Automatic Differentiation for Design Sensitivity Analysis of Structural Systems Using Multiple Processors

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Automatic differentiation tools (ADIFOR) is incorporated into a finite element based structural analysis program for shape and non-shape design sensitivity analysis of structural systems. The entire analysis and sensitivity procedures are parallelized and vectorized for high performance computation. Small-scale examples to verify the accuracy of the proposed program and a medium-scale example to demonstrate the parallel-vector performance on the multiple Cray-C90- processors are included in the paper.

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by

Duc T. Nguyen, Olaf O. Storaasli, Jiangning Qin, and Ramzi Qamar

Abstract

Automatic differentiation tools (ADIFOR) is incorporated into a finite element based structural analysis program for shape and non-shape design sensitivity analysis of structural systems. The entire analysis and sensitivity procedures are parallelized and vectorized for high-performance computation. Small-scale examples to verify the accuracy of the proposed program and a medium-scale example to demonstrate the parallel-vector performance on the multiple Cray-C90 processors are included in the paper.

I. Introduction

Using the familiar finite element procedure, the static equilibrium equations for a structural model can be expressed as

\[
\begin{bmatrix} K(b) \end{bmatrix}_{n \times n} \{ z \}_{n \times 1} = \{ F \}_{n \times 1}
\]

where \([K(b)]\), \([z]\) and \([F]\) are referred to the stiffness matrix, nodal displacement vector and nodal force vector, respectively. In Eq. (1), "n" represents the active degree-of-freedom of the discretized structural model.

The stiffness matrix \([K(b)]\), in general, is a function of design variable vector \([b]\) (where \([b] \in \mathbb{R}^k\)). As an example, \([b]\) may represent the cross-sectional areas of various truss members, or thickness of plate members (for non-shape type of design variables), or it may also represent the joint coordinates of various nodes of a structure (for shape type of design variables).

A typical constraint, involving a limit on a displacement or a stress component, may be written as

\[
g(z, b) \leq 0
\]

For the sake of simplified notation, it is assumed that \(g\) depends on only a simple design variable \(b\) (i.e. \([b] \in \mathbb{R}^{k_x}\)). Using the chain rule of differentiation, one obtains

\[
\frac{d g}{d b} = \frac{\partial g}{\partial b} + x^T \frac{d z}{d b}
\]
where $x$ is a vector with components

$$x_j = \frac{\partial g}{\partial z_j}$$

The first term on the right-hand-side of Eq. (3) is usually zero or easy to obtain, thus one discusses only the computation of the second term.

Differentiating Eq. (1) with respect to $b$, one obtains

$$K \cdot \frac{dz}{db} = \frac{\partial F}{\partial b} - \frac{dK}{db} \cdot z$$

Premultiplying Eq. (5) by $x^T K^{-1}$, one obtains

$$x^T \frac{dz}{db} = x^T K^{-1} \left( \frac{\partial F}{\partial b} - \frac{dK}{db} \cdot z \right)$$

Numerically, the computation of $x^T \frac{dz}{db}$ can be performed in two different ways. The first, called the "direct method", consists of solving Eq. (5) for $\frac{dz}{db}$ and then taking the scalar product with $x$. The second approach, called the "adjoint method"[2, 3], defines an adjoint vector $\lambda$ which is the solution of the system

$$K \lambda = x$$

or

$$\lambda = K^{-1} x$$

or

$$\lambda^T = x^T K^{-1} \quad \text{(since matrix } K \text{ is symmetric)}$$

and thus, Eq. (3) can be re-written as

$$\frac{dg}{db} = \frac{\partial g}{\partial b} + \lambda^T \left( \frac{\partial F}{\partial b} - \frac{dK}{db} \cdot z \right)$$

The solution of Eq. (7) for $\lambda$ is similar to a solution for displacement under a "dummy" load vector $\{ x \}$.

Once, the sensitivity information $\frac{dg}{db}$ has been computed, any gradient based optimization softwares[4, 5] can be used to obtain a new, improved design.

The focus of this paper is in the parallel computation of $\frac{dK}{db}$ as shown in Eq. (5), and particularly, the computation of the term $\frac{dK}{db}$.

