The Geomagnetic Field and Radiation in Near-Earth Orbits

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Geomagnetic Field Lines
From 40 W and 140 E

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ABSTRACT

This report shows, in detail, how the geomagnetic field interacts with the particle flux of the radiation belts to create a hazard to spacecraft and humans in near-Earth orbit. It illustrates the geometry of the geomagnetic field lines, especially around the area where the field strength is anomalously low in the South Atlantic Ocean. It discusses how the field will probably change in the future and the consequences that may have on hazards in near space.
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INTRODUCTION

It has been known for more than 40 years that the anomalously weak geomagnetic field in the South Atlantic Ocean, near the coast of Brazil, is important to the radiation environment near Earth. Radiation belt particles, trapped by lines of force of the geomagnetic field, approach closer to the Earth’s surface here than elsewhere. This is an area of intense radiation in near-space and is a subject of extensive research by space scientists. This radiation is usually described on a heuristic basis from measurements made in space and “models” have been developed on this basis. There are also attempts to understand the radiation in terms of the laws of physics, i.e., particle motion in the geomagnetic field, and how to effect shielding from this radiation. Space mission and instrument designers today rely on both these approaches.

There is an increasing use of near-Earth orbits for spacecraft to study the Earth, for space telescopes, for the manned space station, and for military and commercial applications. All these spacecraft and humans in space suffer very significant damaging effects of radiation. Since most of this radiation is related to the geomagnetic field, this report will take a brief but closer look at the correspondence between the geomagnetic field and radiation and how this situation may become worse in the future.

Many studies of radiation in near space have treated the geomagnetic field as if it was due to a simple dipole. With the current field model we examine details of the field at any latitude, longitude, and altitude, out to about 4 Earth radii (25,484 km) where solar time-varying effects become important and the magnetospheric boundary is felt. Many features of the actual geomagnetic field in space have not been illustrated and this report attempts to show that.

The IGRF spherical harmonic coefficients for 1995 (Barton, C.E. et al., 1996) are used to represent the actual geomagnetic field, as of the beginning of the year 1995. This version of the IGRF, as of this writing, is the most recently approved reference field and has been extensively studied in many publications. This IGRF model also carries the time derivative of the spherical harmonic coefficients so one may calculate the time variations of the field (although one is cautioned that the accuracy will degenerate if the calculation is extended too far from 1995). The three-dimensional complexity of the geomagnetic field is such that one should have a computer to display any of the many aspects of the field geometry, or show how the degree and order of the IGRF terms influence the field configuration. Here we used the IGRF to degree (n) and order (m) 10, although fewer terms were occasionally used to see how significantly they

---

A desktop computer was used to calculate the geomagnetic field components from the International Geomagnetic Reference Field (IGRF) coefficients and to produce simple graphics. Computer codes for calculations and graphics were written in TrueBasic. A subroutine to calculate the field from the spherical harmonic coefficients follows that of Quinn (Quinn, et al., 1995) and sources quoted there, converted from the original Fortran to TrueBasic language (see appendix). These programs were run on a Power Macintosh 9600 desktop computer.

There are several programs available to calculate the field components from the IGRF coefficients, for example see http://www.ngdc.noaa.gov/IAGA/wg8/wg8.html.

To make global maps of the geomagnetic field the total field was calculated for each 5 degrees of latitude and 5 degrees of longitude and compared to published tables of these values, where available. The results matched to within a few nT. This matrix of values was entered in a graphics program (Spyglass) and the results contoured to produce maps of the total field of the Earth.

The Geomagnetic Field...
changed the results and to see how the field appears with only the tilted and displaced dipole. It is the total field magnitude, rather than field components, that are sufficient for purposes of this report; however, the directions of the field are used in the ray tracing of the lines of force.

Four principal anomalous regions of the field at the Earth’s surface are the localized highs in northern Siberia, in northern Canada and near Antarctica south of Australia, and the large, elongated low near the coast of Brazil. This later feature is the South Atlantic Anomaly (SAA), with a minimum of about 23,000 nT located at 26 S, 53 W, about 700 km inland from the coast of Brazil. It is frequently said that the SAA is due to the displacement of the geomagnetic dipole axis from the center of the Earth, in the direction of the Pacific Ocean and the tilt of the dipole with respect to the rotational axis.

This point is made clear in figure 2. The upper part of this figure shows the field that would result from an axial, inclined dipole. That is the field that is produced using only the first three terms of the IGRF 1995, all of which have \( n = 1 \). In equatorial latitudes the contours follow the magnetic equator.

The lower part of figure 2 shows the field that would be produced by an offset, inclined dipole. This field is due to only the first 8 terms of the IGRF 1995, where terms have \( n = 1 \) and 2. The displacement of the dipole is 526.89 km in the direction of the northwest Pacific (21.47 N, 144.77 E). The inclined, offset dipole creates an Antarctic high south of Australia and an SAA off Brazil but it does not create the highs in Northern Canada and Siberia. The SAA in this figure has many of the characteristics of the SAA of the complete field (figure 1). However, the actual SAA is stretched more towards the southern tip of Africa, showing that terms with \( n \) greater than 2 play a significant role shaping that Anomaly. The shape of the field contours in figure 2 (Lower) greatly resemble the actual field at an altitude of 5000 km (figure 3, lower right), showing that terms with \( n \) greater than 2 have little effect at this altitude.

---

\(^2\)The SAA was also called the Brazilian Anomaly (Dessler and Karplus, 1960) or Capetown Anomaly (Heirtzler and Hirshman, 1960; Cain, et al., 1968).
Figure 2. (Upper) Total Field (nT) due to centered, inclined dipole. (Lower) Total Field (nT) due to offset, inclined dipole.

Figure 3. Contours of total field intensity, in nT, at various altitudes above the Earth's surface, using IGRF 95. Upper left 200 km altitude, upper right 500 km, lower left 1000 km, lower right 5000 km.
Maps of the total field at elevations of 200, 500, 1000, and 5000 km are shown in figure 3. Many low-Earth orbit (LEO) spacecraft fly at altitudes between 500 and 1000 km and the field they experience is shown here. The shape of the contours of the SAA are rather different than at the Earth’s surface. The weakest part of the field is still off the coast of Brazil but there is a tendency for the low to be stretched in the direction of Africa. From 200 to 500 km the strength of the SAA is reduced by about 4000 nT, and from 500 to 1000 km the SAA is reduced by another 4000 nT. At 5000 km the overall field strength is much reduced over its value at lower elevations, rarely reaching 10,000 nT, and the difference between highs and lows is less pronounced.

The variation of the field with altitude over the four anomalous regions is shown in figure 4. The SAA low covers a larger portion of the Earth's surface than the highs and figure 4 shows that the variation with height for the SAA is much less than for the other three areas. For all the regions the field strength at 5000 km is about one-third what it is on the surface. At 10,000 km (somewhat less than 2 Earth radii) the field strength is about one-tenth the surface value. Above about 15,000 km (not shown) the field strength is much reduced and the four anomalous regions (including the SAA) have lost their distinctive appearances. Geostationary orbits (36,000 km) are beyond the influence of the SAA, although there are other radiation hazards, due to other effects, at this altitude.

