NATIONAL ADVISORY COMMITTEE
FOR AERONAUTICS

REPORT:

EQUATIONS FOR THE DESIGN OF
TWO-DIMENSIONAL PROPELLING NOZZLES

By:

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AERONAUTIC SYMBOLS

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2. GENERAL SYMBOLS

- \( W \) Weight = \( mg \)
- \( g \) Standard acceleration of gravity = \( 9.80665 \text{ m/s}^2 \) or \( 32.1740 \text{ ft/sec}^2 \)
- \( m \) Mass = \( \frac{W}{g} \)
- \( I \) Moment of inertia = \( mk^2 \) (Indicate axis of radius of gyration \( k \) by proper subscript.)
- \( \mu \) Coefficient of viscosity

3. AERODYNAMIC SYMBOLS

- \( S \) Area
- \( S_w \) Area of wing
- \( \sigma \) Gap
- \( b \) Span
- \( c \) Chord
- \( A \) Aspect ratio = \( \frac{b^2}{S} \)
- \( V \) True air speed
- \( q \) Dynamic pressure = \( \frac{1}{2} \rho V^2 \)
- \( L \) Lift, absolute coefficient = \( \frac{L}{qS} \)
- \( D \) Drag, absolute coefficient = \( \frac{D}{qS} \)
- \( D_0 \) Profile drag, absolute coefficient = \( \frac{D_0}{qS} \)
- \( D_i \) Induced drag, absolute coefficient = \( \frac{D_i}{qS} \)
- \( D_p \) Parasite drag, absolute coefficient = \( \frac{D_p}{qS} \)
- \( C \) Cross-wind force, absolute coefficient = \( \frac{C}{qS} \)

- \( \alpha \) Angle of attack
- \( \epsilon \) Angle of downwash
- \( \alpha_0 \) Angle of attack, infinite aspect ratio
- \( \alpha_i \) Angle of attack, induced
- \( \alpha_s \) Angle of attack, absolute (measured from zero-lift position)
- \( \gamma \) Flight-path angle
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REPORT No. 907

EQUATIONS FOR THE DESIGN OF
TWO-DIMENSIONAL SUPERSONIC NOZZLES

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National Advisory Committee for Aeronautics

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SUMMARY

Equations are presented for obtaining the wall coordinates of two-dimensional supersonic nozzles. The equations are based on the application of the method of characteristics to irrotational flow of perfect gases in channels. Curves and tables are included for obtaining the parameters required by the equations for the wall coordinates.

A brief discussion of characteristics as applied to nozzle design is given to assist in understanding and using the nozzle-design method of this report. A sample design is shown.

INTRODUCTION

A supersonic nozzle is used to transform parallel flow at sonic velocity into parallel, uniform flow at a supersonic Mach number. The conventional two-dimensional supersonic nozzle consists of the following four main parts arranged in the direction of flow (fig. 1):

1. A subsonic inlet converging in the direction of flow
2. A throat in which the streamlines are parallel to the nozzle axis and sonic velocity of the compressible flow is reached
3. An expanding part with constant or increasing angle of inclination of the nozzle wall to the axis of the nozzle, in which the flow accelerates to supersonic speeds
4. A straightening part of increasing area of cross section in the direction of flow but decreasing angle of inclination of the wall to the nozzle axis; in this part, the flow is turned parallel to the nozzle axis with the desired final Mach number uniform across the exit section.

In a properly designed nozzle, there are no compression or expansion waves in the flow downstream of the straightening portion. A streamline crossing such waves would be altered in direction and Mach number, which is generally undesirable.

The method of characteristics provides a means for obtaining the properties of a fluid moving at supersonic speed past solid surfaces. A particular application of the method of characteristics permits the solution of the inverse problem of obtaining the profile of the solid boundary that would create a desired supersonic flow.

Graphical methods for designing two-dimensional nozzles by the method of characteristics, for example, are reviewed in reference 1. Graphical methods employing characteristics for obtaining nozzles free from waves in the final flow, however, are tedious and subject to the error inherent in construction involving the plotting of many consecutive lines.

The application of the method of characteristics to the analytical design of two-dimensional supersonic nozzles was completed at the NACA Cleveland laboratory in February 1947. Analytical expressions are obtained for the wall contours of the supersonic part of the two-dimensional nozzle. An analytical expression for the straightening part of two-dimensional nozzles, in which source flow is considered to exist in the expanding part, has been derived by Kuno Foelsch of North American Aviation, Inc., but no method is given for creating such source flow. In order to present a complete discussion of two-dimensional nozzle design, the design of nozzle-wall contour for producing source flow in the expanding part of the nozzle and the design of the complementary straightening part are presented. A less complete treatment of this problem from a different point of view has been given by A. O. L. Atkin in a British report.

A working knowledge of the method of characteristics is desirable in order to understand and use the nozzle-design method. For this reason, the form of the method of characteristics most convenient for discussing the method of nozzle design considered is given in an appendix. A summary of the design equations and a sample nozzle design are included.
METHOD OF ANALYSIS

It will be demonstrated that when source flow is created entirely across the nozzle channel at any section, adjacent areas of the flow also have the properties of source flow. On this basis, analytical expressions are derived for the nozzle-wall coordinates required to create a specified source flow in the expanding part of the nozzle and to turn that flow into a uniform stream parallel to the nozzle axis in the straightening part with the desired Mach number. Only irrotational flows are considered in this analysis. The total temperature and the total pressure are constant throughout the flow. The flow adjacent to the nozzle walls is assumed to follow the wall contour at all times.

PROPERTIES OF SOURCE FLOW

In most conventional supersonic nozzles, source flow is approximated at the end of the expanding part of the nozzle. Because of the simple mathematical relations governing source flow, it is desirable to specify that perfect source flow exists at the end of the expanding part of the nozzle to obtain analytical expressions of simple form for the nozzle-wall coordinates.

The essential properties of two-dimensional source flow are illustrated in figure 2. In the supersonic part (solid lines), streamlines are straight and appear to diverge from the apparent upstream source O. All stream tubes with the same included angle \( \theta \) between bounding streamlines carry the same mass flow. From one-dimensional supersonic-flow theory, which applies to this type of flow because the flow is uniform on circular cylindrical surfaces concentric with the apparent source, the Mach number at points a distance \( r \) (fig. 2) from the apparent source is given by the following expression:

\[
A_r = \frac{A_1}{r} = \frac{1}{M_r} \left( \frac{1 + \frac{\gamma - 1}{2} M_r^2}{\gamma + 1} \right)^{\frac{\gamma + 1}{2(\gamma - 1)}} = r
\]

where \( A_r \) is the flow area per unit depth normal to the streamlines at a distance \( r \) from the source and \( A_1 \) is the corresponding flow area at \( M=1 \). (For convenience, all symbols are defined in appendix A.) The parameter \( r_1 \) is the distance from the apparent source to the arc at which the Mach number is unity, corresponding to the location of apparent throat of the source flow. The area of cross section normal to the flow at which \( M=1 \) is

or

\[
r = \frac{A_1}{2 \theta_{max}} \left( \frac{1 + \frac{\gamma - 1}{2} M_r^2}{\gamma + 1} \right)^{\frac{\gamma + 1}{2(\gamma - 1)}}
\]

Equation (1) then becomes

\[
r = \frac{A_1}{2 \theta_{max}} \left( \frac{1 + \frac{\gamma - 1}{2} M_r^2}{\gamma + 1} \right)^{\frac{\gamma + 1}{2(\gamma - 1)}}
\]

EXPANSION WAVES AND CHARACTERISTICS

According to the discussion in appendix B, changes in flow direction and Mach number in diverging channels are produced by a system of expansion waves originating at the channel walls. The change in flow direction due to an expansion wave from one channel wall is constant along Mach lines directed downstream of their point of contact with the channel wall where the wave originates. In the absence of expansion waves from the second wall, these Mach lines are straight and all the flow experiences the same change in direction and Mach number between the same two Mach lines in the expansion wave. If the flow enters the channel with uniform direction and Mach number, the flow direction and the Mach number are constant for the entire flow along these straight Mach lines in the expansion wave. The Mach number and the flow direction are the same as that of the flow moving adjacent to the channel wall at the point of contact with the Mach line. A number can be assigned to the Mach line that is equal to an expansive angular turn about a corner in a wall, bounding the flow, required to convert a sonic flow (\( M=1 \)) to the same Mach number as that along the Mach line, according to the well-known Prandtl-Meyer theory (reference 2). Mach lines so numbered are called characteristics. The characteristics originating at the upper wall of the nozzle (fig. 3) are designated by \( (\Psi_+ \) and from the lower wall by \( (\Psi_- \). Each point in the flow is crossed by a \( (\Psi_+ \) and a \( (\Psi_- \) characteristic corresponding to the two Mach lines through every point in a supersonic flow. The value of \( (\Psi_+ \) assigned to a characteristic represents the counterclockwise angular turning that would be experienced by the streamline coming from the left between the region where the flow is uniform with a Mach number of unity and the \( (\Psi_- \) characteristic in the absence of the system of expansion waves designated by the \( (\Psi_- \) characteristics. Similarly, the value of the \( (\Psi_- \) characteristic represents the clockwise turning experienced by a streamline from the left between the region.
where the Mach number is unity and the \((\Psi_+)\) characteristic in the absence of the system of expansion waves designated by the \((\Psi_+)\) characteristics. The counterclockwise angular turning produced by the expansion wave between two characteristics of the \((\Psi_+)\) set, designated \((\Psi_+)_1\) and \((\Psi_+)_2\), is \((\Psi_+)_2 - (\Psi_+)_1\). Likewise, \((\Psi_-)_2 - (\Psi_-)_1\) represents the clockwise turning of the flow produced by an expansion wave of the \((\Psi_-)\) set. In appendix B, it is also shown that turning the flow in either the clockwise or counterclockwise direction due to the expansion waves from the nozzle walls is accompanied by an increase of the cross section of the flow tubes with a consequent increase in supersonic-flow Mach number.

The deviation of the flow produced by the waves corresponding to one set of characteristics occurs independently of the presence of the wave of the other set. The combined effect of overlapping expansion waves of the \((\Psi_+)\) and \((\Psi_-)\) sets, as shown in zone III of figure 4, is obtained by adding the effect of the two sets of expansion waves considered separately. The total Prandtl-Meyer turning angle \(\Psi\) assigned to a point \(F\) (fig. 4) is the sum of the \((\Psi_+)\) and \((\Psi_-)\) characteristics through the point \(F\). If \(M\) is the Mach number of the flow at \(F\), then from reference 1 or 2

\[
\Psi = (\Psi_+) + (\Psi_-) = \lambda \tan^{-1} \sqrt{M^2 - 1} - \tan^{-1} \sqrt{M^2 - 1} \quad (2)
\]

Also, the total counterclockwise angular deviation of the flow between the direction where the Mach number is unity and the point \(F\) is equal to

\[
\theta = (\Psi_+) - (\Psi_-) \quad (2a)
\]

If the values of \((\Psi_+)\) and \((\Psi_-)\) are known at all points in the irrotational flow, the flow is completely specified because equations (2) and (2a) give the flow Mach number and direction at any point.