Since in the finite element procedure

$$[ K ] = \sum_{e=1}^{\# \text{ elements}} [ k(e) ]$$

Therefore, computation of $\frac{d[K]}{db}$ involves with computation of $\frac{d[k]}{db}$ and the latter can be obtained either by

(i) Finite Difference Method
or

(ii) Analytical Method

In the finite difference method, a small perturbation of a design variable is first applied, then approximate derivative (which can be affected by round-off and truncation errors\(^{[3]}\)) can be generated. The analytical method tends to generate very cumbersome expressions for the derivatives. Thus, the objectives of this paper is to use automatic differentiation (ADIFOR) tools\(^{[6]}\) to compute the derivatives of \(\frac{d[k_r]}{db} \) in a parallel-vector computer environment.

A brief review of ADIFOR tools\(^{[6]}\) is given in Section 2. Parallel generation and assembly\(^{[7]}\) of the stiffness matrix \([K]\) is presented in Section 3. Parallel-Vector equation solver\(^{[8]}\) which will be used to solve system of Eq. (5) is summarized in Section 4. Numerical examples are presented in Section 5, and conclusions are drawn in Section 6.

II. A Brief Review on Automatic Differentiation\(^{[6]}\)

Automatic Differentiation (AD) is essentially an automatic implementation of the chain rule of differentiation based on tracking the connection between the dependent (or output) and independent (or input) variables.

Typically, to calculate the derivative of any output variable in a computer program with respect to any input variable, one modifies the original program by inserting of specialized instruction which identify the relevant output and input variables.

Automatic differentiation produces exact derivatives, limited only by machine precision. There are two modes of AD. In the forward mode, the chain rule is evaluated from the input to the output. In this mode, the computational cost increases with the number of input variables. In the reverse mode, the chain rule is evaluated from the output to the input.

In order to understand the forward mode in AD, let’s refer to Figure 1 where the computation flow to evaluate

\[
y_3 = \frac{-20b_2}{(2b_1b_2 + \sqrt{2b_1b_2})} = \frac{-20b_2}{b_1(2b_2 + \sqrt{2b_1})}
\]

is shown in a form of the directed graph.

The derivatives of \(\frac{dy_3}{db_2}\) and \(\frac{dy_3}{db_1}\) are also shown in a form of the directed graph in Figure 2.

In Figure 2, the connecting link between any 2 vertex represents the chain-rule derivatives. As an example, \(\frac{\partial a}{\partial b_2} = 2b_1\) and \(\frac{\partial d}{\partial a} = 1\).

On the other hand, if the reverse mode of differentiation is used to calculate \(\frac{dy_3}{db_1}\), then the chain-rule of differentiation will start with the output variable \(y_3\), and then proceed as following:
\[
\frac{dy_3}{dd} = \frac{20b_2}{(2b_1b_2 + \sqrt{2}b_1^2)^2}
\]

\[
\frac{dy_3}{da} = \frac{\partial d}{\partial a} = \frac{20b_2 \times 1}{(2b_1b_2 + \sqrt{2}b_1^2)^2}
\]

\[
\frac{dy_3}{dx_2} = \frac{dy_3}{da} \frac{\partial a}{\partial b_2} + \frac{\partial y_3}{\partial b_2} = \frac{20b_2 \times 1 \times 2b_1}{(2b_1b_2 + \sqrt{2}b_1^2)^2} + \frac{-20}{(2b_1b_2 + \sqrt{2}b_1^2)}
\]

It has been concluded from earlier research works\(^{[6,9,10]}\) that using automatic differentiation (AD) method, such as ADIFOR tool\(^{[6]}\), will be more computationally efficient than the finite difference method. In most problems, however, analytical method is more efficient than ADIFOR tool (but at the expense of assuming there is no human errors in deriving analytical derivative expressions).

The comparisons of computational costs and the accuracy to evaluate derivative information between the Finite Difference, Analytical and ADIFOR have been discussed\(^{[6,9,10]}\). This paper, therefore, will focus on the issue of incorporating derivative calculation subroutines (generated by ADIFOR) in a parallel-vector high-performance computer environment.

