

Potential Operating Orbits for the SAFE-400

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Abstract. Safety must be ensured during all phases of space fission system design, development, fabrication, launch, operation, and shutdown. One potential space fission system application is fission electric propulsion (FEP), in which fission energy is converted into electricity and used to power high efficiency ($I_{sp} > 3000s$) electric thrusters. For these types of systems it is important to determine which operational scenarios ensure safety while allowing maximum mission performance and flexibility. Space fission systems are essentially non-radioactive at launch, prior to extended operation at high power. Once high power operation begins, system radiological inventory steadily increases as fission products build up. For a given fission product isotope, the maximum radiological inventory is typically achieved once the system has operated for a length of time equivalent to several half-lives. After that time, the isotope decays at the same rate it is produced, and no further inventory builds in. For an FEP mission beginning in Earth orbit, altitude and orbital lifetime increase as the propulsion system operates. Two simultaneous effects of fission propulsion system operation are thus (1) increasing fission product inventory and (2) increasing orbital lifetime. Phrased differently, as fission products build up, more time is required for the fission products to naturally convert back into non-radioactive isotopes. Simultaneously, as fission products build up, orbital lifetime increases, providing more time for the fission products to naturally convert back into non-radioactive isotopes. Operational constraints required to ensure safety can thus be quantified.

I. INTRODUCTION

The fission process was first reported in 1939, and in 1942 the world's first man-made self-sustaining fission reaction was achieved. Creating a self-sustaining fission chain reaction is conceptually quite simple. All that is required is for the right materials to be placed in the right geometry - no extreme temperatures or pressures required - and the system will operate. Since 1942 fission systems have been used extensively by governments, industry and universities. Fission systems operate independently of solar proximity or orientation, and are thus well suited for deep space or planetary surface missions. In addition, the fuel for fission systems (highly enriched uranium) is essentially non-radioactive, containing 0.064 curies/kg. This compares quite favorably to current nuclear systems (Pu-238 in radioisotope systems contains 17,000 curies/kg) and certain highly futuristic propulsion systems (tritium in D-T fusion systems would contain 10,000,000 curies/kg). An additional comparison is that at launch a typical space fission propulsion system would contain an order of magnitude less onboard radioactivity than did Mars Pathfinder's Sojourner Rover, which used radioisotopes for thermal control. The primary safety issue with fission systems is avoiding inadvertent system start - addressing this issue through proper system design is straightforward. The energy density of fission is seven orders of magnitude greater than that of the best chemical fuels, and if properly utilized is more than adequate for enabling rapid, affordable access to any point in the solar system.

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II. THE 400 KILOWATT THERMAL SAFE AFFORDABLE FISSION ENGINE (SAFE-400)

NASA's reference Phase 1 space fission electric propulsion system is based on the 400 kWt Safe Affordable Fission Engine (SAFE-400) reactor. Hardware-based research and development related to Phase 1 space fission system development has been ongoing since 1996 (Houts, 1997). In addition to early module tests, a 30 kWt SAFE core has been fabricated and tested, and a 100 kWt core is currently in fabrication. The next step after fabrication and testing of the 100 kWt core is fabrication and testing of a 400 kWt core that is nearly flight-prototypic. Design details are being added to the 400 kWt core, and fabrication should begin in 2003. While SAFE performance must be adequate to enable missions of interest, the emphasis of Phase 1 system design and development is on safety, affordability, and schedule. Key features of Phase 1 systems include a high level of testability, utilization of established technology, and utilization of existing / operational facilities. Details of recent efforts at NASA MSFC related to the SAFE are given in Van Dyke, 2002. Details of the reference Phase 1 400 kWt design (SAFE-400) are given in Poston, 2002. A comparison of Phase 1 system options is given in Houts, 2002. A picture of a coupled SAFE-30 / Stirling engine test is shown in Figure 1. A picture of a control zone test of a SAFE-100 mock-up is shown in Figure 2.