In the nozzles considered, the throat section is followed by a part that produces a uniform flow parallel to the axis at section 1-1' (fig. 3) at a Mach number \(M_t\) greater than unity. Methods for creating this uniform flow with the required value of \(M_t\) are discussed elsewhere herein. The nozzle walls at section 1-1' are parallel to the nozzle axis. The first expansion wave emanating from the upper wall due to the counterclockwise turning of the wall at point 1 is bounded upstream by the \((\Psi_+)\) characteristic, making the Mach angle \(\beta_1\) (equal to \(\sin^{-1} \frac{1}{M_1}\)) with the uniform flow of Mach number \(M_1\). Similarly, the first expansion wave emanating from the lower wall due to the clockwise turning of the lower wall at 1' is bounded upstream by the \((\Psi_-)\) characteristic, making the Mach angle \(\beta_1\) with the uniform flow \(M_1\).

The flow in the nozzle between section 1-1' and the downstream characteristics through 1 and 1' is uniform and has the Mach number \(M_t\) because this space is not traversed by waves from either wall. In this zone the value of \(\Psi\) is constant and is designated \(\Psi_t\), corresponding to \(M_t\) (equation (2)). Because the flow is uniform and parallel to the axis at all points in this zone, from equation (2a)

\[
\theta = 0 = (\Psi_+) - (\Psi_-)
\]

and from equation (2)

\[
\Psi_t = \Psi_+ + \Psi_- = 2(\Psi_+) = 2(\Psi_-)
\]

The downstream characteristics through the points 1 and 1' therefore have a value

\[
(\Psi_+)_t = (\Psi_-)_t = \frac{\Psi_t}{2}
\]

Because of the axial symmetry of the flow produced by similar upper and lower nozzle walls, the characteristics through 1 and 1' (fig. 4) arrive at the opposite walls at corresponding points \(E'\) and \(E\), respectively. At any point \(B\) (fig. 4) on the upper wall upstream of \(E\), the wall makes an angle \(\alpha\) with the nozzle axis. Between the points 1 and \(B\), the streamlines moving along the wall will be turned counterclockwise through an angle \(\alpha\). The value of the \((\Psi_+)\) characteristic through \(B\) is therefore

\[
(\Psi_+)_B = (\Psi_+) + \alpha = \frac{\Psi_t}{2} + \alpha
\]

Also, the total counterclockwise angular deviation of the flow between the direction where the Mach number is unity and the point \(F\) is equal to

\[
\theta = (\Psi_+) - (\Psi_-) \quad (2a)
\]

If the values of \((\Psi_+)\) and \((\Psi_-)\) are known at all points in the irrotational flow, the flow is completely specified because equations (2) and (2a) give the flow Mach number and direction at any point.

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\theta = 0 = (\Psi_+) - (\Psi_-)
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and from equation (2)

\[
\Psi_t = (\Psi_+) + (\Psi_-) = 2(\Psi_+) = 2(\Psi_-)
\]

The downstream characteristics through the points 1 and 1' therefore have a value

\[
(\Psi_+)_t = (\Psi_-)_t = \frac{\Psi_t}{2}
\]

Because of the axial symmetry of the flow produced by similar upper and lower nozzle walls, the characteristics through 1 and 1' (fig. 4) arrive at the opposite walls at corresponding points \(E'\) and \(E\), respectively. At any point \(B\) (fig. 4) on the upper wall upstream of \(E\), the wall makes an angle \(\alpha\) with the nozzle axis. Between the points 1 and \(B\), the streamlines moving along the wall will be turned counterclockwise through an angle \(\alpha\). The value of the \((\Psi_+)\) characteristic through \(B\) is therefore

\[
(\Psi_+)_B = (\Psi_+) + \alpha = \frac{\Psi_t}{2} + \alpha
\]
and the value of the \((\Psi_+\) characteristic through \(E'\) (fig. 4) is

\[
\Psi_+ = \Psi'_+ + \alpha + \frac{\Psi_+}{2}
\]  

(4b)

Between the upper nozzle wall and the characteristic through \(I'\) (zone I, fig. 4), the \((\Psi_+)\) characteristics are straight lines because the expansion waves from the upper wall are not crossed by any waves from the lower wall. (See appendix B.) Likewise, in zone II the \((\Psi_-)\) characteristics are straight for corresponding reasons. In zone III expansion waves from the upper and lower walls overlap and the characteristics are curved.

The complete wave pattern for nozzles of the type considered is schematically shown in figure 3. The first expansion waves to leave the nozzle wall at points \(1\) and \(1'\) are bounded upstream by the \((\Psi_+)\) and \((\Psi_-)\) characteristics, respectively. Because of the symmetry of the nozzle, these characteristics arrive at corresponding points \(E'\) and \(E\) on the opposite walls. Therefore, between points \(1\) and \(E\) no expansion waves are incident upon the nozzle walls. In the straight-walled part between sections \(E-E'\) and \(S-S'\), expansion waves are emitted having strength equal to the incident waves from the opposite wall. In order that no expansion waves be emitted from the portion of the wall between \(S\) and \(N\) (straightening part), the wall in this part of the nozzle is curved toward the nozzle axis. The curvature of the nozzle wall is the same as that assumed by the streamline moving along the wall under the influence of the incident expansion waves from the opposite wall. (See appendix B.) No waves are emitted by the wall between points \(S\) and \(N\), therefore, and zones IV and V are traversed by one set of expansion waves whose characteristics are straight.

**SOURCE FLOW IN NOZZLES**

The nozzle-design method considered in this report is based upon establishing source flow at circular-arc section \(E-E'\) (fig. 4). At this section the inclination of the wall to the axis has an assigned value \(\alpha\) and the assigned Mach number of the flow is \(M_e\). The choice of the values of \(\alpha\) and \(M_e\) at section \(E-E'\) is considered in the section "Design of complete nozzle." It will first be shown that if source flow exists at section \(E-E'\) it exists everywhere in zone III. The flow between points in zone III is then related by equation (1a).

This fact, together with the fact that the characteristics in zones I and II are straight, is the basis for establishing an analytical expression for the nozzle-wall contour producing the stipulated source flow at section \(E-E'\).

The point of intersection of the straight line tangent to the nozzle wall at section \(E-E'\) (fig. 4) and the nozzle axis represents the location \(O\) of the apparent source creating the source flow through section \(E-E'\). At all points on section \(E-E'\), the Mach number is constant. At a point on section \(E-E'\) where the flow makes the angle \(\theta\) with the axis, the following relations from equations (2) and (2a) apply:

\[
\Psi_e = (\Psi_+) - (\Psi_-) = \lambda \tan^{-1} \sqrt{\frac{M_e^2 - 1}{\alpha}} - \tan^{-1} \sqrt{\frac{M_e^2 - 1}{\alpha}}
\]

where \(\lambda = \frac{\gamma}{\gamma - 1}\)

\[
\theta = (\Psi_+) - (\Psi_-)
\]

(5a)

At a point \(F\) on section \(E-E'\) through which the flow makes the angle \(\theta\) with the axis, from equations (5) and (6a),

\[
\Psi_e = \frac{\Psi_+ - \theta}{1}(\Psi_- - \theta)
\]

(6b)

holds for all points on section \(E-E'\). The \((\Psi_+)\) and \((\Psi_-)\) characteristics \(GJ\) and \(GH\) make the Mach angle \(\beta\) with the streamline through point \(F\), so that the length of arcs \(HF\) and \(FJ\) are equal according to the equation

\[
r d\theta = HF = dr \tan \beta = FJ = rd\theta_i
\]

(7)

Therefore

\[
d\theta = d\theta_i
\]

(7a)

At point \(G\)

\[
\theta_G = (\Psi_+) - (\Psi_-) = \left[ (\Psi_+) \frac{d(\Psi_+)}{d\theta} d\theta_i \right] - \left[ (\Psi_-) \frac{d(\Psi_-)}{d\theta} d\theta_i \right]
\]

(7b)
From equations (6b) and (7a)
\[ \theta_o = \frac{\partial (\Psi_1)}{\partial \theta} - \frac{\partial (\Psi_2)}{\partial \theta} - \frac{d \theta}{d \theta} \\ \theta_o = (\Psi_1)_{\theta} - (\Psi_2)_{\theta} - \theta_o \tag{8} \]
The streamline through G therefore has the same direction as the streamline through F. Also, from equations (2) and (6b) and the expressions for \((\Psi_+)_G\) and \((\Psi_-)_G\) used in equation (7b), there is obtained
\[ \Psi_0 = (\Psi_1)_G + (\Psi_2)_G = (\Psi_1)_{\theta} + (\Psi_2)_{\theta} + \left[ \frac{\partial (\Psi_1)}{\partial \theta} - \frac{\partial (\Psi_2)}{\partial \theta} \right] d \theta_i \]
From equation (7)
\[ d \theta_i = \frac{dr}{r \tan \beta} \]
and
\[ \Psi_0 = (\Psi_1)_{\theta} + 2 \frac{\partial (\Psi_2)}{\partial \theta} \frac{dr}{r \tan \beta} \tag{9a} \]
But from equation (6b)
\[ \frac{\partial (\Psi_1)}{\partial \theta} - \frac{\partial (\Psi_2)}{\partial \theta} = \left[ (\Psi_1)_{\theta} - (\Psi_2)_{\theta} \right] = -2 \frac{\partial (\Psi_2)}{\partial \theta} \]
From equations (9a) and (9b)
\[ \Psi_0 - \Psi_2 = -d \psi = \frac{dr}{r \tan \beta} \]
and
\[ \frac{d \psi}{dr} = \tan \beta \tag{10} \]
This expression is independent of \(\theta\). Therefore, because \(\tan \beta\) is constant on section E-E', on arcs a constant distance \(dr\) from section E-E' the value of \(d \psi\) is constant. The circular-arc section OG is therefore also a section on which source flow exists. A repetition of the developments just described would establish that source flow exists in zones adjacent to circular arc OG.

In this way, source flow can be shown to exist in zone III (fig. 6) to the left of section E-E'. In the upper half of the nozzle, source flow is limited to the zone (zone III) between section E-E' and the \((\psi_r)\) characteristic through the nozzle wall at point E. (See fig. 4.) At all points in this zone, both \((\psi_r)\) and \((\psi_-)\) characteristics belong to the system of characteristics giving source flow at section E-E'.

At a point K outside this zone, the \((\psi_-)\) characteristic is not part of the system of characteristics that is associated with the source flow specified for section E-E' and the source flow does not include point K.

The existence of the zone of source flow (zone III) (fig. 6) can be shown by physical reasoning as well. If the flow through the nozzle were reversed, the supersonic flow being from the test section to the throat, and if source flow existed at section E-E', then the expansion wave from the wall at point E would be bounded upstream by the \((\psi_-)\) characteristic through E. The influence of the change of wall contour at E would be effective in the flow downstream of this characteristic. Between section E-E' and the \((\psi_-)\) characteristic, no change from source flow would occur.

**COORDINATES OF WALL CONTOUR OF EXPANSION PART OF NOZZLE**

The coordinates of the nozzle walls X, Y that produce source flow at section E-E' are obtained in the following development:

The origin of the coordinates is taken as the apparent throat, and X and Y are taken parallel and normal to the axis of the symmetrical nozzle, respectively. The coordinates of points on the characteristics will be designated \(x, y\).