### III. Parallel Generation and Assembly on Distributed- and Shared Memory Computers\(^[7]\)

The choice of the storage scheme for the global stiffness matrix in any finite element analysis code is based on whether it will save the memory or it will enhance the vector speed, or both. The row-oriented storage scheme\(^{[8]}\) is good for saxpy operation and shared memory type computers, while the skyline storage is good for dot product (daxpy) operation. Moreover, the skyline storage scheme requires less memory and this feature is important for computers with distributed-memory (since each processor usually has less memory capacity as compared to shared-memory computers). Fortunately, the Intel iPSC/860 computers have good vector performance for daxpy operation. In order to use the vector-unrolling technique to improve the vector performance, a block-skyline columns storage and block rows storage schemes for the stiffness matrix is used on the Intel and Cray type computers, respectively (as shown in Figure 3). To simplify the discussion, assuming the global matrix is full and three processors are used to store different portions of the global stiffness matrix.

The size of the block is called k if there are k columns (or k-rows) in each block. It is realized that the choice of k will have the effects on

1. the in-core memory requirement,
2. the vector performance,
3. the communication performance.

For the Intel iPSC/860 parallel computers, the block size in MPFEA is set to be 8. Since each processor only has certain block-columns (or block rows) of the global stiffness matrix, the generation and assembly of this matrix can be done in parallel without any communications among processors. The work involved in the generation and assembly procedure can be summarized as (for each processor \(i\), where \(i = 1, 2, \ldots, NP\)): 185
Task 1. To identify (but not to search for!) the elements that contribute to the columns (or rows) which belong to processor i.

Task 2. To generate these elements stiffness matrices.

Task 3. To assemble the global stiffness matrix with these element stiffness matrices.

It should be noted here that even for the case of nonlinear structural analysis, Task 1 of the above procedure needs to be done only once, while Task 2 and Task 3 have to be performed repeatedly since the global matrix will be updated in each nonlinear iteration.

IV. Parallel-Vector Choleski Method Development

In the sequential Choleski method, a symmetric, positive-definite stiffness matrix, \([K]\), can be decomposed as

\[
[K] = [U]^T [U]
\]  

with the coefficients of the upper-triangular matrix, \([U]\):

\[
u_{ij} = 0 \quad \text{for} \quad i > j
\]

\[
u_{11} = \sqrt{K_{11}} \quad \text{for} \quad j \geq 1
\]

\[
u_{ij} = \sqrt{K_{ij} - \sum_{k=1}^{i-1} u_{ki}^2} \quad \text{for} \quad i > 1
\]

\[
u_{ij} = \frac{K_{ij} - \sum_{k=1}^{i-1} u_{ki} u_{kj}}{u_{ii}} \quad \text{for} \quad i, j > 1
\]

For example, \(u_{57}\) can be computed from Eq. (18) as:

\[
u_{57} = \frac{k_{57} - u_{15} u_{17} - u_{25} u_{27} - u_{35} u_{37} - u_{45} u_{47}}{u_{55}}
\]

The calculations in Eq. (17) for the term \(u_{57}\) (of row 5) only involve columns 5 and 7. Furthermore, the "final value" of \(u_{57}\) cannot be computed until the final, updated values of the first four rows have been completed. Assuming that only the first two rows of the factored matrix, \([U]\), have been completed, one still can compute the second partially-updated value of \(u_{57}\) as designated by superscript (2):

\[
u_{57}^{(2)} = k_{57} - u_{15} u_{17} - u_{25} u_{27}
\]

If row 3 has also been completely updated, then the third partially-updated value of \(u_{57}\) can be calculated as:

\[
u_{57}^{(3)} = \nu_{57}^{(2)} - u_{35} u_{37}
\]

This observation suggests an efficient way to perform Choleski factorization in parallel on NP
processors. For example, each row of the coefficient stiffness matrix, [K], is assigned to a separate processor.

From Eq. (17), assuming NP = 4, it is seen that row 5 cannot be completely updated until row 4 has been completely updated. In general, in order to update the i\textsuperscript{th} row, the previous (i-1) rows must already have been updated. For the above reasons, any NP consecutive rows of the coefficient stiffness matrix, [K], will be processed by NP separate processors. As a consequence, while row 5 is being processed by a particular processor, say processor 1, then the first (5-NP) rows have already been completely updated. Thus, if the i\textsuperscript{th} row is being processed by the p\textsuperscript{th} processor, there is no need to check every row (from row 1 to row i-1) to make sure they have been completed. It is safe to assume that the first (i-NP) rows have already been completed as shown in the triangular cross-hatched region of Figure 4.