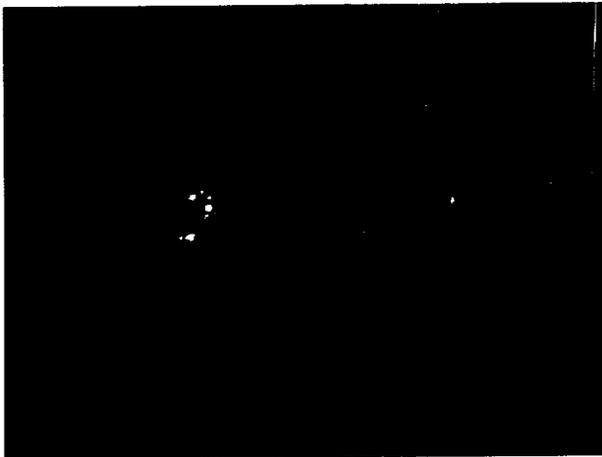


FIGURE 1. SAFE-30 Providing Thermal Power to a Stirling Engine, Resulting in Electricity Production.

III. ORBITAL LIFETIME AS A FUNCTION OF ALTITUDE

Orbital lifetime as a function of altitude was calculated for three spacecraft mass-to-frontal area ratios: 82 kg/m², 250 kg/m², and 1000 kg/m². Spacecraft altitudes of 550 to 700 km were examined. The range was chosen based on Space Shuttle capabilities and atmospheric drag. The

payload capability (including attachment hardware) of the space shuttle to a 550 km orbit exceeds 13,500 kg. This payload capability is significantly greater than that required by all but the most ambitious Phase 1 FEP missions. At 700 km atmospheric drag is extremely low (effectively zero above 800 km); however, the space shuttle is not capable of reaching that orbit. The code "LIFTIM" (Alford, 1974) was used to perform the calculations. An initial orbit insertion date of 31 December 2011 was assumed, and predicted fluctuations of the Earth's atmosphere were taken into account. Results of the calculations are shown in Figure 3. The calculations are conservative in that the launch date is chosen to be just prior to peak atmospheric density at altitude. Atmospheric density values two standard deviations above the predicted value (+2s) and two standard deviations below the predicted value (-2s) are plotted.

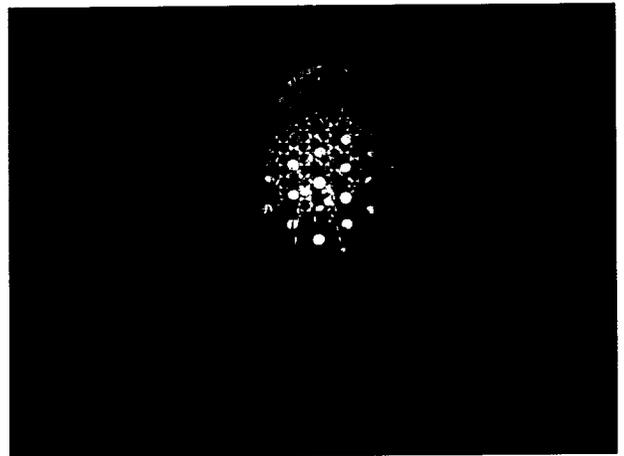


FIGURE 2. SAFE-100 Mockup Control Zone Test.

As shown in Figure 3, an 82 kg/m² spacecraft (e.g. an 8200 kg FEP vehicle with 100 m² of frontal area) placed in a 550 km orbit is estimated to require 20 months to re-enter, assuming a 12/31/2011 launch and a +2s (worst-case) atmosphere. The same spacecraft placed in a 600 km orbit would require over nine years to re-enter, given the same assumptions. Figure 2 also shows that at 250 kg/m² (e.g. 10,000 kg, 40 m² area) orbital lifetime for missions starting at 550 km exceeds ten years. Orbits of 700 km or higher all have lifetimes much greater than 10 years (for the 82, 250, and 1000 kg/m² spacecraft analyzed).

Given a reasonable system specific power and specific impulse, increasing the orbital altitude of an FEP system from 550 to 600 km will require about one day of full-power operating time. If the expelled propellant mass is considered negligible, then for the 82 kg/m² case one day of operation effectively increases the orbital lifetime by

over seven years. For spacecraft configurations that result in greater than 82 kg/m^2 , the orbital lifetime would be

increased even more.