According to figure 6, the equation of any \((\psi_-)z\) characteristic in zone III is given as
\[ x = r \cos \theta \tag{11a} \]
\[ y = r \sin \theta \tag{11b} \]
where
\[ r = \frac{A_1}{2 \alpha M} \left( 1 + \frac{\gamma - 1}{2} \frac{M^2 - 1}{\gamma + 1} \right)^{\frac{\gamma + 1}{2(\gamma - 1)}} \tag{11c} \]
from equation (1a). From equation (2a)
\[ \theta = (\psi_+) - (\psi_-)_z = (\psi_+) + (\psi_-)_z - 2(\psi_-)_z \]
\[ = \frac{\lambda}{\gamma - 1} \frac{M^2 - 1}{\gamma + 1} \frac{\lambda}{\gamma - 1} \frac{M^2 - 1}{\gamma + 1} \]
\[ = \frac{\psi - 2(\psi_-)_z}{\gamma - 1} \]

The justification for substituting \(A_1\) for \(A_1\) and \(\alpha_M\) for \(\theta_{\text{inlet}}\) in equation (1a) to obtain equation (11b) is based on the following considerations: The mass flow across section E-E' is the same as the mass flow through the nozzle throat, where \(M = 1\). If source flow actually existed for all the flow upstream of section E-E', the flow would be contained between straight lines OE and O' (fig. 4), which make the angle \(\alpha_M\) with the axis. At the hypothetical section \(r_1\) (fig. 2), where \(M = 1\) in source flow, the density and the flow velocity would be the same as the corresponding values for the nozzle throat. Because the mass flow is the same across the \(A_1\) section and the nozzle throat, the flow area must be the same in both cases:
\[ A_1 = A_1 = 2 \alpha_M r_1 \]

In particular, for the \((\psi_-)_{\theta}\) characteristic bounding zone III (fig. 4)
\[ \theta = \frac{\psi - 2(\psi_-)_{\theta}}{\gamma - 1} \]
From equation (4)
\[
\langle\Psi\rangle_r = \frac{\Psi_r}{2}
\]
and from equation (11c)
\[
\theta = \Psi - \Psi_r
\]
and \(r\) is given by equation (11b).

The nozzle-wall coordinates of the expanding part can now be directly obtained from the following argument: Because zone I contains expansion waves from only the upper wall, a characteristic such as UV (fig. 6) in zone I is straight and the flow at every point on the line has the same Mach number and flow direction \(\theta\). (See appendix B.) The flow lines crossing each characteristic at any point in zone I make the same angle \(\beta = \sin^{-1} \frac{1}{M}\) with the characteristic. Source flow exists on circular arc VW concentrice with O and the Mach number is constant for all points on the arc. Point V is common to the arc VW and the Mach line UV and, inasmuch as there are no discontinuities in the flow, the Mach number is constant along the line UVW. Because the flow is considered to have constant total pressure and total temperature, the properties of the fluid, such as density, static pressure, static temperature, and flow speed, are constant along line UVW. The continuity condition for steady flow requires that the mass flow be the same across section E-E' and UVW. If source flow did exist in the entire wedge-shaped zone between the nozzle axis and the straight line OE, the Mach number of the flow across arc TV concentric with O would be the same as actually exists along VW or UV. The mass flow that crosses Mach line UV would cross arc TV with the same density and velocity. The area \(l \sin \beta\) normal to the flow crossing Mach line UV must therefore be equal to the area normal to the assumed source flow crossing TV. As TV is the arc normal to the direction of the assumed source flow,
\[
l \sin\beta = r(\alpha_k - \theta)
\]
By means of the relation \(\sin\beta = \frac{1}{M}\)
\[
l = Mr(\alpha_k - \theta)
\]
If \(X\), \(Y\) and \(x\), \(y\) are taken as the wall coordinates and the coordinates of the \((\Psi)_{+}\) characteristic, respectively, then from figure 6
\[
X = x - l \cos (\beta - \theta) = r \cos \theta - Mr(\alpha_k - \theta) \cos (\beta - \theta)
\]
\[
Y = y + l \sin (\beta - \theta) = r \sin \theta + Mr(\alpha_k - \theta) \sin (\beta - \theta)
\]
Negative values of \(X\) are possible.

All terms in equations (13) and (13a) are functions of \(\lambda\). These functions, taken from equations (11b) and (11d), are listed here for convenience:

\[
r = \frac{A_i}{2\alpha_i M} \left(1 + \frac{\gamma - 1}{2} M^2\right)^{\frac{\gamma + 1}{\gamma - 1}}\]
\[
\theta = \lambda \tan^{-1} \sqrt{\frac{M^2 - 1}{\lambda} - \tan^{-1} \sqrt{M^2 - 1 - \Psi_i}}
\]
\[
\beta = \sin^{-1} \frac{1}{M}
\]
Values for
\[
r(\frac{2\alpha_i}{A_i}) = \frac{1}{M} \left(1 + \frac{\gamma - 1}{2} M^2\right)^{\frac{\gamma + 1}{\gamma - 1}}
\]
and
\[
\theta + \Psi_i = \lambda \tan^{-1} \sqrt{\frac{M^2 - 1}{\lambda} - \tan^{-1} \sqrt{M^2 - 1}}
\]
are given in table I, column 4 and column 3, respectively.
EQUATIONS FOR THE DESIGN OF TWO-DIMENSIONAL SUPERSONIC NOZZLES

VALUE OF $\Psi_I$

If the uniform parallel flow across section $l-l'$ (fig. 4) were at a Mach number of unity, both limiting Mach lines, or characteristics, $l'E$ and $l'E'$ would leave their respective nozzle walls with direction normal to the nozzle axis and would arrive at the opposite wall without displacement downstream. In this case the length of the expanding section of the nozzle would be zero.

The minimum value of $\Psi_I$ required to obtain a length of nozzle sufficient to permit an assigned value of $\alpha_B$ at section $E-E'$ is obtained from the physical requirement that the value of $M$ must always increase with increasing value of $X$, the nozzle-wall coordinate given in equation (13). The minimum value of $M_I$, corresponding to the minimum value of $\Psi_I$, (equation (2)) is obtained from

$$\alpha_B = \frac{(M_I^2 - 1)^{3/2}}{0.6M_I^6}$$

for $\gamma=1.400$. The development of this equation is given in appendix C. Values of $M_I$ less than those given by equation (14) give negative values of $\frac{\partial X}{\partial M}$ in the neighborhood of section $l-l'$. A plot of equation (14) is given in figure 8.

The highest value $\alpha_B$ can have (fig. 8) is about 31°, corresponding to a value of $M_I=2$. Source flow cannot be produced in nozzles with $\alpha_B$ greater than 31°. The corresponding values of $\Psi_I$ given by equation (14) lie between 0 and $\Psi_I$ corresponding to $M_I=2$. The values of $\Psi_I$ plotted in figure 8 are minimum values. Over-all design considerations or ease of computation may suggest values of $\Psi_I$ greater than these minimum values. If a higher value is chosen for $\Psi_I$, the corresponding value of $\alpha_B$ required to obtain the desired value of $M_I$ is computed in a manner to be considered in the section "Design of Complete Nozzle."

WALL CONTOUR OF STRAIGHTENING PART OF NOZZLE

The straightening part of the nozzles considered converts a supersonic source flow into a uniform flow parallel to the nozzle axis. Consider a supersonic source flow at circular-arc section $S-S'$ concentric with apparent source (fig. 7). Circular-arc section $S-S'$ may be coincident with section $E-E'$ or may be a section downstream of section $E-E'$. If it is downstream, source flow exists across the entire straight-walled channel of the nozzle between sections $E-E'$ and $S-S'$. Because the nozzle-wall curvature between points $S$ and $N$ will influence the flow only downstream of the forward Mach line through point $S$ ($\Psi_+s$ characteristic), the source flow ends at the ($\Psi_+s$) characteristic upstream of point $F$.

The straightening part of the nozzle is designed on the principle that the wall contour is shaped to conform to the curvature of the streamline adjacent to the wall that is turned by the incident expansion wave from the opposite wall. No emission of either expansion or compression waves occurs from the wall so shaped. This point is discussed in appendix B. The ($\Psi_+)s$ characteristic therefore represents the downstream limit of all expansion waves emanating from the upper nozzle wall. The zone enclosed by the lines joining the points $SFN$ contains waves that originate at the lower nozzle wall only. The ($\Psi_-$) characteristics in this zone are therefore straight. (See appendix B.)

The equation of the limiting characteristic ($\Psi_+)s$ is obtained by making use of the fact that source flow exists in the adjacent area upstream of the ($\Psi_+)s$ characteristic. If $x$ and $y$ are the coordinates parallel and normal to the nozzle axis, respectively, of the ($\Psi_+)s$ characteristic with the origin taken at the apparent source, then

$$x = r \cos \theta$$
$$y = r \sin \theta$$

where $r$ is given as a function of $M$ by equation (11b). By the same reasoning used to obtain equation (11d), $\theta$ is obtained as

$$\theta = (\Psi_+)S - (\Psi_-) = -[\Psi_+)S + (\Psi_-)] + 2(\Psi_+)S = 2(\Psi_+)S - \Psi$$

$$= 2(\Psi_+)S - \lambda \tan^{-1} \sqrt{M_I^2 - 1} + \tan^{-1} \sqrt{M_I^2 - 1}$$

The values of $M$ range from $M_S$ to $M_F$. The value of ($\Psi_+)S$ is obtained from the observation that a streamline along the nozzle axis arriving at point $F$ (fig. 7) will have crossed all expansion waves emanating from both walls and will therefore be at the final flow Mach number $M_F$ corresponding to a total turning angle $\Psi_F$. Because the inclination of the flow to the axis is zero at point $F$, values of ($\Psi_+$) and ($\Psi_-$) of the characteristics through point $F$ are related by the equation

$$\theta = (\Psi_+)F - (\Psi_-)F = 0$$
Moreover, the (\(\Psi_1\)) and (\(\Psi_2\)) characteristics through \(F\) are the limiting characteristics (\(\Psi_1\)) and (\(\Psi_2\)), respectively; therefore

\[
(\Psi_1)_F - (\Psi_1)_S = (\Psi_1)_F
\]

Because in a symmetrical nozzle (\(\Psi_1\)) = (\(\Psi_2\)), and

\[
\Psi_F - (\Psi_1)_F + (\Psi_2)_F - 2(\Psi_1)_F = 2(\Psi_1)_S
\]

\[
(\Psi_1)_S = \frac{\Psi_F}{2}
\]  

(16a)

From the fact that the flow through point \(F\) is at the final Mach number \(M_F\), (\(\Psi_F\)) = (\(\Psi_2\)) and equation (16) can then be written

\[
\theta = \Psi_F - \Psi = \Psi_F - \Psi = \frac{\tan^{-1}\sqrt{M^2 - 1} + \tan^{-1}\sqrt{M^2 - 1}}{M}
\]  

(16b)

where \(M\) has values between \(M_s\) and \(M_F\). The value of \(M_s\) corresponds by equation (2) to \(\Psi_s\) (\(\Psi_1\)) + \(\Psi_2\). Because the flow direction at point \(S\) makes the angle \(\alpha_e\) with the nozzle axis (fig. 7),

\[
(\Psi_1)_S = (\Psi_2)_S = \alpha_e
\]

Therefore, from equation (16a)

\[
\Psi_s - 2(\Psi_1)_S - \alpha_e = \Psi_F - \alpha_e
\]

(16c)

The coordinates \(X, Y\) of the nozzle wall for the straightening part are obtained in a manner similar to those for the expanding section. A characteristic (such as \(GH\) in zone IV (fig. 7)), included in area S\(F\)N is straight and the Mach number is constant along the characteristic. (See appendix B.2.1) Subsequently, the flow direction, pressure, temperature, and velocity are constant along such characteristics. Along the circular are GD, source flow exists and the Mach number, pressure, and temperature are constant. Only the flow direction varies along GD. As point \(G\) is common to \(GH\) and are GD, the physical properties of the fluid and the flow speed along GD are the same as along \(GH\). The area of flow normal to the streamline along HGD is

\[
A = r\theta - l \sin \beta
\]  

(17)

If source flow had existed downstream of section S-S', as it would have if the nozzle walls had continued downstream straight through \(S\), then the mass flow across are BGD would have the same value as across HGD. The fluid would also have had the same pressure, temperature, density, and flow Mach number as actually exists on arc GD, which does support source flow. The area normal to the flow across BGD would therefore be the same as for the flow that does cross HGD and from equation (17)

\[
r\alpha_e - r\theta + l \sin \beta = \frac{1}{M}
\]  

(17a)

As \(\sin \beta = \frac{1}{M}
\]

\[
l = M(r\alpha_e - \theta)
\]  

(17b)

Therefore, if \(X, Y\) and \(x, y\) are the nozzle-wall coordinates of the straightening part and the (\(\Psi_1\)) characteristic, respectively, then

\[
X = r + l \cos (\theta + \beta) = r \cos \theta + M(r\alpha_e - \theta) \cos (\theta + \beta)
\]  

(18)

\[
Y = y + l \sin (\theta + \beta) = r \sin \theta + M(r\alpha_e - \theta) \sin (\theta + \beta)
\]  

(18a)

The values of \(r\) are obtained from equation (11b) and are tabulated in table 1, column 4. The value of \(\theta\) is obtained from \(\Psi_2 - \theta\) (given in table 1, column 3) corresponding to the value of \(M\) for which the point \((X, Y)\) (equations (18) and (18a)) is being obtained. The value of \(\Psi_2\) corresponding to \(M_F\), the final Mach number of the nozzle, is also obtained from table 1, column 3. The value of \(M\) to be used in equations (18) and (18a) ranges from \(M_s\) to \(M_F\).