Synchronization checks are required only for the rows between (i-NP + 1) and (i-1) as shown in the rectangular solid region of Figure 4. Since the first (i-NP) rows have already been completely factored, the i\textsuperscript{th} row can be "partially" processed by the p\textsuperscript{th} processor as shown in Eq. (18, 19).

V. Numerical Applications

Different finite element types (such as 2-D Truss, and Plate/Shell elements) and different type of design variables (such as cross-sectional areas, joint coordinates of truss elements and thickness of plate elements) are considered in this section. The first two examples are small-size for the purpose of verifying the accuracy of derivatives (\( \frac{d [k^{(x)}]}{d b} \)) generated by ADIFOR\textsuperscript{6} as compared to the ones obtained by finite difference technique. The last example is medium-size for the purpose of evaluating the parallel-vector performance of the entire finite element and Design Sensitivity Analysis (DSA) process.

Example 1: Plate-Structure With (Non-Shape) Thickness Design Variable

In this example, 32 plate elements\textsuperscript{11} are used, a point force is applied at the center of the fixed plate (see Figure 5). Thickness of a plate is selected as (non-shape) design variable in this case. The original thickness is 0.03 and a perturbation of 0.5% is used in the finite (central) difference scheme.

The derivatives of element stiffness matrix (in global reference and using ADIFOR) with respect to the thickness t for typical members such as members 5, 12, and 19 are presented in Table 1. These derivatives are in good agreement with the ones obtained by finite (central) difference scheme.

Example 2: Truss-Structure With (Shape) Joint Coordinate Design Variables

In this example, a 1 bay x 1 story truss structure is shown in Figure 6. This small-scale structure has 4 joints and 5 members. All joint x-coordinates of this structure are selected as (shape) design variables. A horizontal force F is applied at node 1. The dimensions for each base and height of this structure are 12" and 9", respectively. Young modulus and cross-sectional area are 29000 Ksi and 4 in\textsuperscript{2}, respectively. A perturbation of 1% is used in the finite (central) difference scheme. The derivatives of element stiffness matrix (in global reference and using ADIFOR) with respect to a typical x-coordinate of joint 2 for members 1 and 5 are presented in Table 2. Again, these derivatives are in good agreements with the ones obtained by finite (central)
difference scheme.

Example 3: A 2-D Truss Structure With 80 Bays and 190 Stories

In this example, a 80 bay x 190 story truss structure is also shown in Figure 6. A horizontal force F is applied at node 100. All other datas are the same as in Example 2. There are 96 cross-sectional areas selected as (non-shape) design variables in this example. This structure has 60,990 elements. The resulted structural stiffness matrix has 30,780 degree-of-freedom. Using the variable bandwidth storage scheme(8) will require a real 1-dimensional array with 5,171,574 words to store the stiffness matrix in the core memory. The average bandwidth for this stiffness matrix is 168.

The performance of the entire finite element analysis and design sensitivity analysis (using ADIFOR tool) on 1, 8, and 16 Cray-C90 processors are shown in Table 3. The total speed-up for the ENTIRE PROCESS are 7.32 and 12.93 when 8 and 16 Cray-C90 processors are used, respectively.

VI. Conclusions

Based upon the numerical results presented in this paper, the following conclusions can be made:

1. Automatic Differentiation (ADIFOR)[6] tool has been successfully applied to both simple (TRUSS) and complex PLATE/SHELL[11] finite elements.

2. Both non-shape and shape design variables can be successfully treated.

3. For the first time (to the authors' knowledge), ADIFOR tool can be applied in a parallel-vector computer environment for non-shape and shape sensitivity analysis.

4. The entire finite element and sensitivity analysis can be done with excellent parallel and vector speed (using all 16 Cray-C90 processors).

VII. Acknowledgments

The financial support from NASA grant NAG1-858 are acknowledged. The authors are also deeply indebted to Drs. L. Green, P. Newman, J. Barthelemy (all from NASA Langley Research Center), C. Bischof (from Argonne National Laboratory) and A. Carle (from Rice University) for helpful discussions during the ADIFOR user workshop (September 13-14, 1993), held at Building 1192C-E, the CFD Laboratory, NASA LaRC). Helpful discussions with Dr. A. Tessler on using his plate/shell element (NASA Langley Research Center) is also appreciated.