**GEOMAGNETIC FIELD LINES IN SPACE**

Usually early drawings of the Earth’s magnetic field lines were more cartoon than fact and did not reflect their true configuration. However, there were attempts to describe the geomagnetic field in a fashion useful for studying the radiation environment (for example McIlwain, 1961; Mlodnosky and Helliwell, 1962). With the availability of inexpensive but fast desktop computers it is easy to get an accurate presentation of the field using all the spherical harmonic components. We show some examples here.
The straightforward procedure for drawing a line of force (ray tracing) is to start with a point at a given latitude, longitude and altitude. For this point the IGRF is used to find the direction of the field (declination and inclination). The next point will be a specified distance in the direction indicated by the field. The accuracy of the position of the line of force will depend upon this step size. The smaller the step size the more accurate the line but the longer it will take the program to run. Using different step sizes we found that a 110-km step produced sufficient accuracy, although this was slightly degraded along arcs at distances of about 4 Earth radii.

For this second point the direction of the field was again calculated, and position of the third position determined, etc., with the process stopping when the next altitude would be less than zero. This provides a series of points defining a line of force with the coordinates and field components specified for each point. Usually we took lines originating in the Southern Hemisphere (because the SAA is in the Southern Hemisphere) with a loop stepping the origin to different starting latitudes and/or longitudes as each line is finished. In special cases, for example, where we are plotting lines crossing the shuttle’s path (figure 7), it was necessary to start the line on the shuttle’s path and proceed in both south and north directions from there.

To illustrate the asymmetry of the lines of force we show (figure 5) lines of force originating along 40 W near the longitude of the SAA and also along 140 E, which is separated by 180 degrees and is near the longitude of the Antarctic and Siberian highs. Here the lines originate at 20, 40, and 60 S. Lines which originate at 60 S reach nearly 4 ER near the equator along 140 E but only about 1 ER on the opposite side of the Earth. Notice that the line which leaves 60 S terminates near 80 N along 140 E, but only at about 45 N along 40 E. The tilt of the dipole axis is evident. When the Earth rotates during the course of the day, this asymmetric field creates a changing, magnetically complex region in near space.

![Figure 5](image-url)
Figure 6 is another display of field lines, originating at various longitudes but always from 70 S. This displays the range of altitudes reached by lines originating from different longitudes. For this latitude (70 S) lines starting from 80 W are closest to the surface of the Earth and the line staring from 120 E are the most distant. It is clear that any charged particle within 1 or 2 ER would experience different forces depending upon which side of the Earth it approached.

![Figure 6. Lines of force from 70 S, for various longitudes.](image)
Another way of getting a broad view of field lines is to show their projection on the surface of the Earth (figure 7). Here lines originate at 10, 30, and 50 S and at various longitudes. The projection of the line which originates near the SAA (30 S, 30 W) appears to cross the line originating at 50 S, 30W. Other lines appear to cross as well.

Figure 7. Lines of force originating from 10 S, 30 S, and 50 S, and from each 30 degrees of longitude, as projected on the Earth’s surface.
Actually they are at different altitudes and do not physically cross in space. If one looks at the lines of force crossing the shuttle’s path one gets another perspective (figure 8). Since the shuttle is at constant altitude (480 km here) it will cross more polar or more equatorial lines as it moves from high to low latitudes. These are shown for two common orbital inclinations: 28.5 and 57 degrees. In both cases an orbit was chosen which goes through the SAA. On the 28.5 degree inclination orbit none of the lines of force reach high (auroral) latitudes. With the 57 degree orbit the shuttle is frequently near the ends of a line of force. In those higher latitudes the lines of force are nearly vertical and do not extend much past the flight path. At the equator the shuttle is near the top (maximum altitude) of the line of force and the lines of force are more horizontal.

**Figure 8.** Lines of force crossed by the shuttle on an orbit crossing the SAA. On the left is an orbit with 28.5 degrees inclination. On the right is an orbit with 57m degrees inclination.
Figure 9 is yet another illustration of the lines near the SAA. It shows the field strength along three lines of force starting at latitudes 20 S, 30 S, and 40 S, all along 60 W and along three lines originating at 0, 30 W, and 60 W and all at 30 S. The further south the origin, the weaker the field near the equator because of its higher altitude, but the field is stronger at the ends of the lines where the lines are at higher latitudes. The field strength at the equator of the line starting on the Greenwich meridian is less than the others because that line rises to higher altitudes along the equator. One notices that all lines have the same strength at about 28 S, very near the latitude of the SAA. Of course, this does not mean that all lines have the same altitude at the crossover point shown.

![Figure 9](image.png)

**Figure 9.** (Left) Field strength vs latitude for starting latitudes 20 S, 30 S, and 40 S, all at 60 W longitude. (Right) Field strength vs latitude for starting longitudes 0, 30 W, and 60 W.

**PARTICLE MOTIONS AND THE UPPER ATMOSPHERE**

For human safety and for engineering purposes it is important to know the radiation environment expected in near-space missions. Ad hoc “models” have been developed from observational data. There are two models. One model describes the electron flux and another the proton flux. NASA Trapped Radiation Models AE-8 and AP-8, for electron- and proton-produced radiation respectively, are available from the National Space Science Data Center (http://nssdc.gsfc.nasa.gov) and described by Fung, 1996. Countries other than the U.S., notably Britain and Russia, have other models. The models give the particle flux as a function of particle type, particle energy, position, and time.

No extensive discussion of charged particles in space will be given here, only enough of the basic principles to understand the effect of the geomagnetic field on the radiation. A more complete and convenient description can be found on the homepage of the (Belgium) Space Environment Information System (http://www.spenvis.oma.be/spenvis/).
The motion of a charged particle is characterized by the particle's energy, mass, charge and pitch angle at
the equator (angle made with the line of force). Their motion in the radiation belts is depicted as being of
three basic types, all occurring simultaneously and all controlled by the geomagnetic field (for general
discussions see Walt, 1994, or Roederer, 1970). The first type of motion is a circular motion with a
gyroradius about the field line (period of the order of milliseconds). This radius depends upon the strength
of the field where the particle is located and its local pitch angle. Stronger fields cause the radius to be
less; the radius is greater for greater pitch angles and greater energies. Figure 10 shows the radius for
selected points, selected field lines, and selected energies for electrons and protons. It is seen that these
radii are of the order of meters for electrons and kilometer for protons. The radius will change as the
particle moves along a field line into different field strengths.

![Diagram of gyroradius calculation](image)

**Figure 10.** Heights of lines of force starting at 20 S, 30 S, and 40 S, all along longitude 60 W. The gyroradius for electrons (e), and protons (p), are given in meters for 1, 10, and 20 Mev particles at the equator with pitch angles of 90 degrees.

