A simultaneous luminescence pressure and temperature mapping system is developed including improved dye application techniques for surface temperature and pressure measurements from 5 torr to 1000 torr with possible upgrade to from 0.5 torr to several atmospheres with improved camera resolution. Adsorbed perylene dye on slip-cast silica is pressure (oxygen) sensitive and reusable to relatively high temperatures (~150°C). Adsorbed luminescence has an approximately linear color shift with temperature, which can be used for independent temperature mapping and brightness pressure calibration with temperature.
FIG. 1
FIG. 2

\[ Ko = 0.0011 \text{ l/torr} \]
\[ a = -0.0023 \text{ l/C} \]
\[ b = -0.0040 \text{ l/C} \]

FIG. 3

\[ c = 0.0024 \text{ l/C} \]
FIG. 5a

Applicable Range

c (1/C)

Ko (1/torr)

FIG. 5b

Ro

absorbed film thickness, arb. units
**FIG. 10a**

![Graph showing dP/P against pressure, torr for Imaging Resolution. The graph includes curves for <8-Bit, 10-Bit, and 12-Bit resolution.](image)

**FIG. 10b**

![Graph showing dP/torr against pressure, torr for <8-Bit, 10-Bit, and 12-Bit resolution.](image)
SIMULTANEOUS LUMINESCENCE PRESSURE AND TEMPERATURE MAPPING

This is a continuation of application Ser. No. 08/1492,686 filed on Jun. 20, 1995, now abandoned.

ORIGIN OF THE INVENTION

The invention described herein was made by an employee of the United States Government and may be used by and for the Government for governmental purposes without the payment of any royalties thereon or therefor.

BACKGROUND OF THE INVENTION

1. Technical Field of the Invention

The present invention relates generally to pressure and temperature mapping systems, and more particularly to simultaneous temperature mapping systems for use at high temperatures.

2. Discussion of the Related Art

Advanced diagnostic techniques capable of nonintrusive global measurement of aerodynamic pressure and heating distributions on models tested in hypersonic wind tunnels will significantly enhance the design and assessment process of future aerospace vehicles. Considerable progress has been made in measuring global temperature distributions with time, hence global heating distribution, via relative intensity thermographic phosphor and infrared emission techniques, but global pressure measurements at hypersonic conditions continue to elude experimental aerothermodynamicists.

Pressure mapping techniques using photoluminescence offer a potential savings in time and expense for wind-tunnel testing over conventional pressure instrumentation (i.e. discrete measurements via transducers) and are sought for hypersonic wind-tunnel applications.

Luminescence pressure mapping techniques have been developed and tested broadly by government agencies, industries, and universities for lower-speed applications. Current techniques use oxygen sensitive luminophores in polymer binders for pressure sensitive paints which can be applied to a variety of surfaces. Paint applications, however, are limited in high-temperature stability and time response by polymers. Temperature effects on induced luminescence further complicate pressure measurements in high-speed application with aerodynamic heating. Recent work was done to calibrate temperature effects on paint luminescence for wind-tunnel measurements using separate temperature mapping data. The results of this work and similar applications may expand near-term testing capabilities for relatively low enthalpy, long test duration facilities, but innovative solutions are needed for the bulk of hypersonic testing (i.e. M=6, moderate to high enthalpy, blowdown and impulse facilities) including an internal method for thermal calibration, higher-temperature stability and faster response times.

It is accordingly an object of the present invention to provide a system for simultaneously mapping temperature and pressure.

It is another object of the present invention to provide a system for simultaneously mapping temperature and pressure at high temperatures.

It is another object of the present invention to provide a system for simultaneously mapping temperature and pressure with reduced response time.

It is a further object of the present invention to provide a system for simultaneously mapping temperature and pressure which is suitable for use in hypersonic facilities.