VIII. References


Figure 1: Computational Graph for \( y_3 = \frac{-20 b_2}{d} \)

\[ y_3 = \frac{-20 b_2}{d} \]

\[ d = a + b \]

\[ b = \sqrt{2} b_1^2 \]

\[ a = 2 b_1 b_2 \]

\[ b_1 \]

\[ b_2 \]

Figure 2: Computational Graph for \( \frac{dy_3}{db_2} = \frac{40 b_2 b_1}{(2b_1 b_2 + \sqrt{2} b_1^2)^2} + \frac{-20}{(2b_1 b_2 + \sqrt{2} b_1^2)} \)

\[ \frac{dy_3}{db_2} = \left( \frac{20 b_2}{d^2} \right) (1)(2b_1) + \left( \frac{-20}{d} \right) \]

\[ \frac{dy_3}{db_1} = \frac{(20 b_2) (2 b_2 + 2 \sqrt{2} b_1)}{[b_1 (2 b_2 + \sqrt{2} b_1)]^2} \]

\[ \frac{dy_3}{db_1} = \left( \frac{20 b_2}{d^2} \right) (1)(2b_2) + \left( \frac{20 b_2}{d^2} \right) (1)(2 \sqrt{2} b_1) \]
Figure 3. Block-skyline columns storage and block rows storage schemes

Figure 4: Information required to update row i
Figure 5: Clamped Plate - Structure

Figure 6: 2-D Truss Structure
\[
\frac{\partial [k^{(5)}]}{\partial t} = [5576.925, -4780.2198, 0, 0, 0, -15934.066, \ldots \ldots \ldots \ldots]
\]

\[
\frac{\partial [k^{(12)}]}{\partial t} = [15934.068, 5576.925, 0, 0, 0, -5576.923, \ldots \ldots \ldots \ldots]
\]

\[
\frac{\partial [k^{(19)}]}{\partial t} = [21510.99, 5576.925, 0, 0, 0, 8.268E-12, \ldots \ldots \ldots \ldots]
\]

Table 1: ADIFOR Derivatives of Plate Element Stiffness Matrix with Respect to Thickness (Non-shape) Design Variable
Table 2: ADIFOR Derivatives of Truss Element Stiffness Matrix with Respect to x-coordinate of Joint 2 (Shape) Design Variable.

el. stiff [k] for member 1
0.966667E+04 0.000000E+00 -0.966667E+04 0.000000E+00
0.000000E+00 0.000000E+00 0.000000E+00 0.000000E+00
-0.966667E+04 0.000000E+00 0.966667E+04 0.000000E+00
0.000000E+00 0.000000E+00 0.000000E+00 0.000000E+00

Gradient of stiff [k] w.r.t DV 2 = \frac{\partial \left[ k^{(1)} \right]}{\partial x_2}

\begin{align*}
-805.55555555556 & 0. & 0. & 805.55555555556 & 0. \\
0. & 0. & 0. & 0. & 0. \\
805.55555555556 & 0. & -805.55555555556 & 0. & 0. \\
0. & 0. & 0. & 0. & 0.
\end{align*}

el stiff [k] for member 5
0.494933E+04 0.371200E+04 -0.494933E+04 -0.371200E+04
0.371200E+04 0.278400E+04 -0.371200E+04 -0.278400E+04
-0.494933E+04 -0.371200E+04 0.494933E+04 0.371200E+04
-0.371200E+04 -0.278400E+04 0.371200E+04 0.278400E+04

Gradient of stiff [k] w.r.t DV 2 = \frac{\partial \left[ k^{(5)} \right]}{\partial x_2}

\begin{align*}
32.995555555555 & -284.58666666667 & -32.995555555555 & 284.58666666667 \\
-32.995555555555 & 284.58666666667 & 32.995555555555 & -284.58666666667 \\
\end{align*}
Table 3: Parallel-Vector Performance For DSA of 80 Bays x 190 Stories Truss Structure Using ADIFOR Tool on Multiple Cray-C90 Processors