The second motion is the bounce back and forth along a field line from one hemisphere to the other,
reversing direction at a mirror point. This period is on the order of seconds. This motion is an adiabatic
invariant, requiring that the magnetic field strength divided by the square of the sine of the pitch angle is a
constant anywhere along the field line, regardless of the charge, mass, or energy of the particle. Since the
pitch angle at the mirror point ($\Phi_m$) is 90 degrees we can find the value of the field strength at the mirror
point \( B_m \) by the equation

\[
\frac{B_e}{\sin^2(\Phi_e)} = \frac{B_m}{\sin^2(\Phi_m)} = B_m
\]

where \( B_e \) is the field strength at the equator and \( \Phi_e \) is the pitch angle at the equator and \( \Phi_m \) is the pitch angle at the mirror point (90 degrees).

Since we know the angles of the field along the field line we can find the place where the dip angle changes sign, i.e., the equator, and the value of \( B_e \) there. Equation 1 shows that there is a different value of \( B_m \) for each equatorial pitch angle. Knowing \( B_m \) we can find the altitude of the mirror point. Figure 11 shows the heights of lines of force vs latitude for south points of origin along longitude 53 W. Only the lower 1000 km of these lines are shown and mirror altitudes are shown along each line for various equatorial pitch angles. Although the longitude of these lines goes through the SAA, this figure does not give any indication why charged particles are trapped at the SAA. The longer (higher altitude) field lines have lower pitch angles near the Earth’s surface.

**Figure 11.** Height of field lines vs latitude for lines starting along longitude 53 W. Numbers on lines are mirror points for particles with equatorial pitch angles indicated. Particles are scattered or absorbed in the atmosphere below about 100 km altitude.
It should be kept in mind that the particle flux density is not the same for all angles at the equator. Here one relies on the cumulative observations that have been made on many spacecraft. The distribution of particle flux at the equator is expressed as function of pitch (Φ) angle by the expression

\[ J(\Phi) = J \sin^n(\Phi) \]  \hspace{1cm} (2)

Where J is a constant (value of J(Φ) when Φ=90) and n is the anisotropy index and has been given by Fung, 1996. Figure 12 is a plot of \( \sin^n(\Phi) \), or the ratio of \( J(\Phi)/J \), vs Φ for values of n from 1 to 8, indicating that, even for relatively small values of n, most of the flux will be at very high pitch angles.

Fung shows the electron and proton particle density of a function of n, but the number of particles with lower values of n is relatively small. Then, referring to figure 11 we see that most particles have a mirror location in the upper part of their field lines and much fewer come to lower elevations where they might interact with the atmosphere.

The third type of motion is the drift of particles around the Earth from east to west or west to east, usually taking about an hour. As particles gyrate about a field line they reach neighboring field lines to the east or west. There may be a stronger or weaker field on the neighboring line, shortening or lengthening the gyroradius. This will cause the particle to drift to the east (electrons) or west (protons). This effect may be different at different altitudes, latitudes, and longitudes because the field strength varies with these variables. This motion of charged particles is called the “ring current.”
Scattering by the atmosphere removes particles from the lower end of the radiation belts. This is illustrated by the reduction in astronauts’ dose rate during sunspot maximum, compared to sunspot minimum (Badhwar, et al., 1997). At sunspot maximum the atmosphere is believed to expand due to heating, removing more particles from the lower end of the radiation belts. Although there are more solar flares—producing more particles in the radiation belts—during solar max, this effect is of shorter duration and is counter to the effect of removing radiation-belt particles by the expanded atmosphere.

Models of the upper atmosphere are available on-line from the National Space Science Data Center (NSSDC) at URL (http://nssdc.gsfc.nasa.gov/space/model/atmos/msice.html). This source provides data for the MSIS-E-90 (Mass-Spectrometer-Incoherent-Scatter) model, which in turn is based largely on the COSPAR International Reference Atmosphere (CIRA) 1986 model. This model is an improvement over the frequently quoted U.S. Standard Atmosphere (1976). Figure 13 shows, as an example, a height-density profile of the atmosphere for 06 hrs UT, 1 January 1995, in the vicinity of the SAA.

On a worldwide basis the atmospheric density is highest in October and April and lower in July and January. It also varies with latitude, longitude, time of day, geomagnetic activity and, as just mentioned, sunspot cycle. NSSDC atmospheric models, plotted for different locations and times, show that most of the density change occurs in the lower atmosphere and that figure 13 is a typical example.

The number density is dominated by molecular nitrogen below 80 to 100 km. Above this altitude atomic oxygen dominates the particle number density. However, the total particle density (O plus N$_2$) drops off drastically by about 7 orders of magnitude. The sensible top of the atmosphere is usually taken to be 100 km.
Interactions with the atmosphere are not necessarily responsible for the increase in radiation over the SAA. However, the atmosphere is responsible for removing particles from the radiation belts.

RADIATION DAMAGE TO SPACECRAFT

There are many types of hazards to spacecraft. Aside from the hazards caused by space debris and the problems associated with the vacuum of space, there are effects caused by the interaction of spacecraft with its plasma environment. These include spacecraft wake effects, spacecraft contamination effects, spacecraft power system effects, and spacecraft electromagnetic effects. There are major effects due to the natural, undisturbed radiation in space and these we explore here.

All spacecraft suffer radiation damage. Designing electronics and other equipment to minimize the effect of this damage is a major engineering concern, affecting both the choice and redundancy of electronic components and circuits, as well as software design. Especially over the SAA, there is significant radiation damage to electronic, optical, and computer systems. This radiation is of various types, depending upon the type of particle and its energy, the amount of radiation shielding, and the sensitivity of the materials receiving this radiation. One common type of radiation damage is spacecraft charging whereby charge is built up with time until arcing occurs. Another and very common type of damage is a “Single Event Upset” or SEU, whereby a single and critical digital component changes state. Clever design can frequently cause such problems to be reduced, although the spacecraft may be inoperative for a period of time. Some spacecraft turn off electrical circuits or optical systems when passing over the SAA.

Most ground controllers keep records of the type of damage, the time and where such damage occurred. An archive of such records is available from the National Geophysical Data Center in Boulder, URL address (http://spider.ngdc.noaa.gov:800/production/html/GOES/anom5jt.txt) or by ftp://ftp.ngdc.noaa.gov/1STP/ANOMALIES/.

An example of the SEU’s for one spacecraft is given in figure 14. This is for TOPEX, at an altitude of 1340 km, for the period 1992–1998, with total field strength contours shown for approximately the same altitude. The points do not cluster evenly within a contour but are more scattered in a northeast-southwest direction. (Characteristically, SEU’s for various spacecraft are not rigidly confined by any total field contour line.)
There were 282 SEU’s over this 6-year period. This rate of a few per week to a few per month seems typical. It is seen that most of the SEU’s are located in and about the SAA, although there is no clear contour that defines where they will occur. An examination was made of South America geomagnetic observatory records for times when there was an SEU nearby. These records indicated no unusual geomagnetic activity. Another investigation looked at the 3-hour Ap activity index when there was an SEU. Figure 15 shows the distribution of Ap when there was an SEU. There are relatively few high Ap and the distribution of Ap is not greatly different from the distribution of all Ap during the 1992-1998 period. Although it is known that there is a burst of radiation in space when there are solar flares it is impossible to predict exactly when an SEU will occur. Table 1 is a list of some spacecraft which have been, are, or will be, in Low-Earth Orbit and suffer radiation damage. A notable case is the Hubble Space Telescope,
whose instruments are turned off over the SAA to prevent damage to its optical fibers, electronic components, and solar arrays. Other spacecraft have had problems with star cameras and other instruments. The shuttle, Mir, and the International Space Station Alpha are manned space missions which have operated or will operate for extended periods of time in this environment. Spacecraft in higher orbits also suffer radiation damage. For example, spacecraft in geosynchronous orbits, although at an altitude of 35,790 km (about 6 Earth radii) and in an equatorial orbit also suffer radiation damage. These spacecraft rotate with the line of force on which they are located and these lines terminate at very high latitudes on the Earth’s surface.

