“BIMODAL” NUCLEAR THERMAL ROCKET (BNTR) PROPULSION
FOR FUTURE HUMAN MARS EXPLORATION MISSIONS

Stan Borowski
National Aeronautics and Space Administration
Glenn Research Center
Cleveland, Ohio
Artificial Gravity “Bimodal” NTR Crew Transfer Vehicle (CTV) for Mars DRM 4.0 (1999)
• During short, high thrust propulsion phase, each BNTR produces ~340 MWt and ~15 klbf of thrust
• During long, power generation phase, each BNTR operates in “idle mode” producing just ~150 kWt
• A Brayton conversion unit on each BNTR produces up to 25 kW_e to enhance stage capabilities
Rover/NERVA* Program Summary (1959-1972)

- 20 Rocket/reactors designed, built and tested at cost of ~ $1.4 billion
- Engine sizes tested
  - 50-250 klf
- H$_2$ exit temperatures achieved
  - 2,350-2,550 K (Graphite fuel)
- $I_{sp}$ capability
  - 825-850 sec (hot bleed cycle)
- Burn duration
  - 62 mins. (NRX-A6 -- single burn)
  - >4 hrs. (NRX-XE -- 28 burns) (accumulated)
- Engine thrust-to-weight
  - ~3 for 75 klf NERVA

"Open Air" testing at Nevada Test Site

*NERVA: Nuclear Engine for Rocket Vehicle Applications

NERVA program experimental engine (XE) demonstrated 28 startup/shutdown cycles during tests in 1969.
Nuclear Thermal Rocket (NTR) Propulsion

What’s New?

Then (Rover/NERVA: 1959–72)

- **Engine sizes tested**
  - 50–250 klbf

- **H₂ exit temps achieved**
  - 2,350–2,550K (Graphite)

- **Isp capability**
  - 825–850 sec (hot bleed)

- **Engine thrust-to-weight**
  - ~3 for 75 klbf NERVA

- **Testing (Rover/NERVA)**
  - “Open Air” exhaust at Nevada test site

Now

- **“Current” focus is on smaller NTR sizes**
  - 5–15 klbf (Code S science–humans)

- **Higher temp. fuels being developed**
  - 2,700K (Composite),  2,900K (Cermet) and ~3,100K (Ternary Carbides)

- **Isp capability**
  - 915–1005 sec (expander cycle)

- **Advances in chemical rockets/materials**
  - ~2–6 for small NTR designs

- **Small NTR allows full power testing in**
  - “Contained Test Facility” at INEL with “scrubbed” H₂ exhaust

Smaller, Higher Performance

Easier to test

Environmentally “Green”

For Public Acceptance

‘Propelling Us to New Worlds’
Nuclear Thermal Rocket (NTR) Propulsion
-- Key Technology / Mission Features --

- NTR engines have negligible radioactivity at launch / simplifies handling and stage processing activities at KSC
  - < 10 Curies / 3 NTR Mars stage vs ~400,000 Curies in Cassini's 3 RTGs
- High thrust / Isp NTR uses same technologies as chemical rockets
- Short burn durations (~25-50 mins) and rapid LEO departure
- Less propellant mass than all chemical implies fewer ETO launches
- NTR engines can be configured for both propulsive thrust and electric power generation -- "bimodal" operation
- Fewest mission elements and much simpler space operations
- Engine size aimed at maximizing mission versatility
  -- robotic science, Moon, Mars and NEA missions
- NTR technology is evolvable to reusability and "in-situ" resource utilization (e.g., LANTR -- NTR with LOX "afterburner" nozzle)
“Bimodal” NTR Cargo & Crew Transfer Vehicles for 1999 Mars Design Reference Mission (DRM) 4.0

6 - “80 t” SDHLVs plus Shuttle for Crew & TransHab Delivery

2011 Cargo Mission 1
Habitat Lander
IMLEO= 131.0 t

Optional “In-Line” LH₂ Tank (if needed)

2011 Cargo Mission 2
Cargo Lander
IMLEO= 133.7 t

2014 Piloted Mission
Artificial Gravity
Crew Transfer Vehicle
IMLEO= 166.4 t
Modular “Bimodal” NTR Transfer Vehicle Design for Mars Cargo and Piloted Missions

**Bimodal NTR:** High thrust, high $I_{sp}$ propulsion system utilizing fissioning U$^{235}$ produces thermal energy for propellant heating and electric power generation enhancing vehicle capability

**Vehicle Characteristics**
- Versatile design
- “Bimodal” stage produces 50 kW$_e$
- Power supports active refrigeration of LH$_2$
- Innovative “saddle” truss design allows easy jettisoning of “in-line” LH$_2$ tank & contingency consumables
- Vehicle rotation ($\omega$ 4-6 rpm) can provide Mars gravity to crew outbound and near Earth gravity inbound (available option)
- Propulsive Mars capture and departure on piloted mission
- Fewest mission elements, simple space ops & reduced crew risk
- Bimodal NTR vehicles easily adapted to Moon & NEA missions

**Engine Characteristics**
- Three 15 klb$_f$ engines, $T/W_{eng}$ ~3.1
- Each bimodal NTR produces 25 kW$_e$
- Utilizes proven Brayton technology
- Variable thrust & $I_{sp}$ optional with “LOX-afterburner” nozzle (LANTR)

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**Piloted Transfer Vehicle**

**TransHab**

**Bimodal NTR Stage**
Mars DRM 4.0: “Bimodal” NTR Crew Transfer Vehicle (CTV) with Inflatable “TransHab” Module & Artificial Gravity Capability

“Bimodal” NTR Core Stage w/Refrigeration
(Sized for Delivery by “Shuttle-Derived” HLV)

3 x 15 klbf BNTRs
(F/W_{eng} = 3.1)

48.6t Capacity LH\textsubscript{2} Tank
Refrigeration System

50 kWe CBC w/Radiator

RCS

7.4m I.D.