<table>
<thead>
<tr>
<th>Tasks</th>
<th>Number of Cray-C90 Processors</th>
<th>Speed-Up Factors</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>1 proc.</td>
<td>8 proc.</td>
</tr>
<tr>
<td>(A)</td>
<td>0.4855 sec</td>
<td>0.09954 sec</td>
</tr>
<tr>
<td>(B)</td>
<td>0.9582 sec (0.9906*)</td>
<td>0.1320 sec (0.1433*)</td>
</tr>
<tr>
<td>(C)</td>
<td>2.6290 sec</td>
<td>0.3568 sec</td>
</tr>
<tr>
<td>(D)</td>
<td>0.1019 sec</td>
<td>0.1015 sec</td>
</tr>
<tr>
<td>(E)</td>
<td>2.3717 sec</td>
<td>0.3034 sec</td>
</tr>
<tr>
<td>(F)</td>
<td>9.6934 sec</td>
<td>1.2128 sec</td>
</tr>
<tr>
<td>Entire Process</td>
<td>16.2740 sec</td>
<td>2.2221 sec</td>
</tr>
</tbody>
</table>

Notes:
(A) To generate column heights of stiffness matrix
(B) To generate and assemble stiffness matrix
(C) To factorize stiffness matrix
(D) To get static (forward/backward) solution (sequential computation)
(E) To generate the right-hand-side vectors for sensitivity equations
(F) To solve for displacement sensitivity vectors
* Wall-Clock-Time
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OBJECTIVES

1. To obtain accurate derivatives of complex finite elements and/or complex design variables

2. Design variables can be either non-shape (such as areas, thickness) or shape types (such as joint coordinates)

3. The entire solution process should be parallelized and vectorized to reduce solution time

MOTIVATION

1. Analytical (hand-coded) derivatives are not feasible for complex and shape variables.

2. Finite difference derivatives are expensive and can be inaccurate.
APPROACH USED

1. Utilizing Automatic Differentiation (ADIFOR) tool developed by ANL & CRPC at Rice
   Application tool back from AAOB & ESB at LaRC

2. Parallelizing and Vectorizing every step of the entire solution process
UNIQUE FEATURES OF THIS WORK

1. Both simple and complex finite elements (2-D truss, and 3-D plate/shell) are treated.

2. Both (non-shape) design variables (such as areas of truss members, or thickness of plate and shell members), and shape design variables (such as joint coordinates) are considered.

3. The entire solution process has been parallelized and vectorized.

4. EXCELLENT speed-up has been achieved even on "small-scale" example.
GENERAL FORMULATION FOR DESIGN SENSITIVITY ANALYSIS (D.S.A.)

- Equilibrium Equations

\[
[K] \{Z\} = \{F\}
\]  \hspace{1cm} (1)

Take derivatives of both sides of Equation (1) with respect to design variable vector \( b \)

\[
\frac{\partial [K]}{\partial b} * \{Z\} + [K] * \frac{\partial \{Z\}}{\partial b} = [0]
\]  \hspace{1cm} (2)

- Sensitivity Equations

\[
[K] \frac{\partial \{Z\}}{\partial b} = - \frac{\partial [K]}{\partial b} * \{Z\}
\]  \hspace{1cm} (3)

Note:
Parallel-Vector Eq. Solver is needed for CSM & CFD!
D.S.A.
FOR 2-D TRUSS ELEMENT

\[
[k_{\text{Global}}] = \frac{EA}{L} \begin{bmatrix}
  C_x^2 & C_x C_y & -C_x^2 & -C_x C_y \\
  C_y^2 & C_x C_y & -C_y^2 & -C_x C_y \\
  C_x^2 & C_y^2 & C_x C_y & -C_x C_y \\
  C_y^2 & C_y^2 & C_x C_y & -C_x C_y \\
\end{bmatrix}_{4 \times 4}
\]

where

\[
L = \sqrt{(x_k - x_j)^2 + (y_k - y_j)^2}
\]

\[
C_x = \frac{x_k - x_j}{L}
\]

\[
C_y = \frac{y_k - y_j}{L}
\]
(A) **Non-Shape Design Variables**

\[
\frac{\partial [k^{(e)}]}{\partial \mathbf{A}} = \frac{\partial [k^{(e)}]}{\partial \mathbf{A}}_{Global} = \frac{E}{L} \times [T] = EASY!
\]