**DESIGN OF COMPLETE NOZZLE**

Supersonic nozzles are generally specified in terms of the cross-sectional area of final uniform flow \(A_1\) and the final Mach number \(M_F\). The nozzle-throat area is obtained by the one-dimensional-flow equation

\[
A_1 = \left(1 + \frac{\gamma - 1}{2} M^2 \right)^{\frac{\gamma + 1}{2}}
\]

for which values are tabulated in table 1, column 4.

**NOZZLE WITHOUT STRAIGHT-WALLED PART**

The shortest nozzles may be designed by the method reported are those without a straight-walled part between sections E-E' and S-S'. The straightening part immediately follows the expanding part. For a given value of \(M_F\) and given final Mach number \(M_F\), the value of \(\alpha_e\) is fixed by the following consideration: Because \(\alpha_e\) is the angle through which the nozzle wall turns between section 1-1' and section E-E' (fig. 3), then

\[
(\Psi_1)_k - (\Psi_1)_1 = \alpha_e
\]  

(19)

By equation (4)

\[
\alpha_e = \frac{(\Psi_1)_k - (\Psi_1)_F}{2}
\]  

(19a)

The value of (\(\Psi_1\)) remains constant at (\(\Psi_1\)) downstream of the (\(\Psi_1\)) characteristic because no additional waves are emitted from the upper wall of the shortest nozzle (fig. 9). The value of (\(\Psi_2\)) likewise remains constant at (\(\Psi_2\)) downstream of the (\(\Psi_2\)) characteristic. At the end of the nozzle, where the flow is parallel to the nozzle axis with a uniform Mach number \(M_F\),

\[
\theta = 0 = (\Psi_1)_k - (\Psi_1)_F
\]

\[
(\Psi_2)_F = (\Psi_1)_F + (\Psi_2)_F - 2(\Psi_1)_F - 2(\Psi_2)_F
\]

(19b)

From equation (19a), therefore

\[
\alpha_e = \frac{\Psi_2 - \Psi_1}{2}
\]  

(19c)
EQUATIONS FOR THE DESIGN OF TWO-DIMENSIONAL SUPersonic Nozzles

The angle $\alpha_k$ is always less than one-half the equivalent turning angle $\Psi_I$ required to obtain the final Mach number $M_f$.

Considerations of nozzle construction or flow stability may suggest a desirable value of $\alpha_k$. Then $\Psi_I$ is given by equation (19c) for a nozzle of given final Mach number $M_f$. The value of $\alpha_k$ chosen must correspond to a value of $\Psi_I$ by equation (19c) that is equal to or greater than the minimum $\Psi_I$ computed by equation (14) for the same value of $\alpha_k$. (See fig. 8.) A small saving in length of nozzle is made if a value of $\alpha_k$ and the corresponding value of $\Psi_I$ are obtained from the simultaneous solution of equations (14) and (19c). These values are given in a plot of $\alpha_k$ and the corresponding minimum value of $\Psi_I$ required is given in figure 10 for a range of values of $M_f$ from 1 to 10. In the high range of values of final Mach number $M_f$, $\Psi_I$ exceeds $\alpha_k$. If large values of $\Psi_I$ are undesirable, lower values may be used in conjunction with a straight-walled part of the nozzle as discussed in the next section.

NOZZLE WITH STRAIGHT-WALLED PARTS

If nozzles are desired having known values of $\alpha_k$ and $\Psi_I$ less than those given by equations (14) and (19c) (fig. 10), then a straight-walled portion of the nozzle is required downstream of section $E-E'$ to obtain the desired value of $M_f$. The length of the required straight-walled part is obtained as follows: According to equation (11b), which applies to source flow, the axial distance between circular arc sections $E-E'$ and $S-S'$ is

$$r_s - r_e = \frac{M_f}{2\alpha_k} \left[ \frac{1}{M_s} (1 + \frac{\gamma - 1}{2} M_s^2) \right]^{\frac{\gamma + 1}{2(\gamma - 1)}}$$

The values of $M_e$ and $M_s$ are obtained from the corresponding values of $\Psi_K$ and $\Psi_s$ evaluated in the following manner:

The expression for $\Psi_K$ is obtained from equations (4) and (4a) and figure 4 as

$$\Psi_K = (\Psi_1)_K + (\Psi_2)_K = \frac{\Psi_f}{2} + \alpha_K + \frac{\Psi_f}{2} - \Psi_I \div \alpha_K \quad (20a)$$

From equation (10c)

$$\Psi_s = \Psi_f - \alpha_k \quad (20b)$$

The values of $\Psi_K$ and $\Psi_s$ from equations (20a) and (20b) provide by means of table 1, columns 1 and 3, the corresponding value of $M_K$ and $M_s$ required in equation (20). The values of $r_s$ and $r_e$ likewise can be obtained from table 1, column 4. The only theoretical condition on the choice of

**Figure 9. Limiting characteristics in nozzle without straight-walled part.**

**Figure 10. Minimum initial turning angle $\Psi_I$ and maximum wall-expansion angle $\alpha_k$.**

$\Psi_I$ and $\alpha_k$ is that $\Psi_I$ shall not be less than the value given by equation (14) (fig. 8) for the value of $\alpha_k$ chosen (less than 31°).

**DESIGN OF INITIAL EXPANSION PART**

Exact nozzle-wall contours for converting a uniform flow at Mach number unity to a uniform supersonic flow at Mach number $M_f$ can be obtained by shaping the nozzle walls to conform to the streamlines corresponding to the turning of a sonic flow about a corner according to Prandtl-Meyer theory. (Complete nozzles built according to this method have excessive length for high final Mach numbers. This length is undesirable if thick boundary layers on the nozzle walls are to be avoided.)
Four applications of the use of the solution for the turn about a corner to obtain the wall coordinate of the initial expansion part are illustrated in figure 11. In figure 11 (a) is shown the subsonic entrance part, the nozzle throat, the initial expansion part, and the expanding part of the nozzle. The lower wall of the initial expansion part is a sharp corner at C with an angle equal to \( \Psi \). The upper wall has the contour of a streamline of the flow around the sharp corner. In figure 11 (b) is illustrated the same type of initial expansion part in which the sharp corner at C of figure 11 (a) is replaced by a streamline of the flow around the sharp corner. In the arrangements of both figures 11 (a) and 11 (b), the axis of the subsonic entrance makes the angle \( \Psi \) with the axis of the supersonic part of the nozzle. The axis of the subsonic inlet can be made parallel to, but offset from, the axis of the supersonic part of the nozzle by producing the initial expansion of the flow by means of a counterclockwise and clockwise turning of the flow about a corner at the upper wall (point C, fig. 11 (c)) and the lower wall (point C') each of angle \( \Psi/2 \). As in the case shown in figure 11 (b), the

Figure 11. Methods of designing initial expansion part of nozzle.
corners at $C_1$ and $C_2$ can be replaced by streamlines. The arrangement illustrated in figure 11 (d) uses a plug whose contours downstream of the throat are shaped to conform to streamlines for the flow around the corners $C$ and $C'$ on the upper and lower walls, respectively. The turning angle at $C$ and $C'$ is $\Psi_1$ degrees. The initial expansion of the flow is, in effect, accomplished by two separate initial expansion parts in parallel. The axis of the subsonic entrance is in line with the axis of the supersonic part of the nozzle.

An alternative form of the arrangement of figure 11 (d) is shown in figure 11 (c). No plug is required in this initial expansion part. The expansion waves arising at the turns at $C$ and $C'$ are intercepted without further remission by the opposite walls. As all the streamlines cross the expansion waves from both the upper and lower wall, the turning angles at $C$ and $C'$ are $\Psi_1/2$. The wall contours of the arrangement shown in figure 11 (e) are not streamlines of a Prandtl-Meyer turn about a corner but must be obtained by the standard graphical method to be discussed.

The expressions for the coordinates of the wall contour in which the initial turning of the flow is produced are now obtained. In figure 12 is shown the supersonic flow about the corner of a two-dimensional wall in a supersonic flow of infinite extent. According to Prandtl-Meyer theory, the Mach number of the flow is constant along radial lines from the corner and all flow lines crossing a given radial line are parallel at the radial line. For a flow line a distance $d_1$ from the corner $C_1$ along a radial line, the total flow area normal to the flow, $A_{t_1}$, is $d_1 \sin \beta$. From the geometrical relation shown in figure 12, the coordinates of a given stream-line (wall coordinates) are

$$X_1 = d_1 \cos (\beta + \Psi_1 - \Psi)$$
$$Y_1 = d_1 \sin (\beta + \Psi_1 - \Psi)$$

where $\beta = \sin^{-1} \frac{1}{M} (1 \leq M \leq M_0)$. The value of $d_1$ is obtained from the one-dimensional flow relation

$$d_1 = d_0 \left(1 + \frac{\gamma - 1}{2} M^2\right)^{\frac{\gamma + 1}{2(\gamma - 1)}}$$

$$d_0 = \frac{d_1}{\sin \beta} \frac{1}{M} \left(1 + \frac{\gamma - 1}{2} M^2\right)^{\frac{\gamma + 1}{2(\gamma - 1)}}$$

When the short wall of the initial expansion part is a sharp corner, then $d_0$ is equal to the width of the nozzle throat. If both walls in the initial turning portion of the nozzle are to conform to streamlines as illustrated in figure 11 (b), the throat width is given by $d_0 - b$. The coordinates of the long wall are given by equations (21) and (21a) and of the short wall by the same equation with $b_1$ and $b_0$ substituted for $d_1$ and $d_0$, respectively, in equations (21), (21a), and (21b). The values of $d/d_0$ are given in table I, column 5.