![Figure 15. Distribution of Ap-index values for times of TOPEX SEU's.](image)
Table 1. Some LEO Spacecraft.

<table>
<thead>
<tr>
<th>Name</th>
<th>Country</th>
<th>Launch Date</th>
<th>Incl. (deg.)</th>
<th>Mean Alt. (km)</th>
</tr>
</thead>
<tbody>
<tr>
<td><strong>EXISTING</strong></td>
<td></td>
<td></td>
<td></td>
<td></td>
</tr>
<tr>
<td>ASTRID-2</td>
<td>SWEDEN</td>
<td>12/98</td>
<td>83</td>
<td>1000</td>
</tr>
<tr>
<td>COBE</td>
<td>US</td>
<td>11/89</td>
<td>99</td>
<td>882</td>
</tr>
<tr>
<td>ERS-1</td>
<td>ESA</td>
<td>7/91</td>
<td>99</td>
<td>784</td>
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<td>EUVE</td>
<td>US</td>
<td>7/92</td>
<td>28</td>
<td>506</td>
</tr>
<tr>
<td>GRO</td>
<td>US</td>
<td>4/91</td>
<td>28</td>
<td>434</td>
</tr>
<tr>
<td>HST</td>
<td>US</td>
<td>4/90</td>
<td>28</td>
<td>595</td>
</tr>
<tr>
<td>LANDSAT-5</td>
<td>US</td>
<td>4/84</td>
<td>98</td>
<td>703</td>
</tr>
<tr>
<td>LANDSAT-7</td>
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<td>4/99</td>
<td>98</td>
<td>700</td>
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<td>RUSSIA</td>
<td>2/86</td>
<td>52</td>
<td>392</td>
</tr>
<tr>
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<td>US</td>
<td>5/90</td>
<td>99</td>
<td>815</td>
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<tr>
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<td>DEN.</td>
<td>2/99</td>
<td>97</td>
<td>715</td>
</tr>
<tr>
<td>ROSAT</td>
<td>GER.</td>
<td>6/90</td>
<td>53</td>
<td>544</td>
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<td>US</td>
<td>7/92</td>
<td>82</td>
<td>593</td>
</tr>
<tr>
<td>SEAWTFS</td>
<td>US</td>
<td>9/97</td>
<td>98</td>
<td>705</td>
</tr>
<tr>
<td>SHUTTLE</td>
<td>US</td>
<td>many</td>
<td>28.5, 57</td>
<td>300</td>
</tr>
<tr>
<td><strong>SPOT</strong></td>
<td>FRANCE</td>
<td>several</td>
<td>99</td>
<td>825</td>
</tr>
<tr>
<td>SUNSAT</td>
<td>S. Afr.</td>
<td>2/99</td>
<td>96</td>
<td>715</td>
</tr>
<tr>
<td>TOPEX</td>
<td>US/FR</td>
<td>6/92</td>
<td>66</td>
<td>1338</td>
</tr>
<tr>
<td>TRIMM</td>
<td>US</td>
<td>11/97</td>
<td>35</td>
<td>350</td>
</tr>
<tr>
<td>UARS</td>
<td>US</td>
<td>8/91</td>
<td>57</td>
<td>578</td>
</tr>
<tr>
<td><strong>PLANNED</strong></td>
<td></td>
<td></td>
<td></td>
<td></td>
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<tr>
<td>CHAMP</td>
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<td>/00</td>
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<td>375</td>
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<tr>
<td>DMSP-B5</td>
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<td>8/99</td>
<td>polar</td>
<td>840</td>
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<tr>
<td>FEDSAT</td>
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<td>/01</td>
<td>70</td>
<td>750</td>
</tr>
<tr>
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<td>DEN.</td>
<td>/02</td>
<td>polar</td>
<td>600</td>
</tr>
<tr>
<td>NPOESS</td>
<td>US</td>
<td>/08</td>
<td>polar</td>
<td>833</td>
</tr>
<tr>
<td>SAC-C</td>
<td>ARGENT.</td>
<td>12/99</td>
<td>98</td>
<td>702</td>
</tr>
<tr>
<td>SAC1-1</td>
<td>BRAZIL</td>
<td>7/99</td>
<td>99</td>
<td>750</td>
</tr>
<tr>
<td>VCL</td>
<td>US</td>
<td>2/00</td>
<td>65</td>
<td>400</td>
</tr>
<tr>
<td>GLAS</td>
<td>US</td>
<td>/01</td>
<td>94</td>
<td>600</td>
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RADIATION HAZARDS TO HUMANS

Radiation in the radiation belts offer several constraints and hazards to humans in low-Earth orbit. The ionizing radiation found in LEO is thought to come, largely, from two sources: (1) trapped radiation which originates on the Sun and with the proton component being the most dangerous, and (2) galactic cosmic rays, which are extremely energetic and which occur at unpredictable times. Shielding offers little protection from radiation belt particles from the Sun and virtually no protection from galactic rays (see Wilson, et al., 1997). Outside the Earth's magnetic field (outside the magnetosphere) there are other very significant radiation hazards but, since those are not controlled by the Earth's magnetic field, they will not be discussed here. The environment of low-Earth orbit is where humans will be spending much time in the coming years and decades. Within NASA the Spacecraft Radiation Health Program at Johnson Space Center (http://sn-io.jsc.nasa.gov/sreg/techmemo.htm) leads in the research and development activities that address the health effects of space radiation exposure to astronauts. Astronauts carry personal radiation dosage badges and radiation monitors are located throughout the shuttle. There is a space radiation analysis group (SRAG) which predicts, in real time, the exposure that might occur during a solar flare and terminates extravehicular activity if that is necessary.

It is impossible to predict how much radiation an astronaut will receive, although one can make an estimate of the probability of receiving a given amount. However, there is a chance that these limits will be exceeded in space. NASA regulations for limits of exposure, measured in REM's, call for 15 REM for 30 day exposure, 50 REM for annual exposure, 150 to 400 REM for exposure for men (more REM with increasing age), 100 to 300 for exposure for women (more REM with increasing age).