(B) **Shape Design Variables**

\[
\frac{\partial [k^{(i)}]}{\partial \mathbf{x}}_{Global} = \text{VERY TEDIOUS}!
\]

\[
\begin{align*}
\partial & \left\{ \begin{array}{c}
x_j \\
x_k \\
y_j \\
y_k \\
\end{array} \right\} \\
\end{align*}
\]
D.S.A. FOR TRIANGULAR PLATE BENDING ELEMENT

(Reference: Prezziemiernicki’s Book)

\[ k(\epsilon) \]_{Local} = \int_0^{D^3} D^T k D \, dV = \frac{D^3}{12(1 - \nu^2)} \]

\[
\begin{bmatrix}
4 & 0 & 2(1 - \nu) \\
0 & 4 & 0 \\
12x & 4x & 36x^2 \\
\end{bmatrix}
\]

Symmetric

\[
(12 - 8x(1 + \nu)) \begin{bmatrix}
4(1 - \nu) & 0 & 12x(1 + \nu) \\
0 & 12y & 36xy \\
12x(1 + \nu) & 36xy & 36y^2 \\
\end{bmatrix}
\]
Then

\[
[k^{(e)}]_{Global} = [k^{(e)}]_{Local} \cdot [T]
\]

Functions of element nodal coordinates also!

(A) **Non-Shape Design Variables**

\[
\frac{\partial [k^{(e)}]_{Global}}{\partial \mathbf{t}} = E A S Y
\]

(B) **Shape Design Variables**

\[
\frac{\partial [k^{(e)}]_{Global}}{\partial \mathbf{t}} = V E R Y \ T E D I O U S
\]
D.S.A. FOR TRIANGULAR PLATE/ SHELL BENDING ELEMENT

(Reference: Alex Tessler's Element)

Note: In this work, Alex Tessler's Plate/Shell finite element is used.
Applications

Example 1 (80 bays, 190 stories)
- NEL = 60,990 elements (truss)
- NEQ = 30,780 equations
- NTERMS = 5,171,574 terms
- AVEBW = 168 average bandwidth
- NDV = 96 non-shape design variables

Example 2 (1 bay, 1 story)
- NEL = 5 elements (truss)
- NEQ = 4
- NDV = 8 shape design variables

Example 3 (plate structure)
- NEL = 32 elements (Alera Tensile's plate/shell)
- NDV = 75 shape design variables
NUMERICAL RESULTS AND PARALLEL PERFORMANCE

<table>
<thead>
<tr>
<th>Tasks</th>
<th>Number of Cray - C90 Processors</th>
<th>Speed-Up Factors</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>1 proc.</td>
<td>8 proc.</td>
</tr>
<tr>
<td>(A)</td>
<td>0.4855(^{sec})</td>
<td>0.09954(^{sec})</td>
</tr>
<tr>
<td>(B)</td>
<td>0.9582(^{sec}) (0.9906(*)</td>
<td>0.1320(^{sec}) (0.1433*)</td>
</tr>
<tr>
<td>(C)</td>
<td>2.6290(^{sec})</td>
<td>0.3568(^{sec})</td>
</tr>
<tr>
<td>(D)</td>
<td>0.1019(^{sec})</td>
<td>0.1015(^{sec})</td>
</tr>
<tr>
<td>(E)</td>
<td>2.3717(^{sec})</td>
<td>0.3034(^{sec})</td>
</tr>
<tr>
<td>(F)</td>
<td>9.6934(^{sec})</td>
<td>1.2128(^{sec})</td>
</tr>
<tr>
<td>Entire Process</td>
<td>16.2740(^{sec})</td>
<td>2.2221(^{sec})</td>
</tr>
</tbody>
</table>
### Design Variables

<table>
<thead>
<tr>
<th>Design Variable</th>
<th>Total Memory Needed</th>
</tr>
</thead>
<tbody>
<tr>
<td></td>
<td>1000, 7881649</td>
</tr>
</tbody>
</table>