When the initial expansion to $M_1$ is accomplished in two steps, as shown in figure 11 (c), the coordinates of the walls of the first part are given by equations (21) and (21a) with $\Psi_1$ replaced by $\Psi_1/2$. The coordinates of the wall of the second section about point $C_2$ are obtained from the geometric relations illustrated in figure 13. The angle between the flow direction at $R$ and at $G$, where the flow direction is parallel to the nozzle walls, is $\Psi_1 - \Psi$. If $D$ is a point on the wall opposite to the location of corner $C_2$ and $d_2$ is the variable length $C_2 D$, then the coordinates of the wall are

$$X_2 = d_2 \cos (\beta + \Psi_1 - \Psi)$$
$$Y_2 = d_2 \sin (\beta + \Psi_1 - \Psi)$$

The coordinate axes at $C_2$ are turned at an angle $\Psi_1/2$ with respect to the axes at $C_1$. The value of $M$ ranges from $M_n$ to $M_1$, where $M_n$ corresponds to $\Psi_n/2$, or

$$\frac{\Psi_1}{2} \leq \Psi \leq \frac{\Psi_1}{2}$$

![Figure 12. Two-dimensional supersonic flow about corner.](image-url)
Moreover,

\[ d_z = d_n \left( \frac{1 + \frac{\gamma-1}{2} M_z^2}{\frac{\gamma+1}{2}} \right)^{\frac{\gamma+1}{\gamma-1}} \]  

(22b)

(from equation (21b)). Values of \( d_z/d_n \), shown as \( d/d_n \), are given in table 1, column 5. Point \( C_z \) can be coincident with point \( B \) (fig. 13).

If the coordinates of the walls with smooth turns (fig. 14) are desired in place of the sharp turns at \( C_1 \) and \( C_2 \), they are obtained as before with \( b_1 \), \( b_2 \), and \( b_3 \), substituted for \( d_1 \), \( d_2 \), and \( d_3 \), respectively, in equations (21) to (22a). The nozzle-throat width is then \( d_n - b_3 \).

When a plug is used in the initial expansion part of the nozzle, as in figure 11(d), each wall has a turn equal to \( \Psi_2 \); each turn influences the flow between the corresponding wall and the plug. The coordinates of the plug (fig. 11(d)) downstream of the throat are given by equations (21) and (21a), in which \( d_0 \) is now the distance from the wall to the plug at the throat section \( CE \). Smooth turns can be substituted for the corners at \( C \) by the method discussed in connection with figure 11(b). Boundary-layer development on the plug may produce an undesirable wake. This condition may be alleviated by the boundary-layer-removal arrangements illustrated in figure 11(d).

A graphical method for obtaining wall contours for the initial expansion corresponding to the configuration shown in figure 11(e) is illustrated in figure 15. The system of \((\Psi_2)\) and \((\Psi_4)\) characteristics emanating from the corners \( C \) and \( C' \) are curved in zone I to account for the effect of one set of expansion waves on the other in accordance with the discussion of appendix B. In zones II and III, the characteristics are straight because the nozzle walls are curved to prevent emission of waves downstream of points \( C \) and \( C' \). Because all expansion waves from \( C \) and \( C' \) remain upstream of the \((\Psi_2)\) and \((\Psi_4)\) characteristics equal to \( \Psi_2/2 \), \((\Psi_2)\) is constant everywhere in zone II at a value of \( \Psi_2/2 \). In zone III, \((\Psi_4)\) is likewise constant everywhere at \( \Psi_2/2 \). If \( M_t \) represents the Mach number of the flow at section 1-1', (fig. 15), then the width of the nozzle at section 1-1' is obtained from the one-dimensional supersonic-flow relation

\[ \frac{w}{d_n} = \frac{1 + \frac{\gamma-1}{2} M_t^2}{\frac{\gamma+1}{2}} \left( \frac{\frac{\gamma+1}{\gamma-1}}{\frac{\gamma+1}{2}} \right) \]  

(23)

Values of the right side of equation (23) are given in table I, column 4.

Because the change in wall contour between points \( B \) and \( G \) (fig. 15) is made to conform to the curvature of the adjacent streamlines produced by the incident expansion waves, the change in wall angle \( \alpha \) between \( B \) and \( G \) on the upper wall (zone II), is

\[ \Delta \alpha = -\theta_1 - \theta_2 = (\Psi_2)_a - (\Psi_2)_b - (\Psi_4)_a + (\Psi_4)_b \]  

(23a)

or, because \((\Psi_4)_a = (\Psi_4)_b = \frac{\Psi_4}{2}\),

\[ \Delta \alpha = -(\Psi_2)_b - (\Psi_2)_a \]  

(23b)

Similarly, on the lower wall (zone III),

\[ \Delta \alpha = -(\Psi_4)_b - (\Psi_4)_a \]  

(23b)

Beginning the graphical layout of the walls from section 1-1', which has the calculated width \( w \), in the manner shown at the lower wall (fig. 15), is advisable. This procedure ensures that the ratio of the area at section 1-1' to the throat section is correct and gives the desired value of \( M_t \). The line \( HJ \) is drawn, making the angle \( \Delta \alpha \) with the nozzle axis as determined by equation (23b). The line \( JK \) is drawn likewise, making the angle \( \Delta \alpha \) with line \( HJ \) as determined from equation (23b) with \( K \) and \( J \) substituted for \( J \) and \( H \), respectively. The polygon obtained by the method just described is replaced by a smooth curve through the vertices of the polygon.
The system of characteristics for zone I of the initial expansion part of nozzles having values of $M_1$ up to 1.536, which corresponds to an initial turning angle of $13^\circ$, is reproduced in figure 16. The zone I characteristics for an initial expansion part of equivalent angle $\Psi_1$ are obtained by selecting all $(\Psi_+)$ and $(\Psi_-)$ characteristics having values equal to and less than $\Psi_1/2$. The zone II characteristics are obtained by continuing the set of $(\Psi_+)$ characteristics as straight lines in the direction of the tangent to the characteristics at their point of intersection with the $(\Psi_-)$ characteristic equal to $\Psi_1/2$. The zone III characteristics are obtained by continuing the set of $(\Psi_+)$ characteristics as straight lines in the direction of the tangents to the characteristics at their intersection points with the $(\Psi_-)$ characteristic equal to $\Psi_1/2$. A plot similar to that given in figure 15 results. Because the wall contour is determined by the zone II and zone III characteristics, the zone I characteristics need not be plotted. From zone I is obtained the direction and the coordinates of the zone II and zone III characteristics at the point of contact with the limiting $(\Psi_+)$ and $(\Psi_-)$ characteristics equal to $\Psi_1/2$. The direction and the coordinates of the characteristics can be obtained from the coordinate system given in figure 16. Tracings from figure 16 will be inaccurate because of the distortion of the figure during reproduction. The system of characteristics is given for a nozzle having a throat width of 24 inches.

The coordinates of the characteristics for nozzles having a different throat width $A_t$ are obtained by multiplying all coordinates given in figure 16 by $A_t/24$. The slopes of the characteristics remain unaltered.

**ESTIMATION OF NOZZLE LENGTH**

The length of the supersonic part of the nozzle (fig. 17) is

$$L = X_f - X_t + L_e$$  \hspace{1cm} (24)

As $X_f$ is the coordinate of the downstream end of the nozzle where $M$ is equal to $M_E$, its value is given by equation (18) with $\theta = 0$

$$X_f = r_f (1 + M_{AE} \cos \beta)$$  \hspace{1cm} (24a)
where \( r_f\) and \( \beta_f\) are obtained from table I with \( M=\frac{M}{f} \). The value of \( X_I\), given by equation (13), corresponds to the coordinate of section 1-1' where \( M\) equals \( M_{f} \) and \( \theta=0 \)

\[
X_I = r_f (1 - M_{\alpha e} \cos \beta_f) \quad (24b)
\]

Negative values of \( X_I\) are possible.

The length of the initial expansion part \( L_e\) (measured along the nozzle wall) is generally less than 10 percent of the total length of the nozzle. The following approximate expressions for \( L_e\) will in general suffice:

1. For one turn about a corner (fig. 11 (a)),

\[
L_e = d_o \tan \gamma = d_o \tan (90^\circ + \psi_f - \beta_o) \quad (24c)
\]

where \( \psi_f\) is obtained from table I for \( \theta = M_{f} \).

2. For two turns in succession about a corner at each wall (fig. 11 (c)),

\[
L_e = d_o \tan \gamma + w_t \tan \gamma = d_o \tan (90^\circ + \frac{\psi_f}{2} - \beta_o) + w_t \tan (90^\circ - \beta_o) = d_o \cot \left( \frac{\beta_o}{2} \right) + w_t \cot \beta_f \quad (24d)
\]

where \( \beta_o = \sin^{-1} \frac{1}{M_{f}} \) and \( M_{f}\) corresponds to \( \frac{\psi_f}{2} \) from table II.

3. For the nozzle with the plug (fig. 11 (d)), the value of \( L_e\) is 0.

4. For the short nozzle at the throat (fig. 11 (e)), the axial length of the corresponding initial expansion part is approximately

\[
L_e \approx d_o + w_t \cot \beta_f \quad (24e)
\]

REMARKS ON APPLICATION OF DESIGN METHOD

Mathematical expressions for the wall coordinates of supersonic nozzles in which source flow is developed are valid for values of \( \alpha e\) equal to or less than 31°. The assumption that the flow follows the nozzle wall for values of \( \alpha e\) up to 31° must be verified by experiment. The use of sharp corners at the initial expansion part must be checked as well. Until this check is made, \( \alpha e\) may well be restricted to known safe values and smooth turns used instead of sharp corners. Because of the favorable pressure gradient in the expansion part of the nozzle, however, the flow will probably follow the nozzle wall for all values of \( \alpha e\) permitted by the theory. Satisfactory flow around sharp corners is also likely for the same reason.

A sample calculation is given in table III of all the design parameters and typical wall coordinates for two nozzles having a final Mach number of \( M=3.50\) and a final width of 10 inches. One nozzle has an initial expansion part consisting of one turn about a sharp corner and belongs to the class of shortest nozzles. The other nozzle has an initial turning part consisting of two turns about sharp corners and contains a straight-walled part.

No account was taken of the effect of boundary-layer growth on the walls on the nozzle flow. If the proper distribution of boundary-layer displacement thickness is known, the local \( Y\) coordinates obtained by the equations of this report should be increased by this boundary-layer thickness. It is important to correct the shape of the straight-walled part of the nozzle for the boundary layer in order to avoid the emission of uncompensated compression waves that may produce a shock front somewhere in the flow.