SUMMARY OF THE INVENTION

According to the present invention, the foregoing and additional objects are obtained by providing a method for simultaneous mapping of temperature and pressure on an aerodynamic model, including the steps of applying a luminescent material, preferably perylene, to the surface of the model, exciting the luminescent material, measuring the resulting emissions, and determining the pressure and temperature on the surface of the member. The luminescent material on the surface may be excited using a blue light or long-wave ultraviolet light and the resulting emissions are measured using a two-color imaging system. The preferred method for applying the luminescent material is polishing the surface of the member, saturating the surface with the luminescent material, heating the saturated member, and repolishing the heated member. The member may be a slip-casted ceramic.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 is a figure showing the mounting of test samples;
FIG. 2 is a graph showing emission brightness vs. pressure model at constant temperatures;
FIG. 3 is a graph showing ratio difference vs. temperature;
FIG. 4 is a graph showing color ratio data versus temperature;
FIG. 5a is a graph showing temperature sensitivity and quenching constants vs. adsorbed film thickness;
FIG. 5b is a graph showing reference color ratio vs. adsorbed film thickness;
FIG. 6a is a graph showing time vs. data points;
FIG. 6b is a graph showing irradiance vs. data points;
FIG. 6c is a graph showing pressure vs. data points;
FIG. 6d is a graph showing temperature vs. data points;
FIG. 7a is a graph showing color ratio vs. data points for sample region 2;
FIG. 7b is a graph showing brightness vs. data points for sample region 2;
FIG. 7c is a graph showing pressure vs. data points for sample region 2;
FIG. 7d is a graph showing temperature vs. data points for sample region 2;
FIG. 8a is a graph showing color ratio vs. data points for sample region 3;
FIG. 8b is a graph showing brightness vs. data points for sample region 3;
FIG. 8c is a graph showing pressure vs. data points for sample region 3;
FIG. 8d is a graph showing temperature vs. data points for sample region 2;
FIG. 9a is a graph showing reference brightness vs. data points;
FIG. 9b is a graph showing reference color ratio vs. data points;
FIG. 9c is a graph showing luminescence pressure vs. data points;
FIG. 9d is a graph showing luminescence pressure vs. data points;
FIG. 10a is a graph showing relative pressure uncertainty for various imaging resolutions; and
which was developed for luminescence thermography. Ultraviolet (UV) (365 nm) excitation was also used for irradiation. The quantum yields and quenching constants for quantum yield and oxygen quenching and phosphorescence. For some materials, quenching rates and sensitivities, respectively. Dye application and testing procedures varied, such as with successively finer grits of silicon carbide paper to a highly-smooth finish. Polishing residue is removed with solvent. The surface is then saturated by brush or dipped in a specific dye concentration with toluene. It is then allowed to dry and baked in a clean convection oven for thirty minutes at 160°C. Saturation and baking are repeated to ensure uniform application. Finally, the surface is re-polished with silicon-carbide paper removing accumulated dye layers on the exposed surface, hence, isolating film adsorption to ceramic pores which are more directly regulated by saturated dye concentration.

Ceramic disk samples were machined, 1-mm thick, from slip-cast silica rods 1.27-cm in diameter and drilled with 0.60-mm holes for mounting. Following dye application, samples were mounted as shown in FIG. 1 to electrical connectors in a ceramic block with butt-welded 0.076-mm diameter type K thermocouple wire and 0.51-mm diameter leads. As many as 5 test samples were mounted per test.

Through a test chamber window, image sampling regions are indicated in FIG. 1 with numbers 1 through 4. Region 1 is taken from an analysis sample prepared with low dye concentration. Regions 2 and 3 are from a test sample with optimum dye concentration and region 4 is taken from a test sample with much greater dye concentration. The test chamber was constructed from a 20.5-cm diameter pipe, lined with 2.54-cm thick low-density alumina ceramic insulation, fitted with an electrical resistance heater and o-ring sealed with a 2.54-cm thick quartz viewing window.

Chamber pressure was controlled via a vacuum pump and venting to atmosphere and was measured with a 1000 torr variable capacitance pressure transducer and Hastings vacuum gage down to 0.1 torr. Chamber insulation was effective in providing stable temperatures up to 150°C, taking approximately 4 to 6 hours for the chamber to cool from 150°C to room temperature at 0.5 torr. Measurements made in the vacuum oven revealed an approximately linear shift in quantum efficiency, and quenching coefficient K in Equation (1) with temperature. Experimental data could be approximated to first order from Eqn. (1) for green brightness measurements as:

\[
\frac{I}{I_0} = \frac{1}{1 + K_0(1 + aT)T^b}
\]
plot of Eqn. (4) with experimentally derived constants is shown in FIG. 2 for air pressure at constant temperatures. An approximately linear shift in emission color with temperature was observed from adsorbed perylene dye from 20° C. to 150° C. with a two-color imaging system. This characteristic was not expected but is highly desirable for temperature measurement and calibration. This allows for development of a system with a single dye application and simplifies filtering of broadband emission for diagnostics. In addition, color-ratio measurements were found to be independent of pressure and excitation irradiance. A linear approximation for green-to-red brightness ratio with temperature is given as

$$\frac{I_g}{I_r} - R_0 = cT$$

(5)

where c is the color-ratio temperature sensitivity and was found to be constant over a broad range of applications. $R_0$ is a calibration coefficient for color ratio at zero temperature and was found to vary from sample to sample with apparent film thickness. A plot of Eqn. (5) is shown in FIG. 3 for temperature with experimentally derived constant c.

Image sampling regions consisted of 10x10 element pixel arrays and were averaged and treated as point data. Individual brightness response was calibrated separately for each sample region and color using a calibrated uniform intensity light source, or integrating sphere, with micrometer controlled aperture brightness and 2855K color temperature. Indepedant effects on quenching coefficients also include thinner adsorption. Determination of quenching constants showed similar values for regions 1 through 3 (K\textsubscript{c}=0.0011 torr\textsuperscript{-1}), and much lower value for region 4 (K\textsubscript{c}=0.0005 torr\textsuperscript{-1}).