The radiation environment experienced by astronauts on Skylab and cosmonauts on Mir, as a function of their geographic location, has been studied by Badhwar (1997). Figure 16 shows figures from that paper, clearly illustrating that there is a major peak in radiation at the SAA, although there are minor peaks in other places. The 300 nGy/min peak over the SAA corresponds to about 30 mRad/min (30 mREM/min). Figure 8 shows the line of force cut by the shuttle, which is at 480 km altitude, orbital inclination 28.5 degrees. Although this line is steeply inclined to the vertical, the shuttle is well beyond the sensible atmosphere. The average value of radiation in the SAA is less than this peak value and is about half of this or 15 mREM/min. An astronaut would have to spend 1000 minutes (17 hrs) in the SAA during 30 days to reach his/her exposure limit. For a U.S. astronaut or Russian cosmonaut who stays in space for a year, perhaps on the space station, there is the 50 REM annual limit (note that is much less than 12 times the monthly limit). The radiation dosage of astronauts on the space station will be continuously monitored from Earth. During their flights to and from the moon the Apollo astronauts were lucky because they would have received a lethal dose of radiation if a strong solar event had occurred while they were in space—and such disturbances occur several times per year. A recent National Academy of Science study (National Academy of Science, 1996) showed that manned flights to Mars, because they are of long duration and outside the shielding of the magnetosphere, will not be possible, regardless of wishful thinking, until the shielding problem can be solved.

It is interesting to note that TOPEX, at an altitude of 1334 km, inclination 63 degrees, spent 42 minutes in the SAA over a 3-year period (14 min/yr). It received 2.3 rad per orbit or an accumulated dose of 12,000 rad/yr behind 1 gm/cm² Al shielding. This is far in excess of the lifetime limits for humans.
Figure 16. Distribution of dose rates as functions of latitude and of longitude for Skylab (Dec. 1973-Jan. 1974) and for Mir (Mar. 1995). Mir’s results were adjusted to Skylab’s to compensate for altitude difference (after Badhwar (1997)).

Shuttle flights with a 28.5 degree inclination do not pass through the entire SAA. Shuttle flights with 57 degree inclination do pass through the entire SAA but receive less radiation because they spend less time in the SAA.

THE FUTURE OF THE SOUTH ATLANTIC ANOMALY

Since the geomagnetic field is not static it is worthwhile investigating what the SAA and associated radiation hazard might be in the future. Figure 17 shows the secular change in total intensity of the field for the year 1995, as given by the IGRF. It shows that the greatest rate of decrease of the field is in the South Atlantic southwest of Capetown where the secular variation is more than 100 nT/yr. There is a secondary area of rapidly decreasing field in the western North Atlantic where it exceeds 80 nT/yr. Measurements over the past century, for much of the world, show essentially the same rate of change as for 1995 (Sabaka, et al., 1997). It is understood that data taken at some specific locations will differ from those shown by the spherical harmonic contours since the process of generating the spherical harmonic coefficients will possibly smooth over all values to fit a spherical harmonic expression.
Precisely extrapolating the rate of change of the geomagnetic field into the future is impossible because the reason for change is unknown. Here we can only draw upon what has happened in the past and assume that will continue for the next few decades. For some observatories, especially in the Northern Hemisphere, and in India, the slope of field strength vs time has shown a complete change of slope in the past. Many investigators have not tried to predict the field in the face of this erratic behavior. For the SAA region we can look at the records from 11 observatories that exist around the South Atlantic region. Those with records extending for more than 25 years are listed in table 2, and their locations shown in figure 18. Table 2 also gives the period of time for which there are yearly mean records and the rate of change of total field that those records show, calculated from a least-squares fit to a straight line. These data were obtained from NGDC. Figure 18 also shows the data for five observatories with the longer records and illustrates that the field is changing at a constant rate. Comparing these rates of change to figure 17 it is seen that the change at Hermanus (HER) and Syowa Base (SYO) are greater than figure 17 shows, but other observatories are about the same as the global map of figure 17 indicates.
Table 2. Secular variation of the total field at observatories around the South Atlantic.

<table>
<thead>
<tr>
<th>Code</th>
<th>Sec, Var. (nT/yr)</th>
<th>Years</th>
<th>Name</th>
</tr>
</thead>
<tbody>
<tr>
<td>HER</td>
<td>-110.4</td>
<td>1940–1999</td>
<td>Hermanus, S. Africa</td>
</tr>
<tr>
<td>KOU</td>
<td>-105.4</td>
<td>1958–1975</td>
<td>Surinam</td>
</tr>
<tr>
<td>LAS</td>
<td>-46.7</td>
<td>1965–1995</td>
<td>Las Acacias, Argentina</td>
</tr>
<tr>
<td>LQA</td>
<td>-47.4</td>
<td>1928–1987</td>
<td>La Quiaica, Argentina</td>
</tr>
<tr>
<td>MBO</td>
<td>-14.5</td>
<td>1952–1988</td>
<td>M’Bour, Senegal</td>
</tr>
<tr>
<td>PIL</td>
<td>-57.6</td>
<td>1905–1997</td>
<td>Pilar, Argentina</td>
</tr>
<tr>
<td>SYO</td>
<td>-101.3</td>
<td>1955–1997</td>
<td>Syowa Base, Antarctica</td>
</tr>
<tr>
<td>TRW</td>
<td>-77.9</td>
<td>1958–1995</td>
<td>Trelew, Argentina</td>
</tr>
<tr>
<td>TSU</td>
<td>-57.7</td>
<td>1952–1997</td>
<td>Tsumba, Namibia</td>
</tr>
<tr>
<td>TTB</td>
<td>-69.2</td>
<td>1960–1992</td>
<td>Tatuoca, Brazil</td>
</tr>
<tr>
<td>VSS</td>
<td>-25.2</td>
<td>1920–1997</td>
<td>Vassouras, Brazil</td>
</tr>
</tbody>
</table>

Annual Means From IGPP Monthly Means Web Database:

<table>
<thead>
<tr>
<th>Code</th>
<th>Sec, Var. (nT/yr)</th>
<th>Years</th>
<th>Name</th>
</tr>
</thead>
<tbody>
<tr>
<td>HER</td>
<td>-87.4</td>
<td>1992–1994.8</td>
<td>Hermanus, S. Africa</td>
</tr>
<tr>
<td>HER</td>
<td>-81.4</td>
<td>1995–1998.5</td>
<td>Hermanus, S. Africa</td>
</tr>
<tr>
<td>LAS</td>
<td>-65.1</td>
<td>1992–1994.6</td>
<td>Las Quiaica, Argentina</td>
</tr>
<tr>
<td>TSU</td>
<td>-59.6</td>
<td>1992.2–1994.9</td>
<td>Tsumba, Namibia</td>
</tr>
<tr>
<td>TSU</td>
<td>-65.6</td>
<td>1995–1998.5</td>
<td>Tsumba, Namibia</td>
</tr>
</tbody>
</table>
Figure 18. Secular variation of total field as observed at longer operating geomagnetic observatories around the South Atlantic. The location of the observatories is shown in the upper left, with secular change from figure 17 superimposed. A least-squares fit of the observed data to a straight line is shown with the slope of that line indicated.