Flight Propulsion Research Laboratory,
National Advisory Committee for Aeronautics,
Cleveland, Ohio, June 1, 1948.
APPENDIX A

SYMBOLS

The following symbols are used in this report and are illustrated in the figures:

- A: area normal to flow direction (Because unit depth is assumed at all nozzle sections, the area at any section is numerically equal to the width of that section.)
- \( A_1 \): source-flow area normal to flow direction at section where \( M = 1 \) (equivalent throat area)
- \( A_d \): area normal to flow direction in expansive turn around corner
- \( A_f \): cross-sectional area of nozzle bearing uniform flow at \( M_f \) (nozzle exit)
- \( A_r \): source-flow area normal to flow lines at radial distance \( r \) from apparent source
- \( A_t \): nozzle-throat area
- \( b, b_n, b_1, b_2 \): radial distances from "corner" to streamline representing adjacent nozzle wall (See figs. 11 to 14.)
- \( d, d_0, d_1, d_2 \): radial distances from "corner" to streamline representing remote nozzle wall (See figs. 11 to 14.)
- \( D \): displacement
- \( L, L_e \): length of supersonic part of nozzle
- \( l \): distance along characteristic from nozzle wall to limiting characteristic \( (\Psi_2)_{S'} \)
- \( M, M_f \): Mach number
- \( M_e \): Mach number of flow at circular-arc section \( E-E' \)
- \( M_i \): final Mach number of nozzle flow
- \( M_n \): Mach number of flow at first half of initial expansion part
- \( M_r \): Mach number of flow at circular-arc section bearing source flow at distance \( r \) from source
- \( M_s \): Mach number of flow at circular-arc section \( S-S' \)
- \( r \): radial distance along streamline or nozzle axis from apparent source
- \( r_1 \): radial distance between apparent source and circular-arc section at which sonic velocity \((M=1)\) exists in source flow
- \( r_k \): radial distance of circular-arc section \( E-E' \) from apparent source
- \( r_l \): radial distance between apparent source and location of point on axis where \( M_f \) is first attained
- \( r_s \): distance along nozzle axis from apparent source to \((\Psi_2)_{S'} \), or \((\Psi_2)_{S''} \)
- \( w_f \): width of section \( I-I' \)
- \( X, Y \): nozzle-wall coordinates
- \( X_1, Y_1 \): nozzle-wall coordinates of initial expansion part opposite first corner
- \( X_2, Y_2 \): nozzle-wall coordinates of initial expansion part opposite second corner
- \( X_f \): distance of downstream end of nozzle from apparent source
- \( X_i \): distance of section \( I-I' \) from apparent source
- \( x, y \): coordinates of characteristic
- \( \alpha \): inclination of nozzle wall to nozzle axis
- \( \alpha_0 \): maximum inclination of nozzle wall to nozzle axis (corresponds to wall inclination between circular-arc sections \( E-E' \) and \( S-S' \))
- \( \beta \): Mach angle \((\beta = \sin^{-1} 1/M)\), angle between streamlines and Mach line or characteristic
- \( \beta_f \): Mach angle in final uniform nozzle flow
- \( \beta_i \): Mach angle at section \( I-I' \)
- \( \beta_n \): Mach angle at first half of initial turning part
- \( \gamma \): ratio of specific heats
- \( \zeta \): angle between characteristics bounding zone of expansion waves from corner \( C \)
- \( \theta \): angle of inclination of streamline to nozzle axis
- \( \theta_{aux} \): one-half included angle between boundary streamlines of source flow (maximum possible \( \theta \) in source flow)
- \( \lambda \): 
- \( \xi \): angle between downstream characteristic through corner \( C_0 \) and section \( I-I' \)
- \( \Psi \): equivalent Prandtl-Meyer turning angle
- \( (\Psi_{+}) \): characteristic originating at upper nozzle wall
- \( (\Psi_{-}) \): characteristic originating at lower nozzle wall
- \( \Psi_f \): value of \( \Psi \) at nozzle exit
- \( \Psi_i \): value of \( \Psi \) at section \( I-I' \)

Points along the nozzle wall or in the flow are designated by letters; letters for points along the lower nozzle wall are primed. Sections (cross sections through the two-dimensional flow, which are therefore only lines) are designated by the two letters ending the lines. Point-designation letters are in some places used as subscripts for clarity. Zones (region of different kinds of flow) are designated by Roman numerals; parts of the nozzle, which, like the zones, have two dimensions, are called by name. The following location letters are used:

- \( C \): corner in nozzle wall bounding supersonic flow
- \( C' \): corner in lower nozzle wall corresponding to \( C \)
- \( E \): point on upper nozzle wall at circular-arc section at which source flow is first established across entire channel of nozzle
- \( E' \): point on lower nozzle wall corresponding to \( E \)
- \( I \): point on upper nozzle wall representing downstream boundary of initial expansion part
- \( I' \): point on lower nozzle wall corresponding to \( I \)
- \( O \): apparent source
- \( S \): point on upper nozzle wall at last circular-arc section at which source flow exists across entire nozzle channel
- \( S' \): point on lower nozzle wall corresponding to \( S \)

Other capital letters are used to designate arbitrarily chosen points and as subscripts referring to those points; \( a, b, c, \) and \( d \) are used as subscripts in appendix B to indicate hypothetical values.
APPENDIX B

METHOD OF CHARACTERISTICS IN NOZZLE DESIGN

The form of the method of characteristics found most convenient for designing two-dimensional nozzles is described. Irrotational flows with total temperature and total pressure constant throughout the field are considered.

The starting point taken in setting up the method of characteristics used is conveniently discussed in terms of a uniform two-dimensional sonic or supersonic flow turning around a sharp corner of a wall along which the flow passes (fig. 18 (a)). The streamlines are turned about the corner with increasing Mach number, as at C₁ and C₂, in wedge-shaped zones BC₂D and EC₁F, in which the static pressure decreases and the velocity increases in the direction of the flow. Such zones of decreasing pressure and increasing velocity are called expansion waves. Along radial lines through C₁ and C₂ the velocity, pressure, density, temperature, flow Mach number, and flow direction are constant. These radial lines are Mach lines that make the Mach angle with the local flow direction \( \beta = \sin^{-1} \frac{1}{M} \). Downstream of the bounding Mach line C₁D, the flow is uniform and parallel to wall C₁C₂. At the corner in the wall at C₂, the second wedge-shaped zone has a Mach line C₂E as the upstream boundary, which makes the same Mach angle \( \beta \) with the flow as does the Mach line C₁D because the flow between these two lines is uniform.

As the length of wall C₂C₂ has no effect on the direction and Mach number of the flow at line C₂E, the point C₂ could be made coincident with C₁ without altering the flow at C₂F. The change in Mach number and direction of the flow can therefore be considered to be a function only of the angle through which the flow is turned. Any stream tube having a supersonic Mach number can be considered to have come from a sonic flow (\( M = 1 \)) turned about a corner of angle \( \Psi \). The expression relating the flow Mach number and corresponding turning angle (reference 2) is, in the notation of this paper,

\[
\Psi = -\lambda \tan^{-1} \sqrt{M^2-1} - \tan^{-1} \sqrt{M^2-1} \quad (B1)
\]

Because the Mach number of the flow is constant along Mach lines radiating from C₁ and C₂, each Mach line is assigned a value of \( \Psi \) equal to the turning of the sonic flow required to give the corresponding Mach number. It is convenient to subscript these values of \( \Psi \) as \( (\Psi) \) to indicate that the flow is deviated in a clockwise direction from the direction of the flow at sonic speed when crossing the Mach line originating at C₁ or C₂. A Mach line to which a value of \( \Psi \) has been assigned will be called a characteristic. The angular turning of the flow produced by an expansion wave is equal to the difference in the values of \( \Psi \) of the characteristics bounding the wave. When the wall curves uniformly from C₁ to C₂, as in figure 18 (b), at each point in the wall the turning of differential angle \( d\Psi \) is considered to take place.
EQUATIONS FOR THE DESIGN OF TWO-DIMENSIONAL SUPERSONIC NOZZLES

Numbered according to the total counterclockwise angular deviation experienced by the flow arriving at the characteristics and indicated by $(\Psi_+)$, then the total turning experienced by the flow going from the sonic zone to point $P$ (fig. 18 (e)), for example, is

$$\Psi = \Psi_+ + \Psi_+ = (\Psi_+) + (\Psi_-) = \lambda \tan^{-1} \sqrt{M_+^2 - 1} = \tan^{-1} \sqrt{M_+^2 - 1}$$

(B2)

The net counterclockwise angular deviation of the flow along $C_1P$ from the flow direction in the sonic zone is

$$\theta = (\Psi_+) - (\Psi_-)$$

(B3)

The values of $(\Psi_+)$ and $(\Psi_-)$ characteristics downstream of point $P$ are the same as at point $P$ because no additional turning of the flow occurs downstream of $C_1P$.

Every point in a supersonic flow is crossed by two Mach lines making the Mach angle $\beta$ with the flow direction. Because the characteristics are Mach lines numbered according to the convention just established, to every point in the supersonic flow a $(\Psi_+)$ and a $(\Psi_-)$ characteristic correspond. If the values of $(\Psi_+)$ and $(\Psi_-)$ are known at a point in the flow, the Mach number and the direction are given by equations (B2) and (B3).

The value of $(\Psi_+)$ assigned to a characteristic is unaltered by its intersection with the characteristics of the $(\Psi_-)$ set or vice versa. Two characteristics of the $(\Psi_+)$ set are shown intersecting the two characteristics of the $(\Psi_-)$ set in figure 18 (d). Three parallel streamlines, which may be considered to be elements of a supersonic stream tube are flowing across the characteristics. Streamline 1 is first given a counterclockwise deviation in flow path equal to $(\Psi_+) = (\Psi_+) = \lambda \tan^{-1} \sqrt{M_+^2 - 1}$ in crossing the $(\Psi_+)$ set of characteristics. It continues in a straight line until it intersects the set of $(\Psi_-)$ characteristics, which give it a clockwise deviation in flow path equal to $(\Psi_-) = (\Psi_-)$. The net deflection in path in the counterclockwise direction is $[(\Psi_+)- (\Psi_+)] = [(\Psi_-) - (\Psi_-)]$. Streamline 3 intercepts the $(\Psi_-)$ set of characteristics first and is deflected in a clockwise direction by an amount $(\Psi_-) = (\Psi_-)$. It continues in a straight line until it intersects the $(\Psi_+)$ set of characteristics, which deflect it in a counterclockwise direction an amount $(\Psi_+) = (\Psi_+)$. The net deflection of streamline 3 in the counterclockwise direction is $[(\Psi_+)- (\Psi_+)] - [(\Psi_-) - (\Psi_-)]$, the same as for streamline 1. The total turning $\Delta \Psi$ experienced by both streamlines 1 and 3 in crossing both sets of characteristics is the same and is equal to $[(\Psi_+)- (\Psi_+)] = [(\Psi_-) - (\Psi_-)]$ if streamlines 1 and 3 had the same Mach number and flow direction before intercepting the $(\Psi_+)$ and $(\Psi_-)$ set of characteristics, they would have the same new Mach number and new flow direction after crossing the characteristics. The stream-tube width has also increased to a value corresponding to the higher Mach number of flow after crossing the characteristics. Streamline 2 crosses both sets of characteristics simultaneously. Each set of characteristics produces its turning of the flow independently of the other. The streamline assumes the resultant direction due to the simultaneous clockwise and counterclockwise turning of the flow. The final flow direction and Mach number at $P$ is the same as for streamlines 1 and 3 after passing through both sets of characteristics.

**CHARACTERISTICS INCIDENT ON CHANNEL WALL**

Only flows that do not separate from the confining channel walls are considered in this report. Consider two characteristics of the $(\Psi_+)$ set, having values $(\Psi_+) = (\Psi_+)$, incident on the straight channel wall shown in figure 18 (c). The streamlines move along the wall instead of following the dotted path under the influence of expansion waves contained between the $(\Psi_+)$ characteristics because an expansion wave belonging to the $(\Psi_+)$ set arises at the wall between points $A$ and $B$ that cancels the tendency of the flow to deviate from the wall. That is,

$$\Delta (\Psi_-) = (\Psi_-) - (\Psi_-) = (\Psi_+) - (\Psi_+)_e$$

(B4)
If, between points A and B, the wall curves (as in fig. 18 (f)) an amount \( \Delta \alpha \), then the expansion wave of the \((\Psi_+)^{\prime}\) set must exceed that of the incident \((\Psi_+)^{\prime}\) set by an amount \( \Delta \alpha \) or

\[
\Delta(\Psi_+)^{\prime} = (\Psi_+)^{\prime}_A - (\Psi_+)^{\prime}_B = \Delta(\Psi_+) + \Delta \alpha = (\Psi_+^{\prime})_A - (\Psi_+^{\prime})_B \div \Delta \alpha
\]  

(B5)

If between points A and B (fig. 18 (g)) the wall curves in the direction of the streamline along the wall under the influence of the wave of the \((\Psi_+)\) set only, then the flow adjacent to the wall follows the wall without requiring the compensating expansion wave of the \((\Psi_+)\) set. In this case no wave of the \((\Psi_+)\) set is generated. The fact that waves emanating from a channel wall can be suppressed by curving the wall to the shape of the adjacent streamline under the influence of the incident expansion waves represents the basis of the method for designing supersonic nozzles used in this report.