**NEP Demo Orbital Lifetime, $\pm 2\sigma$ Atmosphere
Cd = 2.1, Mass/Area = 1000, 250, 82 kg/m²**

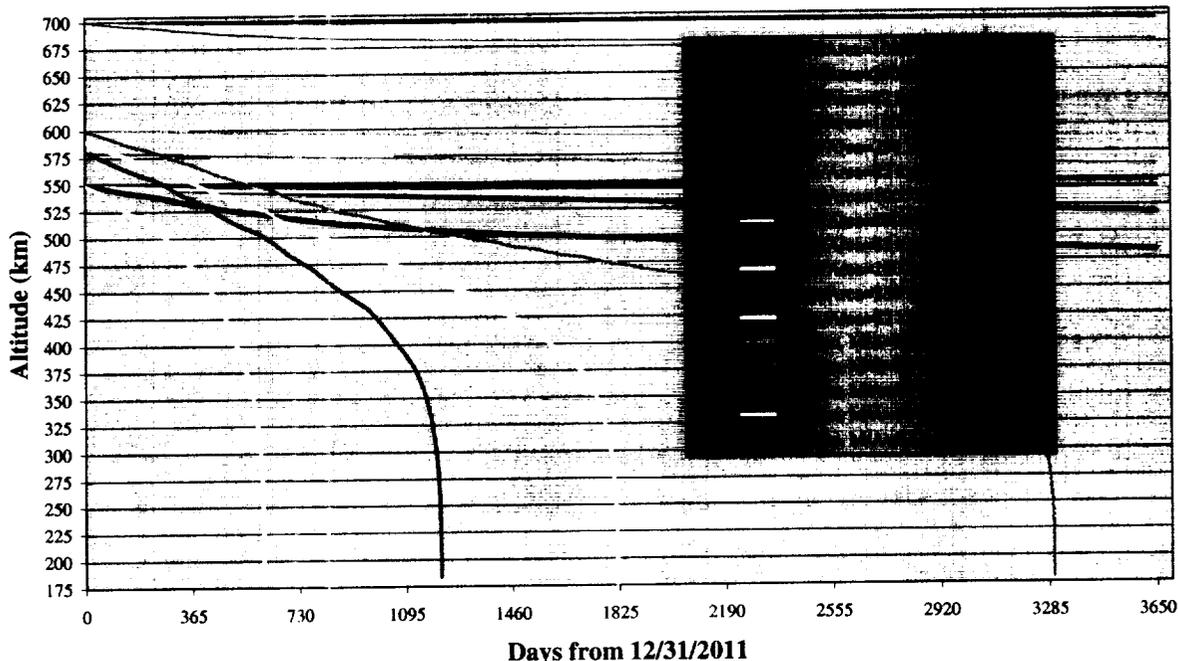


FIGURE 3. Orbital Lifetime as a Function of Altitude for Three Spacecraft Mass-to-Area Ratios.

IV. RADIOLOGICAL INVENTORY AS A FUNCTION OF LIFETIME

Radiological inventory of the SAFE-400 at launch is on the order of 10 Curies (primarily due to U-234). Radiological inventory as a function of time after shutdown was calculated for the SAFE-400 and for spacecraft previously launched by NASA using the code "MONTEBURNS" (Poston, 1999). Radiological inventory as a function of time after shutdown for various SAFE-400 operating times and for various spacecraft currently in Earth orbit is shown in Figure 4. As shown in Figure 4, SAFE-400 radiological inventory decreases rapidly after shutdown, and more rapidly than the inventory in Pu-238 powered spacecraft. Seven days of full power FEP operating time is typically sufficient to raise the orbital altitude of a spacecraft by several hundred kilometers, placing the spacecraft above the point where atmospheric drag is significant for most reasonable starting orbits. As shown in Figure 3, following one week of full-power operation SAFE-400 radiological inventory drops below 100 Curies within 3 years. For full-power

operation of 1 day, radiological inventory drops below 100 Curies in about a year. The actual potential radiological hazard from a reentry would depend not only on radiological inventory, but also on radiation type and system geometry following reentry