Also in table 2 we give rates of change calculated from monthly mean values, for some observatories, with the data obtained from IPGP Paris (http://obsmag.ipgp.jussieu.fr/AM-rqobsmag.html). The monthly mean data only extends for the last 8 years and does not follow so smooth a curve as the yearly means. The slope of these curves was also determined by a least-square fit to a straight line and those slopes are given in table 2. It is seen that the monthly means for Hermanus give values closer to the values in figure 17, than does the yearly mean. This suggests that, in the last few years, the field at Hermanus and Syowa Base may have been decreasing faster than at earlier times.

For periods of time longer than shown in table 2 one might think of the geologic record of the paleomagnetic field. Kreer et al. (1983) reported on a paleomagnetic study of Argentine lake sediments from each of three adjacent lakes located about 600 km northwest of the Trelew geomagnetic observatory. The sediments from one of the lakes extended back 14,000 years, but sediments from the other lakes did not extend back so far. The declination and inclination records from the three lakes could be stacked to produce a record of the direction of the field for the last 6,000 years. The probable error of these directions is high although they show a large excursion to the west and up about 500 years ago.
The authors show intensity and susceptibility profiles from eight cores from the three lakes but were unable to make a stacked intensity profile because of the high variability from lake to lake. From the figures shown the intensity seems to have varied erratically by 10 to 20 percent with numerous short excursions of intensity, which can easily be artifacts. The authors point out that these are the first paleomagnetic measurements from South America and much remains to be done before measurements like this have high reliability. A search of the literature did not reveal any paleomagnetic measurements from lakes from South Africa.

The Deep Sea Drilling Project and the International Program of Ocean Drilling have drilled a number of holes in the South Atlantic Ocean floor. Magnetic measurements are usually made on the deep-sea core material but, contrary to cores from the North Atlantic, no comprehensive report of paleomagnetic intensities has been made for the South Atlantic. In any case, the sedimentation rate in the oceans is usually much slower than in lakes, and time scale so compressed, they are not likely to yield information relevant to the last few hundred years.

It is clear that the spherical harmonic model of the field and its secular change smoothes the field and cannot be expected to give exactly correct values at all points on the Earth. For purposes of this report we will assume that the rate of change as illustrated in figure 17 is correct enough to be used to extrapolate tens to a few hundred years in the future. The calculation is carried out by updating each of the spherical harmonic coefficients, according to their first-time derivative, calculating the X, Y, and Z components of the field and then calculating B from X, Y, and Z.

Figure 19 shows the total field for the year 1995 extrapolated to the years 2025, 2050, and 2100. The principle feature of the geomagnetic field is the change in the SAA. Each few decades it gets larger and the field within it gets weaker. A secondary low shows southwest of Capetown.

One may define the extent of the SAA by the areal extent of the high dose rates for shuttle and Mir (figure 16). The high dose rates extend from approximately 15 S to 40 S and from 20 W to 70 W at 480 km altitude. This corresponds roughly to the 25,000 nT contour on the Earth’s surface. Using this contour to identify the SAA, in the year 2100 the SAA will cover most of South America, the southern part of Africa and the South Atlantic Ocean south of 25 S to the Scotia Sea and Antarctica. The size of the SAA will have increased by a factor of about 4. This suggests that the radiation hazard to humans in space may be increased by a corresponding amount. This will mean that astronauts can spend correspondingly less time in space.

In 1995, the focus of the SAA was clearly over the coast of Brazil. In 2100 there will be two foci: one in northern Brazil and a second and deeper one southwest of Capetown in the South Atlantic Ocean.

Radiation levels with this new SAA will be different from the present for two reasons: (1) because the field is less and the mirror points of particles lower with more being lost to the atmosphere (greater bounce cone loss), and (2) as mentioned before, the area of this low field will be greater than it is now. The first effect would probably reduce the level of radiation, while the second effect would tend to distribute the radiation over a larger area with the integrated effect increasing the radiation hazard. It is tacitly assumed that at present the radiation belts are in approximate equilibrium with the input of particles from
Figure 19. Total field contour map (in nT) for the years 2025 (upper), 2050 (center), and 2100 (lower) extrapolated from IGRF 95.
galactic and solar sources balancing the number of particles lost by atmospheric scattering. In the future the sources would be the same as today while the losses would be greater. This could seriously deplete the particle population of the radiation belts. Able and Thorne, 1999, have made progress in quantifying the factors which create a radiation hazard over the SAA in 1995. A similar analysis for a possible future field is needed with attention given to the balance between the source supply and the atmospheric loss.

![Figure 20. Total field contour map (in nT) for the year 2235, extrapolated from IGRF 95.](image)

It is interesting to continue to extrapolate the field 240 years in the future, to the year 2235. The focus in northern South America intensifies but the field strength approaches zero in the South Atlantic (figure 20). Naturally, all components of the field (X, Y, H, and Z) must approach zero if the total field approaches zero. This does not constitute a global reversal of the geomagnetic field but is a phenomenon confined to the South Atlantic, at that time.

Using the 1995 IGRF secular change in inclination one may calculate how the geomagnetic equator (where the field is horizontal) will change with time. Figure 21 shows that the geomagnetic equator in the Atlantic will move north by 10 to 15 degrees in the next 100 years and by 15 to 30 degrees in the next 240 years. In other parts of the world the change is not great, although it may change long-range high-frequency radio transmissions.
Figure 21. Location of geomagnetic equator (where dip is zero) for years 1995, 2100, and 2235 all based on IGRF 95.
REFERENCES


Roederer, J.G., Dynamics of geomagnetically trapped radiation, Springer-Verlag, New York, 166 pp., 1970.


APPENDIX

This computer code, in True Basic, will plot the field strength B or altitude h vs latitude, with mirror points indicated (similar to figure 11) or it can be modified to make a number of the other plots shown in this report. Note that most of this code, after statement 2400, is the subroutine GEOMAGJ to calculate the field components for a given latitude, longitude, and altitude. The file "IGRF95" is called for in the subroutine and a sample listing of values in this file is given following the listing of the main program. Also note that a large number of lines is to calculate the location of the mirror points and plot them. This part can be skipped by inserting a "Go to" statement and this will materially reduce the number of lines.