The characteristics arising at a wall about which a two-dimensional flow is turned are straight as long as the flow responds to waves from only one wall. The deviation of the flow produced by an intersecting system of waves results in curved characteristics because the characteristics must make the Mach angle \( \beta \) everywhere with the flow direction.
APPENDIX C
DERIVATION OF EXPRESSION FOR MAXIMUM INITIAL EXPANSION ANGLE

In the discussion in appendix B of the expansive turning of a supersonic flow about a continuously curved wall (fig. 18 (b)), characteristics having a finite difference in turning angle were shown to have a finite distance of separation at the wall. If \( D \) be a displacement in the direction of the flow at the wall then

\[
\frac{dD}{dM} > 0 \quad (C1)
\]

In the limiting case of a sharp expansive turn (finite angle) at the wall, all characteristics in the corresponding wedge-shaped expansion wave originate at the sharp corner (fig. 18(a)). The flow adjacent to the wall undergoes an abrupt finite increase in Mach number in crossing the wedge-shaped expansion wave at its vertex where the wave width \( dD \) in the direction of the flow is vanishingly small. In this case

\[
\frac{dD}{dM} = 0 \quad (C2)
\]

The condition expressed by equation (C2) represents a limiting value of \( \frac{dD}{dM} \) because no expansive turn in a wall will give negative values for \( \frac{dD}{dM} \) in the absence of waves incident upon the walls.

In the expansion part of the nozzles considered, no waves are incident upon the nozzle walls. Therefore the condition that \( \frac{dD}{dM} \geq 0 \) applies.

When a value of \( \Psi_f \) or \( M_f \) is chosen too low for the maximum wall-expansion angle \( \alpha_k \) employed, then \( \frac{dX}{dM} \) becomes negative in the neighborhood of section 1-1' where \( \theta = 0 \). For the limiting condition

\[
\frac{dX}{dM} = 0 \quad (C3)
\]

where \( X \) is the coordinate of the wall parallel to the nozzle axis (direction of flow at section 1-1' where \( \alpha = 0 \)).

In order to obtain the allowable values of \( \Psi_f \) and \( \alpha_k \), as governed by equation (C3), the expression for \( X \) must be differentiated with respect to \( M \) and set equal to 0 at section 1-1', where \( \theta = 0, M = M_f \), and \( r = r_f \).

The expression for \( X \) for the expansion part of the nozzle is given by equation (13)

\[
X = r \cos \theta - Mr(\alpha_k - \theta) \cos (\beta - \theta) \quad (C4)
\]

and

\[
\frac{dX}{dM} = \frac{\partial X}{\partial r} dr + \frac{\partial X}{\partial \theta} d\theta + \frac{\partial X}{\partial \beta} d\beta + \frac{\partial X}{\partial M} dM = 0 \quad (C5)
\]

From the values of the parameters at section 1-1', the terms in equation (C5) are obtained: From equation (1), with \( \gamma = 1.40 \),

\[
r = \frac{r_f}{M} \left( \frac{5 + M_f^2}{6} \right)^{\frac{3}{2}} \quad (C6)
\]

Substituting for \( r_f \) the expression preceding equation (C6) yields for section 1-1'

\[
\frac{dr}{dM} = \frac{r_f}{M_f} \left( \frac{5 + M_f^2}{6} \right) \quad (C7)
\]

and

\[
\frac{\partial X}{\partial r} = \cos \theta - M(\alpha_k - \theta) \cos (\beta - \theta) \quad (C8)
\]

Because \( \theta = 0 \) and \( M = M_f \),

\[
\frac{\partial X}{\partial r} = 1 - M\alpha_k \cos \beta \quad (C8)
\]

From equation (11d), with \( \gamma = 1.40 \),

\[
\theta = \Psi - \Psi_f = \lambda \tan^{-1} \frac{\sqrt{M_f^2 - 1}}{\lambda} - \tan^{-1} \frac{\sqrt{M_f^2 - 1} - \Psi_f}{\lambda} \quad (C9)
\]

and

\[
\frac{\partial X}{\partial \theta} = -\frac{r}{M_f \sqrt{M_f^2 - 1} - \alpha_k} \sin (\beta - \theta) \quad (C9)
\]

Therefore, for \( \theta = 0 \)

\[
\frac{\partial X}{\partial \theta} = \frac{r_f}{M_f \sqrt{M_f^2 - 1} - \alpha_k} \quad (C10)
\]

By definition

\[
\beta = \sin^{-1} \frac{1}{M} \quad (C11)
\]

\[
\frac{d\beta}{dM} = -\frac{1}{M_f \sqrt{M_f^2 - 1}} \quad (C11)
\]
From equation (C4)
\[ \frac{\partial X}{\partial \theta} = M_r(\alpha_k - \theta) \sin (\beta - \theta) \]
which becomes at section 1-1:

\[ \frac{\partial X}{\partial \theta} = -r(\alpha_k - \theta) \cos (\beta - \theta) \]

which gives, for \( \theta = 0 \)

\[ \frac{\partial X}{\partial M} = -r(\alpha_k - \theta) \cos (\beta - \theta) \]

\[ \frac{\partial M}{\partial \theta} = -r(\alpha_k - \theta) \cos (\beta - \theta) \]

\[ \frac{\partial M}{\partial M} = -r(\alpha_k - \theta) \cos (\beta - \theta) \]

Solving for \( \alpha \) and \( \theta \) from equations (C7) to (C13) in equation (C5) and

\[ \alpha = \frac{(M_f - 1)}{2} \]

\[ \theta = 0.6M_1 \]

REFERENCES


TABLE 1 VALUES OF \( \alpha \), \( \beta \), \( r/M \), AND \( dM/dh \) FOR FIXED INTERVALS OF \( M \)

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<th>( M_f )</th>
<th>( \alpha )</th>
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<th>( dM/dh )</th>
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\begin{align*}
\alpha & = \frac{(M_f - 1)}{2} \\
\theta & = 0.6M_1
\end{align*}
## TABLE I — VALUES OF $\beta$, $\phi$, $r/r_1$, AND $d/d_1$ FOR FIXED INTERVALS OF $M$—Concluded

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## TABLE II — VALUES OF $M$, $\beta$, $\phi$, $r/r_1$, $d/d_1$ FOR FIXED INTERVALS OF $\phi$

[Values obtained by interpolation from Table I]

<p>| | | | |</p>
<table>
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<td>2162.000</td>
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### Footnotes

21. Equations for the Design of Two-Dimensional Supersonic Nozzles

32. Values obtained by interpolation from Table I
### Table III: Sample Design of Two-Dimensional Supersonic Nozzles for Final Mach Number \( M_f \) of 3.50 and Final Nozzle Width of 10 Inches

**[Symbols defined in appendix A]**

(a) Design parameters

<table>
<thead>
<tr>
<th>Equation</th>
<th>Equation number</th>
<th>Shortest nozzle ¹</th>
<th>Nozzle with straight-walled part ²</th>
</tr>
</thead>
<tbody>
<tr>
<td>( A_1 )</td>
<td>Table 4, ( M_f = 3.50 )</td>
<td>( \frac{1}{3} \times \frac{1}{2} )</td>
<td>( \frac{1}{3} \times \frac{1}{2} )</td>
</tr>
<tr>
<td>( A_1 )</td>
<td>Table 4, ( M_f = 3.50 )</td>
<td>( 1.7388 )</td>
<td>( 1.4728 )</td>
</tr>
<tr>
<td>( d_t )</td>
<td>Table 3, ( M_f = 3.50 )</td>
<td>( \frac{1}{3} \times \frac{1}{2} )</td>
<td>( \frac{1}{3} \times \frac{1}{2} )</td>
</tr>
<tr>
<td>( \Phi_f )</td>
<td>(b) ( M = 3.50 )</td>
<td>( 9.8^\circ )</td>
<td>( 8.39^\circ )</td>
</tr>
<tr>
<td>( \alpha )</td>
<td>Table 2, ( \beta = 10.0^\circ )</td>
<td>( 20.0^\circ )</td>
<td>( 20.0^\circ )</td>
</tr>
<tr>
<td>( M_s )</td>
<td>Table 2, ( \beta = 10.0^\circ )</td>
<td>( 2.429 )</td>
<td>( \Phi_f = 1.222 )</td>
</tr>
<tr>
<td>( \rho )</td>
<td>Table 3, ( \beta = 34.265^\circ )</td>
<td>( 1.222 )</td>
<td>( 2.8428 )</td>
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</tbody>
</table>

(b) Typical coordinates of expansion part (\( M = 1.600 \))

<table>
<thead>
<tr>
<th>Equation</th>
<th>Equation number</th>
<th>Shortest nozzle ¹</th>
<th>Nozzle with straight-walled part ²</th>
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</thead>
<tbody>
<tr>
<td>( M )</td>
<td>Table 1, ( \beta = 1.60 )</td>
<td>( 1.600 )</td>
<td>( 1.222 )</td>
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<tr>
<td>( \rho )</td>
<td>Table 1, ( \beta = 1.60 )</td>
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<td>( \Phi_f = 1.775 )</td>
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<td>( r )</td>
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<td>Table 4, ( M = 1.60 )</td>
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<tr>
<td>( x )</td>
<td>Table 4, ( M = 1.60 )</td>
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<tr>
<td>( \gamma )</td>
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<td>( 3.1666 )</td>
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(c) Length of straight-walled part (equation (20))

<table>
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<td>( r_s )</td>
<td>Table 1, ( M = 3.50 )</td>
</tr>
<tr>
<td>( r_g )</td>
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</tr>
<tr>
<td>( A_s )</td>
<td>Table 1, ( M = 3.50 )</td>
</tr>
<tr>
<td>( A_g )</td>
<td>Table 1, ( M = 3.50 )</td>
</tr>
</tbody>
</table>

¹ No straight-walled part; initial expansion accomplished by 1 turn about sharp corner.
² Straight-walled part with \( \alpha = 1.500^\circ \); initial expansion accomplished by 2 turns in succession about sharp corner at each wall.
EQUATIONS FOR THE DESIGN OF TWO-DIMENSIONAL SUPERSONIC NOZZLES

TABLE III—SAMPLE DESIGN OF TWO-DIMENSIONAL SUPERSONIC NOZZLES FOR FINAL MACH NUMBER $M_f$ OF 3.50 AND FINAL NOZZLE WIDTH OF 10 INCHES—Continued