From this data and similar observations FIGS. 8(a) and (b) were constructed as a guide for future dye applications. Stable values of temperature sensitivity and quenching constants (c and K\textsubscript{c}) are shown to exist over a range of film thicknesses that correspond to a range of calibration coefficient $R_0$ roughly from 0.9 to 1.4. Independent effects on quenching coefficient also include ceramic contaminants, interference with humidity and other gases, and later surface contaminants from handling and tunnel environments. Possible humidity effects are noted but are expected to be less of a concern in dry tunnel air supplies. Later surface contaminants may also be removed with most alcohols since the perylene dye is not easily soluble.

Test conditions for recent experiments are shown in FIGS. 6. (a) to (d). Data series were taken from four contiguous runs made over a three-day period. The longest run was several hours in duration. The first run began shortly after installing samples and evacuating the test chamber. From data points zero through 56, the vacuum oven was sealed except through a vent to atmosphere during pressure sweeps. Between points 56 and 59, the chamber was vented to atmosphere overnight with the quartz window removed.

Points 59 to 61 were taken during re-evacuation from atmosphere to vacuum for the final run. A gap is shown between data points 61 and 64 to indicate changes made in camera gains.

UV irradiance values shown in FIG. 6(b) were monitored using a photoelectric detector attached to the window perimeter. A substantial drift is shown in irradiance values over the longest run. This detector was used in observations only and not in subsequent calculations. Irradiance was purposefully decreased for data points 12 through 14 using neutral density filters (N.D.=0.1, 0.2, and 0.3) in the beam path.

FIGS. 7 and 8 show the calibrated camera output in color ratio and individual brightness measurement for sample regions 2 and 3, respectively. It is noted that color ratio measurements are independent of pressure and irradiance and seem to follow closely thermocouple measurements. Apparent drifts, however, are shown between the color ratio and thermocouple measurements at constant temperatures, which seem to be associated with initial pump down from atmosphere. This may be attributed to atmospheric humidity, which was noticeably greater before the final run evacuation in comparison with initial run evacuation. These effects do not appear to be associated with cycling temperature or other forms of degradation. Suggested methods for wind-tunnel applications would include pulling down test section pressure several hours before run times to stabilize luminescence parameters. Maximum drift rates as demonstrated, however, are well within wind-tunnel run rates.

In FIG. 9 luminescence pressures and temperatures are determined using periodic “wind-off” calibrations ($I_0$ and $R_0$) at reference data points separately for regions 2 and 3. Previously noted drifts are shown in both reference brightness and color-ratio calibration coefficients calculated at periodic intervals. The corresponding pressure and temperature model used in calculations ($K_0$, a, b and c) was shown in FIGS. 1 and 2.

Sensible pressure range as well as resolution is primarily limited by the dynamic resolution of the imaging system. In present measurements, video output is effectively digitized to less than 8-bits. Pressure measurement uncertainty can be derived as a quadratic function of pressure multiplied by relative brightness resolution from Eqn. (1), at constant temperature as

$$dp = \frac{dI}{I} \left(1 + \frac{KP^2}{K}ight)$$

(6)

Values of $I_0/dl$ used in uncertainty calculations were 200, 1000 and 4000, respectively, for the present imaging system and performance estimates for 10-bit and 12-bit systems. FIGS. 10 (a) and (b) show calculations using Eqn. (6), and K=0.001 torr\textsuperscript{-1} at 25° C., for relative pressure uncertainty in vacuum and absolute uncertainty over full range. Calculations show that the present system is operable from 5 torr to 1000 torr with greater than 1 signal-to-noise and from 0.5 torr to several atmospheres with higher imaging resolution.

Relatively bright luminescence was observed from adsorbed perylene dyes with blue-light excitation for temperatures up to 150° C. and greater in vacuum and atmosphere, and was measured using standard video. Comparative applications of a porphyrin dye were vaporized or permanently quenched under similar test conditions. A plot of Eqn. (4) with experimentally derived constants is shown in FIG. 2 for air pressure at constant temperatures.

In subsequent testing and later in wind-tunnel applications, pressure and temperature data are derived from Equations (4) and (5) in the following manner. First it is assumed that $K_0$, a, b, and c are constants which can be applied to all points on a sample or model surface. In a typical application, a “wind-off” measurement is first made for static pressure, temperature and separate color brightness
What is claimed is:

1. A method for simultaneous mapping of temperature and pressure on a member, comprising:
   - applying exactly one luminescent material to the surface of the member;
   - exciting the luminescent material;
   - measuring the resulting emissions; and
   - determining the pressure and temperature on the surface of the member.

2. The mapping method of claim 1 wherein the luminescent material is perylene dye.

3. The mapping method of claim 1 wherein the luminescent material is excited using a blue light.

4. The mapping method of claim 1 wherein the luminescent material is excited using long-wave ultraviolet light.

5. The mapping method of claim 1 wherein the resulting emissions are measured using a two-color imaging system.

6. The mapping method of claim 1 wherein the luminescent material is applied by:
   - polishing the surface of the member;
   - saturating the surface with the luminescent material;
   - heating the saturated member; and
   - repolishing the heated member.

7. The mapping method of claim 1 wherein the member is a ceramic model.

8. The mapping method of claim 7 wherein the ceramic model is a slip casted ceramic.

* * * * *