3.5m

19m

28m

64.15m

IMLEO: ~166.4 t

“In-Line” Propellant Tank
(Tank Jettisoned)

43t Capacity LH\textsubscript{2} Tank

Strongback Truss

Jettisonable Consumables (~6.9t)

9.44m

Shuttle Launched
“TransHab” Module
(Payload ~21.1t)

ECRV (~5.1t)

Mars DRM 4.0: “Bimodal” NTR Crew Transfer Vehicle (CTV) with Inflatable “TransHab” Module & Artificial Gravity Capability

NASA/CP—2004-212963/VOL1
“Bimodal” Crew Transfer Vehicle
Earth Orbit Assembly Sequence

1: Rendezvous

Two “80 t” SDHLV payloads rendezvous and dock prior to Shuttle rendezvous.

ECRV retrieved by SRMS.

2: Assembly

ECRV checked out for crew use.

SRMS used to attach packaged TransHab to CTV.

3: Final CTV Configuration

ECRV transfers crew from Shuttle to CTV. Crew inflates TransHab, deploys flooring and partitions, and checks out CTV systems.
“Bimodal” NTR Crew Transfer Vehicle (CTV) in Artificial Gravity Mode
2014 “Bimodal” NTR Piloted Flight Profile
(210 Day Transit Out, 190 Day Return)

Mars @ Departure
Jan. 3, 2016

Return Inbound Trajectory

Outbound Trajectory

Earth @ Departure
Jan. 21, 2014

Earth @ Arrival
July 11, 2016
(190 days IB)

Mars @ Arrival
Aug. 19, 2014
(210 days OB)

Mars Stay Time: 502 days

Mars Perihelion:
January 22, 2013
December 10, 2014

Earth/Mars Synodic Period:
The proper alignment with Mars occurs every 2.13 yrs allowing the “opening” of the TMI window.
Human Mars Mission Architecture Mass Comparison
(Shown at 80 t steps)

- TMI Stage
- MOC/TEI Stage
- Chemical Descent Stages
- Aerobrake/Descent Shells
- Payload (Surface, Habs, etc.)

ISS @ Assembly Complete (470 t)

IMLEO (t)

- Bimodal NTR
- SEP/Chem
- Chem TMI

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<th>IMLEO (t)</th>
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"LOX-Augmented" Nuclear Thermal Rocket (LANTR)  
"Afterburner" Nozzle Concept Demonstration

**LANTR Concept and Benefits:**
- "Afterburner" nozzle increases thrust by injecting & combusting GO₂ downstream of the NTR throat
- Enables NTR with variable thrust and Isp capability by varying the nozzle O/H mixture ratio (MR)
- Operation at modest MRs (<1.0) helps increase bulk propellant density for packaging in smaller volume launch vehicles
- LANTR's bipropellant operation enables smaller, faster Moon / Mars vehicles when using extraterrestrial sources of H₂ and O₂

**LANTR Test Program Objectives:** (Aerojet & GRC)
- Measure thrust augmentation from oxygen injection and supersonic combustion using small, fuel-rich H/O engine with two different area ratio nozzles (@ 25:1 and 50:1) as "non-nuclear" NTR simulator.
- Use results to calibrate reactive CFD assessment of bimodal LANTR engine

**Status:** LANTR afterburner nozzle demonstrated
- Oxygen injection into hot supersonic flow
- Supersonic combustion in the nozzle
- Elevated nozzle pressures measured
- Benign nozzle wall environment observed
- Increase O₂ consumption rate with nozzle length
- Thrust augmentation >50% measured

Baseline H/O Thrust: 2100 lbf at 1000 psia and MR = 1.5. With GO₂ injection into nozzle, measured thrust due to supersonic combustion is 3200 lbf (~52% thrust augmentation achieved at 50:1 and MR~3.0)
Fully Reusable NTR-Powered Transfer Vehicle
“The Key to Affordable Lunar Transportation”

Ref: Borowski, NASA/TM 106739
Robotic Science “Hybrid” BNTEP Vehicle

Elevation View

- Xenon Thruster Clusters
- Docking Interface
- Saddle Truss
- Electrical and Coolant Conduit Lines
- LH₂ Refrigeration System & Radiator
- "Jetisonable" LH₂ Tank
- Saddle Truss-Mounted Radiator & Foldout Panels
- Science Payload
- Conical Radiator (~26 m²)
- Top View

- "Core" Stage
- Toroidal LOX Tank
- LH₂ Tank
- shield
- LH/W

2 - 60 kW BRUs @ 50% power (enclosed)
5 kib, BNTR

NASA/CP—2004-212963/VOL1
Significant Technology Development is Underway To Support Design Definition for Future "Bimodal" NTR Human Exploration Missions