<table>
<thead>
<tr>
<th>Equation</th>
<th>Equation number</th>
<th>Shortest nozzle 1</th>
<th>Nozzle with straight-walled part 2</th>
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<tr>
<td>$M$</td>
<td>$M_1 &lt; M &lt; M_f$</td>
<td>Source of computed value</td>
<td>Value</td>
</tr>
<tr>
<td>$\theta$</td>
<td>1, col. 3, $M = 2.80$</td>
<td>$M_s = M_1 = 2.260$ if $M &lt; 1.50$ (Value chosen)</td>
<td>2.800</td>
</tr>
<tr>
<td>$\theta$</td>
<td>1, col. 3, $M = 2.80$</td>
<td>$1.650$</td>
<td>1.650</td>
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<tr>
<td>$\psi$</td>
<td>1, col. 3, $M = 2.80$</td>
<td>$56.50^\circ - 65.74^\circ$</td>
<td>12.74^\circ</td>
</tr>
<tr>
<td>$\gamma$</td>
<td>1, col. 4, $M = 2.80$</td>
<td>$3.500$</td>
<td>3.500</td>
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<tr>
<td>$\rho$</td>
<td>1, col. 4, $M = 2.80$</td>
<td>$3.500 \times 1.768$</td>
<td>6.060 in.</td>
</tr>
<tr>
<td>$\alpha - \theta$</td>
<td>$X = \cos \theta + M_s (\alpha - \theta) \cos (\beta + \theta)$</td>
<td>$24.26^\circ - 12.74^\circ$</td>
<td>$11.05^\circ$</td>
</tr>
<tr>
<td>$\beta + \theta$</td>
<td>$Y = \sin \theta + M_s (\alpha - \theta) \sin (\beta + \theta)$</td>
<td>$29.93^\circ + 12.74^\circ$</td>
<td>$33.52^\circ$</td>
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<tr>
<td>$\psi / 2$</td>
<td>$d_1 = 1.22$</td>
<td>$6.060 \times \cos 12.74^\circ + 2.800 \times 0.080 \times 0.200$</td>
<td>$8.725$ in.</td>
</tr>
<tr>
<td>$\psi / 2$</td>
<td>$d_2 = 1.22$</td>
<td>$2.260 \times \sin 12.74^\circ + 2.800 \times 0.080 \times 0.200$</td>
<td>$3.241$ in.</td>
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... (Continued) ...

(e) Typical coordinates of initial expansion part

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<th>Equation</th>
<th>Equation number</th>
<th>Shortest nozzle with single initial turn: $\psi_1 = 10.000^\circ$</th>
<th>Nozzle with straight-walled part and double initial turn: $\psi_1 = 5.000^\circ$</th>
</tr>
</thead>
<tbody>
<tr>
<td>$\psi_1$</td>
<td>$\psi_1$</td>
<td>Source of computed value</td>
<td>Value</td>
</tr>
<tr>
<td>$M$</td>
<td>$M_1 \leq M \leq M_f$</td>
<td>Source of computed value</td>
<td>Value</td>
</tr>
<tr>
<td>$\theta$</td>
<td>1, col. 2, $M = 1.31$</td>
<td>$5.000^\circ$</td>
<td>$2.500^\circ$</td>
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<tr>
<td>$\psi$</td>
<td>1, col. 2, $M = 1.31$</td>
<td>$1.1555$</td>
<td>$1.1555$</td>
</tr>
<tr>
<td>$\psi$</td>
<td>1, col. 3, $M = 1.31$</td>
<td>$5.755^\circ$</td>
<td>$5.755^\circ$</td>
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<tr>
<td>$\psi$</td>
<td>1, col. 3, $M = 1.31$</td>
<td>$5.755^\circ$</td>
<td>$1.1555$</td>
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<tr>
<td>$d_1$</td>
<td>1, col. 5, $M = 1.31$</td>
<td>$1.2600 \times 1.4728$</td>
<td>$1.0002$ in.</td>
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<tr>
<td>$X_1$</td>
<td>$d_1 \cos (\beta + \psi_1 - \psi)$</td>
<td>$1.0002 \times \cos 59.18^\circ$</td>
<td>$0.956$ in.</td>
</tr>
<tr>
<td>$Y_1$</td>
<td>$d_1 \sin (\beta + \psi_1 - \psi)$</td>
<td>$1.0002 \times \sin 59.18^\circ$</td>
<td>$1.656$ in.</td>
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<tr>
<td>$\alpha - \theta$</td>
<td>$X_2 = \cos \theta + M_s (\alpha - \theta) \cos (\beta + \theta)$</td>
<td>$1.2600 \times 1.4728$</td>
<td>$1.1574 \times 1.4728$</td>
</tr>
<tr>
<td>$\beta + \theta$</td>
<td>$Y_2 = \sin \theta + M_s (\alpha - \theta) \sin (\beta + \theta)$</td>
<td>$2.260 \times \sin 59.18^\circ$</td>
<td>$1.7046 \times \sin 59.18^\circ$</td>
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Second turn

<table>
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<th>Shortest nozzle with single initial turn: $\psi_1 = 10.000^\circ$</th>
<th>Nozzle with straight-walled part and double initial turn: $\psi_1 = 5.000^\circ$</th>
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<tbody>
<tr>
<td>$\psi_1$</td>
<td>$\psi_1$</td>
<td>Source of computed value</td>
<td>Value</td>
</tr>
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<td>$M_1 \leq M \leq M_f$</td>
<td>Source of computed value</td>
<td>Value</td>
</tr>
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<td>$55.852^\circ$</td>
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<tr>
<td>$\psi$</td>
<td>1, col. 3, $M = 1.31$</td>
<td>$4.955^\circ$</td>
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<td>$d_1$</td>
<td>1, col. 5, $M = 1.31$</td>
<td>$55.852^\circ$</td>
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<tr>
<td>$X_2$</td>
<td>$d_2 \cos (\beta + \psi_1 - \psi)$</td>
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<tr>
<td>$Y_2$</td>
<td>$d_2 \sin (\beta + \psi_1 - \psi)$</td>
<td>$1.8625$ in.</td>
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</table>

1 No straight-walled part; initial expansion accomplished by 1 turn about sharp corner.
2 Straight-walled part with $\alpha = 15.000^\circ$; initial expansion accomplished by 2 turns in succession about sharp corner at each wall.
### TABLE III—SAMPLE DESIGN OF TWO-DIMENSIONAL SUPersonic Nozzles for Final Mach Number $M_f$ of 3.50 and Final Nozzle Width of 10 inches—Concluded

<table>
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<th>Equation</th>
<th>Source of computed value</th>
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<td>$r_i$</td>
<td>1, col. 2, $M_f=3.50$</td>
<td>10.002&quot;</td>
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<tr>
<td>$r_i$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>6.7806&quot;</td>
</tr>
<tr>
<td>$r_i$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>6.7806 x 2.8128&quot;</td>
</tr>
<tr>
<td>$r_i$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>9.0088 in.</td>
</tr>
<tr>
<td>$r_i$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>1.3350 in.</td>
</tr>
<tr>
<td>$X_f$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>1.4801 x 2.8128&quot;</td>
</tr>
<tr>
<td>$X_f$</td>
<td>1, col. 4, $M_f=3.50$</td>
<td>2.9529 in.</td>
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</table>

<table>
<thead>
<tr>
<th>Equation</th>
<th>Source of computed value</th>
<th>Value</th>
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</thead>
<tbody>
<tr>
<td>$\beta_i - \psi_1$</td>
<td>$\frac{d_0 \cot (\beta_i - \psi_1)}{4}$</td>
<td>24c</td>
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<tr>
<td>$\beta_i - \psi_1$</td>
<td>44.180 - 10.000 in.</td>
<td></td>
</tr>
<tr>
<td>$b_0$</td>
<td>$\frac{d_0 \cot (\beta_i - \psi_1)}{4}$</td>
<td>34.180 in.</td>
</tr>
<tr>
<td>$b_0$</td>
<td>1.4728 cot 34.180</td>
<td></td>
</tr>
<tr>
<td>$\delta_0$</td>
<td>$\frac{d_0 \cot (\beta_i - \psi_1)}{4}$</td>
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</tr>
<tr>
<td>$\theta_2$</td>
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</tr>
<tr>
<td>$\theta_2$</td>
<td>1.543</td>
<td></td>
</tr>
<tr>
<td>$\theta_2$</td>
<td>$\frac{d_0 \cot (\beta_i - \psi_1)}{4}$</td>
<td>2.114</td>
</tr>
<tr>
<td>$l_1$</td>
<td>$\frac{d_0 \cot (\beta_i - \psi_1)}{4}$</td>
<td>35.986 - 2.169</td>
</tr>
<tr>
<td>$l_1$</td>
<td>2.941 + 2.114</td>
<td></td>
</tr>
<tr>
<td>$l_1$</td>
<td>35.949</td>
<td></td>
</tr>
</tbody>
</table>

1 No straight-walled part; initial expansion accomplished by 1 turn about sharp corner.
2 Straight-walled part with $\alpha$ of 17.000°; initial expansion accomplished by 2 turns in succession about sharp corner at each wall.
Positive directions of axes and angles (forces and moments) are shown by arrows.

<table>
<thead>
<tr>
<th>Axis</th>
<th>Designation</th>
<th>Symbol</th>
<th>Force (parallel to axis) symbol</th>
<th>Moment about axis</th>
<th>Angle</th>
<th>Velocities</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>Longitudinal</td>
<td>X</td>
<td>X</td>
<td>Rolling</td>
<td>Roll</td>
<td>Linear</td>
</tr>
<tr>
<td></td>
<td>Lateral</td>
<td>Y</td>
<td>Y</td>
<td>Pitching</td>
<td>Pitch</td>
<td>Angular</td>
</tr>
<tr>
<td></td>
<td>Normal</td>
<td>Z</td>
<td>Z</td>
<td>Yawing</td>
<td>Yaw</td>
<td></td>
</tr>
</tbody>
</table>

<table>
<thead>
<tr>
<th>Designation</th>
<th>Symbol</th>
<th>Linear (component along axis)</th>
<th>Angular</th>
</tr>
</thead>
<tbody>
<tr>
<td>Longitudinal</td>
<td>X</td>
<td>u</td>
<td>p</td>
</tr>
<tr>
<td>Lateral</td>
<td>Y</td>
<td>v</td>
<td>q</td>
</tr>
<tr>
<td>Normal</td>
<td>Z</td>
<td>w</td>
<td>r</td>
</tr>
</tbody>
</table>

Absolute coefficients of moment

\[
C_1 = \frac{L}{q b S} \quad C_n = \frac{M}{q c S} \quad C_s = \frac{N}{q b S}
\]

Angled of set of control surface (relative to neutral position), \( \delta \). (Indicate surface by proper subscript.)

4. PROPELLER SYMBOLS

- **D**: Diameter
- **p**: Geometric pitch
- **p/D**: Pitch ratio
- **V'**: Inflow velocity
- **V_s**: Slipstream velocity
- **T**: Thrust, absolute coefficient \( C_T = \frac{T}{\rho n^2 D^3} \)
- **Q**: Torque, absolute coefficient \( C_Q = \frac{Q}{\rho n^3 D^5} \)

\[
P \quad \text{Power, absolute coefficient } C_p = \frac{P}{\rho n^3 D^5}
\]

\[
C_s \quad \text{Speed-power coefficient } = \frac{V'^3}{P n^3}
\]

\[
\eta \quad \text{Efficiency}
\]

\[
n \quad \text{Revolutions per second, rps}
\]

\[
\phi \quad \text{Effective helix angle } = \tan^{-1} \left( \frac{V}{2 \pi r n} \right)
\]

5. NUMERICAL RELATIONS

- 1 hp = 76.04 kg-m/s = 550 ft-lb/sec
- 1 metric horsepower = 0.9863 hp
- 1 mph = 0.4470 mps
- 1 mps = 2.2369 mph

- 1 lb = 0.4536 kg
- 1 kg = 2.2046 lb
- 1 mi = 1,609.35 m = 5,280 ft
- 1 m = 3.2808 ft