### Wall Clock Time

- Max. wall clock time for gen+assem = 0.25032322

### Derivatives

<table>
<thead>
<tr>
<th>d(z)/d(b) with respect to DV</th>
<th>Value</th>
</tr>
</thead>
<tbody>
<tr>
<td>d(z)/d(b) = 0.463915E-02</td>
<td>0.197448E-03</td>
</tr>
<tr>
<td>d(z)/d(b) = 0.584470E-04</td>
<td>0.463915E-02</td>
</tr>
<tr>
<td>d(z)/d(b) = 0.463915E-02</td>
<td>-0.150054E-03</td>
</tr>
<tr>
<td>d(z)/d(b) = -0.289055E-03</td>
<td>0.463915E-02</td>
</tr>
</tbody>
</table>

### Process Times

- Total time for gen+assem = 0.143736066

### Additional Times

- Time in boundc = 4.737786E-3
- Time in jointc = 3.0290400000002E-4
- Time in apload = 4.8300000000001E-5
- Time in elconn = 3.33129E-3
- Time in materp = 4.7050644E-2
#result

**ndofpe, nodes, ndofpn, nunrol, nummat, intal, pbays, story, ndv
7, 4, 2*2, 8, 1000, 0, 10, 300, (1000)**

*design variable, total memory needed= 1000, 7881649
max. wall clock time for gen+assem = 0.106087758

\[
d(z)/d(b) \begin{pmatrix} 0.463915E-02 & 0.197448E-03 & 0.463915E-02 & 0.127947E-03 & 0.463915E-02 \\ 0.584470E-04 & 0.463915E-02 & 0.110535E-04 & 0.463915E-02 & 0.805540E-04 \\ 0.463915E-02 & 0.150054E-03 & 0.463915E-02 & 0.110535E-04 & 0.584470E-04 \\ 0.463915E-02 & 0.219555E-03 & 0.463915E-02 & 0.805540E-04 & 0.289055E-03 \\ 0.463915E-02 & 0.358556E-03 & 0.463915E-02 & 0.428056E-03 & 0.463915E-02 \end{pmatrix}
\]

ME, time for generate SD=3, 2.441805E-2
ME, time for generate K =3, 8.4435756E-2
ME, time for Factori. =3, 9.3954744000001E-2
ME, time for generate SD=1, 2.2068600000003E-4
ME, time for generate SD=2, 2.5296000000008E-5
ME, time for generate K =1, 8.4171846000001E-2
ME, time for generate K =2, 8.452527E-2
ME, time for Factori. =1, 9.3851616E-2
ME, time for Factori. =2, 9.3939270000001E-2
ME, time for Solution =1, 6.437399999923E-5
ME, time for Solution =2, 5.441399999971E-5
ME, time for (dK/db)*X =1, 2.156280738
ME, time for (dK/db)*X =2, 2.16884262
ME, time for dX/db =1, 7.22125122
E, time for dX/db =2, 7.196945016
E, time for Solution =3, 2.2304856E-2
E, time for (dK/db)*X =3, 2.163674568
ME, time for dX/db =3, 7.211424354

** Time in boundc =4.723506E-3
** Time in jointc =3.020759999998E-4
** Time in upload =4.834200000007E-5
** Time in elconn =3.328746E-3
** Time in materp =4.6941834E-2
** Time in colht =1.0283598E-2

** TOTAL TIME: (nel, neq, ielm, nterms) 12300, 6600, 4100, 186532
** TOTAL TIME: (nel, neq, ielm, nterms) 12300, 6600, 4131, 186532
** TOTAL TIME: (nel, neq, ielm, nterms) 12300, 6600, 4131, 186532
CONCLUSIONS

1. Automatic Differentiation (ADIFOR) tool has been successfully applied to both simple (TRUSS) and complex (Alex Tessler’s PLATE/SHELL) finite elements.

2. Both non-shape and shape design variables can be successfully treated.

3. For the first time (to the author’s knowledge), ADIFOR tool can be applied in a parallel-vector computer environment for non-shape and shape sensitivity analysis.

4. The entire finite element + sensitivity analysis can be done with excellent parallel and vector speed (using all 16 Cray-C90 processors)
SESSION 6 Mosaic and the World Wide Web

Chaired by

Clyde R. Gumbert and John W. McManus

6.1 Introduction to the World Wide Web and Mosaic - Jim Youngblood
6.2 Use of World Wide Web and NCSA Mosaic at Langley - Michael Nelson
6.3 How To Use the WWW To Distribute Scientific & Technical Information (STI) - Donna Roper