10 !Computer code plot B or h vs lat with mirror pts.
20 !Written in TrueBasic, Nov 1998, JRH
30 !stores graphics on clipboard
40 set zonewidth 6
50 OPTION NOLET
60 OPTION ANGLE degrees
70 DIM h(200),lt(200),ln(200),b(200),dp(200),dc(200) !for pts along line of force
80 DIM bm(100) !for field values at mirror points
90 DIM hms(100),ltms(100),lnms(100),bms(100) !for variables at mirror pts in south hemisphere
100 DIM hmn(100),ltmn(100),lnmn(100),bmn(100) !for variables at mirror pts in north hemisphere
110 !OPEN#4: NAME OUTFILE$,CREATE NEW, ORG TEXT
120 ! ***** initialize for plotting *****
130 LIBRARY "PictLib*"
140 LIBRARY "MacTools*"
150 call Copy_clipboard
160 open#2: screen .05,.4,.05,.6 !to make "trim" fit
170 xmin=-55
180 xmax=55
190 ymin=0
200 ymax=10000
210 set window xmin,xmax,ymin,ymax
220 ! **** draw box ****
230 plot lines: xmin,ymin;xmin+1,ymin
240 plot lines: xmax,ymin;xmax,ymin
250 plot lines: xmax,ymax;xmax,ymax
260 plot lines: xmin,ymax;xmin,ymax
270 ! **** plot tics on y axis ****
280 for n=ymin to ymax step 1000
290 plot lines: xmin,n;xmin+1,n
300 plot lines: xmax-1,n;xmax,n
310 next n
320 ! **** plot tics on x axis ****
330 for n=xmin to xmax step 5
340 plot lines: n,ymin,n,ymin+100
350 plot lines: n,ymax-100,n,ymax
360 next n
370 ! **** calculate and plot field line ****
380 !
390  dlat=1  !lat increment along path, deg
400  for flonk=0 to 0 !step 30  !this steps longitude of origin
410  for nk=-50 to -30 step 10 !this steps latitude of origin
420  flatmin=nk  !starting lat, deg
430  flatmax=-nk  !final lat, deg
440  altl=0  !starting alt, km
450  flat1=flatmin
460  flon1=flonk
470  k=1
480  flat=flat1
490  alt=altl
500  flon=flon1
510  !
520  ***** stepping along path begins here *****
530  !
540  CALL GEOMAGJ (alt,flat,flon,t,dip,dec) !get value of t,dip, dec
550  h(k)=alt
560  lt(k)=flat
570  ln(k)=flon
580  b(k)=t
590  dp(k)=dip
600  dc(k)=dec
610  k=k+1
620  dalt=-110*dlat*tan(dip)
630  alt=alt+dalt
640  alt2=alt
650  if alt2<0 then go to 1050  ! if alt goes into ground we have gone too far
660  flat2=flat1+dlat
670  dlon=110*dlat*tan(dec/60)
680  flon2=flon1+dlon
690  go to 1000
700  if flon1>180 then  ! change from flon,flat to x,y
710  x1=flon1-360
720  go to 810
730  else
740  if flon1<-180 then
750  x1=flon1+360
760  go to 810
770  else
780  x1=flon1
790  end if
800  end if
810  y1=flat1
820  if flon2>180 then
830  x2=flon2-360
840  go to 930
850  else
860  if flon2<-180 then
870  x2=flon2+360
880  go to 930
890  else
900  x2=flon2
910  end if
920  end if
930  y2=flat2

The Geomagnetic Field...
if x2>xmax or x2<xmin then go to 1000 !out of bounds
if y2>ymax or y2<ymin then go to 1000 !out of bounds
base=abs(x1)+abs(x2)
if base>=358 then go to 1000 !to prevent line from jumping from one side to the other
if alt2<600 then box circle x1-.5,x1+.5,y1-.5,y1+.5 !emphasis line is alt is low
plot lines: x1,y1;x2,y2 !plot line segment
flat1=flat2 !advance points
flon1=flon2
alt1=alt2
go to 480

***** plot b(n) or h(n) vs lt(n) from n=1 to kmax *****

kmax=k-1
xx1=-lt(1)
yy1=h(1)
for n=2 to kmax
xx2=-lt(n)
yy2=h(n)
plot lines: xx1,yy1;xx2,yy2
xx1=xx2
yy1=yy2
next n

***** find values for dip equator *****

for k=l to kmax-1 !this loop ends on line 1470
if dp(k)<=0 and dp(k+1)>0 then
factor=abs(dp(k)/(dp(k+1)-dp(k)))
if h(k+1)<=h(k) then
heq=h(k)-factor*(abs(h(k+1)-h(k))) !alt decreasing or no change
else
heq=h(k)+factor*abs((h(k+1)-h(k))) !alt increasing
end if
lneq=ln(k)+factor*abs((ln(k+1)-ln(k))) !long increasing
else
lneq=ln(k)+factor*abs((ln(k+1)-ln(k))) !long is increasing
end if
if b(k+1)<=b(k) then
beq=b(k)-factor*abs((b(k+1)-b(k))) !b is decreasing or no change
else
beq=b(k)+factor*abs((b(k+1)-b(k))) !b is increasing
end if
if dp(k+1)<=dp(k) then
dpeq=dp(k)+factor*abs((dp(k+1)-dp(k))) !dip is increasing
else
dpeq=dp(k)-factor*abs((dp(k+1)-dp(k))) !dip is decreasing or no change
end if
if dc(k+1)<=dc(k) then
dceq=dc(k)+factor*abs((dc(k+1)-dc(k))) !declination is increasing
else
dceq=dc(k)-factor*abs((dc(k+1)-dc(k))) !declination is decreasing or no change
end if
next k

***** calc south and north mirror pts for eq pitch angle alpha *****

for alpha=80 to 30 step -10  !this loop ends at statement 2140

bm(alpha)=beq/(sin(alpha)*sin(alpha)) !value of bm for this alpha

***** search for and plot mirror pts, first south of eq. then north *****

for k=l to kmax-1 !start following along line of force, ends at statement 2130
  if dp(k)<=0 then !south of mag equator
    if b(k)>=bm(alpha) and b(k+1)<bm(alpha) then  !value found, calculate
      factor=abs((bm(alpha)-b(k))/(b(k+1)-b(k)))
      if h(k+1)<=-h(k) then
        hms(alpha)=h(k)-factor*abs(h(k+1)-h(k))  !h is decreasing
      else
        hms(alpha)=h(k)+factor*abs(h(k+1)-h(k))  !h is increasing
      end if
      ltms(alpha)=lt(k)+factor*(lt(k+1)-lt(k))
      if ln(k+1)<=ln(k) then
        lnms(alpha)=ln(k)-factor*abs(ln(k+1)-ln(k))  !ln is decreasing
      else
        lnms(alpha)=ln(k)+factor*abs(ln(k+1)-ln(k))  !ln is increasing
      end if
      bms(alpha)=b(k)-factor*abs(b(k+1)-b(k))  !b is decreasing
      else
        bms(alpha)=b(k)+factor*abs(b(k+1)-b(k))  !b is increasing
      end if
    end if
    xc=ltms(alpha)-lx  !plot lat, h box around south mirror pt
    yc=hms(alpha)-50
    y2=hms(alpha)+50
    if hms(alpha)<>0 then box circle -x2,-xl,yc,y2 !draw box 1
    else !value not found
      go to 2130 !continue search south of mag equator
    end if
  else !north of mag equator
    if b(k)<bm(alpha) and b(k+1)>=bm(alpha) then
      factor=abs((bm(alpha)-b(k))/(b(k+1)-b(k)))
      if h(k+1)<=-h(k) then
        hmn(alpha)=h(k)-factor*abs(h(k+1)-h(k))  !h is decreasing
      else
        hmn(alpha)=h(k)+factor*abs(h(k+1)-h(k))  !h is increasing
      end if
      ltmn(alpha)=lt(k)+factor*(lt(k+1)-lt(k))
      if ln(k+1)<=ln(k) then
        lnmn(alpha)=ln(k)-factor*abs(ln(k+1)-ln(k))  !ln is decreasing
      else
        lnmn(alpha)=ln(k)+factor*abs(ln(k+1)-ln(k))  !ln is increasing
      end if
      bmn(alpha)=b(k)-factor*abs(b(k+1)-b(k))  !b is decreasing
      else
        bmn(alpha)=b(k)+factor*abs(b(k+1)-b(k))  !b is increasing
      end if
    end if
    xc=ltmn(alpha)-lx  !plot lat, h box around north mirror pt
    yc=hms(alpha)-50
    y2=hms(alpha)+50
    if hms(alpha)<>0 then box circle -x2,-xl,yc,y2 !draw box 1
    else !value not found
      go to 2130 !continue search south of mag equator
    end if
end if !end calc for south mirror pt