V. OBSERVATIONS

All potential civilian FEP missions currently under discussion would not return to low Earth orbit following extended operation. For these missions, preliminary calculations indicate that strict mission safety requirements are met if FEP operation begins above an orbital altitude of about 500 km. For these scenarios, system radiological inventory during a worst-case failure / re-entry is comparable to that at system launch. If more detailed calculations sustain this observation, then deploying FEP systems directly out of the space shuttle cargo bay (no additional stages required) may be a viable option. Deployment directly from the space shuttle has several potential advantages, including the extremely high reliability of the space shuttle (>99.8%), the large volume

of the shuttle cargo bay, the shuttle's delivered payload mass capability, the presence of astronauts to help ensure proper FEP system deployment and start, and the ability to return the FEP system to earth if desired. If a decision

is made against using the space shuttle to deploy the FEP vehicle, it will still be important to determine the desired range of initial operating orbits to ensure mission safety while optimizing mission performance and flexibility.

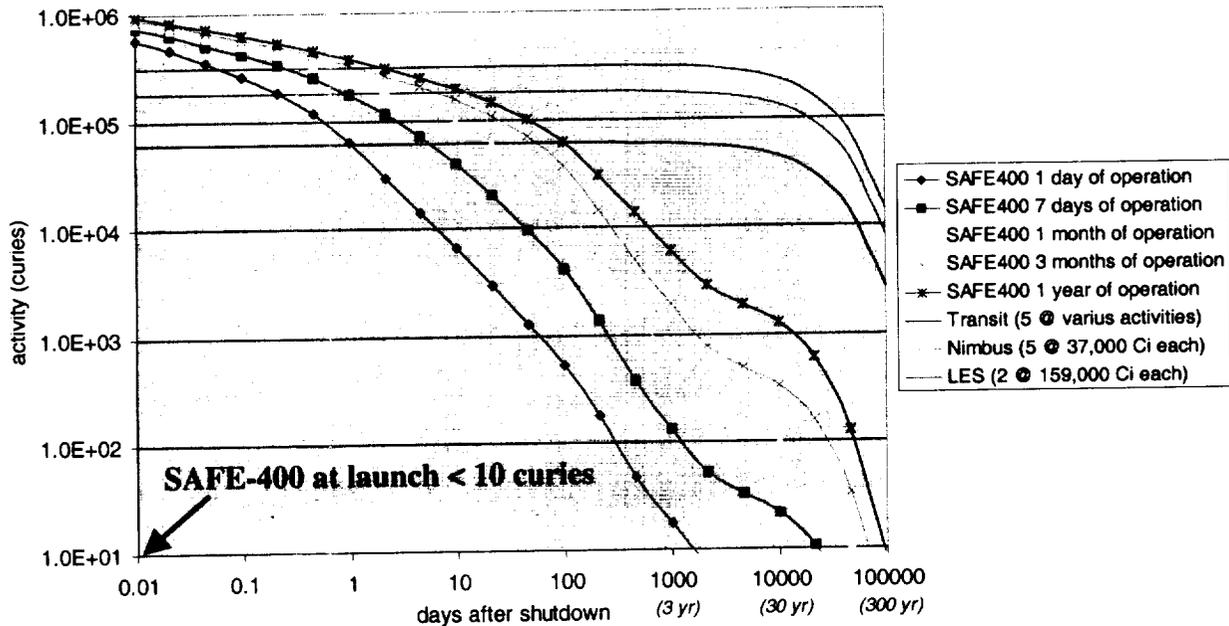


FIGURE 4. Radiological Activity of Currently Orbiting Spacecraft and the SAFE-400 as a Function of Time.

VI. RECOMMENDATIONS FOR FUTURE RESEARCH

Mission options for potential NASA Phase I FEP systems should continue to be explored. More detailed mission design and analysis should be performed to define missions that ensure safety while providing maximum mission performance and flexibility. The benefits and issues associated with using the space shuttle to deploy FEP systems should be further investigated. Increased FEP system definition will add fidelity to mission analyses.

ACKNOWLEDGMENTS

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