The Geomagnetic Field... 31
2040    x2=ltmn(alpha)+1
2050    y1=hmn(alpha)-50
2060    y2=hmn(alpha)+50
2070    if hmn(alpha)<>0 then box circle -x2,-x1,y1,y2 !draw box 2
2080    else !value not found, continue north of mag equator
2090    go to 2130
2100    end if !ends calc north mirror pt
2110    go to 2140 !jump out of this loop and finish this search
2120    end if !ends test of dip
2130    next k !ends checking line of force for this value of alpha
2140    next alpha
2150    for alpha= 80 to 40 step -10 !plot marks for mirror pts
2160    if hms(alpha)<0 then
2170    xx1=ltms(alpha)-.5 !plot lat, b box around south mirror pt
2180    xx2=ltms(alpha)+.5
2190    yy1=bms(alpha)-300
2200    yy2=bms(alpha)+300
2210    !box circle xx1,xx2,yy1,yy2 !draw box 3
2220    else
2230    go to 2250
2240    end if
2250    if hmn(alpha)<0 then
2260    xx3=ltmn(alpha)-.5 !plot lat, b box around north mirror pt
2270    xx4=ltmn(alpha)+.5
2280    yy3=bmn(alpha)-300
2290    yy4=bmn(alpha)+300
2300    !box circle xx3,xx4,yy3,yy4 !draw box 4
2310    else
2320    go to 2340
2330    end if
2340    next alpha
2350    next nk
2360    next flonk
2380    call Copy_done
2390    close #2
2400 END
2410 ! ***** END OF PROGRAM *****
2420 !
2430 ! ***** subroutine to calculate field components *****
2440 SUB GEOMAGJ(alt,flat,flon,t,dip,dec)
2450 ! subroutine GEOMAGJ to calc geomag field components from coefs
2460 ! read in coefs and sec change, after sub getshc of NGDC
2470 option nolet
2480 option angle degrees
2490 dim gh(224)
2500 dim dgh(224)
2510 ddate=2100.0
2520 infile$="IGRF95"
2530 factor=ddate-1995.0
2540 nmax=10
2550 open #3:name infile$
2560 i = 0
2570 for nn = 1 to nmax
2580 for mm = 0 to nn
2590 input#3:m,n,g,h,dg,dh
The Geomagnetic Field...
else if (90. + flat) < .001 then
aa = -89.999
! 300 ft from s. pole
else
aa = flat
end if
clat = cos (aa)
sl(1) = sin (flon)
c1(1) = cos (flon)
X_---O.
dx=O.
y=O.
dy=O.
z=O.
dz=O.
sd = O.
cd = 1.
n=0
m=1
npq = (nmax * (nmax + 3)) / 2
if igdgc = 1 then
aa = a2 * clat * clat
bb = b2 * slat * slat
cc = aa + bb
dd = sqr (cc)
r = sqr (elevkm * (elevkm + 2. * dd)+(a2 * aa + b2 * bb) / cc)
cd = (elevkm + dd) / r
sd = (a2 - b2) / dd * slat * clat / r
aa = slat
clat = slat * cd - clat * sd
end if
ratio = erad / r
aa = sqr (3.)
p(1) = 2. * slat
p(2) = 2. * clat
p(3) = 4.5 * slat * slat - 1.5
p(4) = 3. * aa * clat * slat
q(1) = -clat
q(2) = slat
q(3) = -3. * clat * slat
q(4) = aa * (slat * slat - clat * clat)
for k = 1 to npq ! this loop ends at statement 4110
if n < m then
m = 0
n = n + 1
rr = ratio^n(n + 2)
f n = n
end if
fm = m
if k >= 5 then
if m = n then
aa = sqr (1. - .5 / fm)
j = k - n - 1
p(k) = (1. + 1. / fm) * aa * clat * p(j)
q(k) = aa * (clat * q(j) + slat / fm * p(j))
sl(m) = sl(m-1) * cl(1) + cl(m-1) * sl(1)
cl(m) = cl(m-1) * cl(1) - sl(m-1) * sl(1)

else

aa = sqrt (fn * fn - fm * fm)
bb = sqrt ((fn - 1.)^2 - fm * fm) / aa
cc = (2. * fn - 1.) / aa
i = k - n
j = k - 2 * n + 1

p(k) = (fn + 1.) * (cc * slat / fn * p(i) - bb/(fn - 1.) * p(j))
q(k) = cc * (slat * q(i) - clat / fn * p(i)) - bb * q(j)

end if
end if

aa = rr * gh(l)
da= rr * dgh(l)

if m = 0 then

x = x + aa * q(k)
dx= dx + daa * q(k)
z = z - aa * p(k)
dz = dz - daa * p(k)

l = l + 1

else

bb = rr * gh(l+1)

dbb = rr * dgh(l+1)

cc = aa * cl(m) + bb * sl(m)
dcc = daa * cl(m) + dbb * sl(m)
x = x + cc * q(k)
dx = dx + dcc * q(k)
z = z - cc * p(k)
dz = dz - dcc * p(k)

if clat > 0. then

y = y + (aa * sl(m) - bb * cl(m)) * fm * p(k)/(fn + 1.) * clat

dy = dy + (daa * sl(m) - dbb * cl(m)) * fm * p(k)/(fn + 1.) * clat

else

y = y + (aa * sl(m) - bb * cl(m)) * q(k) * slat

dy = dy + (daa * sl(m) - dbb * cl(m)) * q(k) * slat

end if

l = l + 2

end if

m = m + 1

next k

x = x + factor * dx

y = y + factor * dy

z = z + factor * dz

aa = x

x = x * cd + z * sd

z = z * cd - aa * sd

h=sqrt(x*x+y*y)
sn=0.0001

rad=57.29577951
t=sqrt(x*x+y*y+z*z)

if x=0 and y=0 then dec=0

if x=0 and y<0 then dec=-90

if x=0 and y>0 then dec=90

The Geomagnetic Field...
The following is a listing of the contents of file "IGRF95" called for in the subroutine GEOMAGJ. Only coefficients through nmax=5 are illustrated here. The sequence is as shown below.

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The Geomagnetic Field and Radiation in Near-Earth Orbits

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This report shows, in detail, how the geomagnetic field interacts with the particle flux of the radiation belts to create a hazard to spacecraft and humans in near-Earth orbit. It illustrates the geometry of the geomagnetic field lines, especially around the area where the field strength is anomalously low in the South Atlantic Ocean. It discusses how the field will probably change in the future and the consequences that may have on hazards in near space.