ABSTRACT

Method and system for analyzing, separately or in combination, kinetic energy and potential energy and/or their time derivatives, measured or estimated or computed, for an aircraft in approach phase or in takeoff phase, to determine if the aircraft is or will be put in an anomalous configuration in order to join a stable approach path or takeoff path. A reference value of kinetic energy and/or potential energy (or time derivatives thereof) is provided, and a comparison index for the estimated energy and reference energy is computed and compared with a normal range of index values for a corresponding aircraft maneuver. If the computed energy index lies outside the normal index range, this phase of the aircraft is identified as anomalous, non-normal or potentially unstable.

19 Claims, 3 Drawing Sheets
Fig. 1A

Fig. 1B
Provide estimated values
\[ E(t_n) = C_1 KE(t_n) + C_2 PE(t_n) \]
for kinetic energy and/or potential energy components
at a sequence of times \( \{t_n\}_n \), where
\( C_1 \) and \( C_2 \) are selected real values, not both 0

Provide reference values \( E(t'_n; \text{ref}) \)
for energy components
at a sequence of times \( \{t'_n\}_n \)

Compute an index comparison
\[ C_1 \{E(t_n), E(t'_n; \text{ref})\} \]
for at least one time pair \( (t_n, t'_n) \)

When the comparison index value \( C_1 \) lies outside a selected range for this index, interpret this condition as indicating that the estimated kinetic energy component and/or potential energy component is anomalous.

Fig. 2
Provide estimated values $\frac{dE(t_n)}{dt} = d_3 \frac{d}{dt} KE(t_n) + d_4 \frac{d}{dt} PE(t_n)$ for time derivatives of kinetic energy components and/or of potential energy components at a sequence of times $\{t_n\}_n$; where $d_3$ and $d_4$ are selected real values, not both 0.

Provide reference values $\frac{dE(t''_n, \text{ref})}{dt}$ for energy at a sequence of times $\{t''_n\}_n$.

Compute an index of comparison value $C2 \{ \frac{dE(t_n)}{dt}, \frac{dE(t''_n, \text{ref})}{dt} \}$ for at least one time pair $\{t_n, t''_n\}$.

When the comparison index value $C2$ lies outside a selected range for this index, interpret this condition as indicating that the estimated kinetic energy component time derivative and/or potential energy component time derivative is anomalous.

**Fig. 3**
DETAILED DESCRIPTION OF THE INVENTION

SUMMARY OF THE INVENTION

The airline industry has become concerned with the problem of unstable aircraft approaches, because approach and landing accidents often begin as unstable approaches. An “unstable approach” is often defined as an approach where below a threshold altitude (1,000 feet for IFR and 500 feet for VFR), the aircraft is not established on a proper glide path and with a proper air speed, with a stable descent rate and engine power setting, and with a proper landing configuration (landing gear and flaps extended). Airlines have developed approach procedures that call for abandonment of an approach that is determined to be unstable.

Development and testing of methods for detecting atypical flights by N.A.S.A. has revealed that high energy during an arrival phase (below 10,000 feet but before beginning an approach) is the most common reason for a flight to be identified as atypical or out of a statistically normal range. An atypical high energy arrival phase often corresponds to aircraft kinetic energy and/or potential energy that requires dissipation of 10–30 percent more energy than is required for a normal arrival phase. A normal arrival phase may correspond to an 3° glide path slope and decelerating to an airspeed of about 250 knots during descent through 10,000 feet altitude to a standard reference speed around 2,500 feet altitude, when beginning an approach.

More than half of the high energy arrivals identified by atypicality analysis were brought under control within stabilized approach criteria; some of the remaining of the high energy arrivals were abandoned. In contrast, where these findings were used to define and search for a high-energy arrival exceedance, about three times as many exceedances were detected; and the resulting unstable approaches were found to occur more frequently than the recoveries.

It may be possible to identify a first class of high energy arrivals where recovery and subsequent stabilization is possible and relatively easy, and a second class of high energy arrivals in which recovery and subsequent stabilization is likely to be difficult or impossible. However, the present procedures for determining presence of a normal approach include an electronic glide slope that extends linearly from the end of a target runway to the aircraft, whereas a typical (normal) aircraft approach path is curved and follows the electronic glide slope only from about 1,800 feet above the field to the end of the runway.

A 3-to-1 glide path slope, corresponding to a decrease of 1,000 feet in altitude for every 3 nautical miles horizontal travel, is often desirable during an arrival phase. The airspeed is 250 knots or less by regulation below 10,000 feet, and the aircraft decelerates to a reference speed before joining the approach path. These parameters are directly available but are unlikely to prove to be the only relevant parameters in determining whether a flight arrival phase is normal or other than normal.

What is needed is a system that (1) provides estimates of kinetic energy and potential energy components, and rates of change thereof, of an ascending or descending aircraft at any altitude, (2) provides reference values of kinetic energy and potential energy components, and rates of change thereof, of the aircraft; (3) provides one or more comparison indices for the estimated and reference values; and (4) advises an aircraft operator if the measured comparison indices are too far from the corresponding index values for a normal flight approach.

This invention relates to monitoring and analysis of kinetic energy and potential energy of an ascending or descending aircraft.

A potential energy component, and these components will be discussed in the context of descending aircraft.
component; or may be based on a weighted average over time of the estimated and reference values of the energy components. The comparison index may be chosen from a group of such indices that includes: (1) a first ratio \( \frac{E(t',;\text{ref})}{E(t,;\text{ref})} \); (2) a second ratio \( \frac{E(t',;\text{ref})}{E(t,;\text{ref})} \); (3) a difference \( E(t,;\text{ref}) - E(t',;\text{ref}) \); (4) an absolute difference \( |E(t,;\text{ref}) - E(t',;\text{ref})| \); (5) a normalized difference \( \frac{|E(t,;\text{ref}) - E(t',;\text{ref})|}{\max{E(t,;\text{ref})}, \min{E(t,;\text{ref})}} \); or (6) a weighted average of the differences \( \sum w_i \left( E(t,i;\text{ref}) - E(t',i;\text{ref}) \right) \). An analogous procedure may be applied to time rates of change of the kinetic and potential energy components.

One energy index is a ratio of actual aircraft energy divided by ideal aircraft total energy during an arrival phase. If this ratio lies near a boundary but outside a "normal" range for the energy index (e.g., between about 0.90 and 1.10), this arrival phase may be considered non-normal, and appropriate remedial actions may be taken to recover to a stabilized approach. If this ratio is below a threshold, (e.g., below 0.85) or above a second threshold (e.g., above 1.20) for an arrival phase, the aircraft is unlikely to be able to recover, and the aircraft is better advised to abandon the approach, to execute a go-around, and to re-enter a new arrival phase.

BRIEF DESCRIPTION OF THE DRAWINGS

FIGS. 1A and 1B illustrate environments in which the invention can be practiced.

DESCRIPTION OF BEST MODES OF THE INVENTION

FIGS. 1A and 1B illustrate environments for an ascending aircraft (1A) and for a descending aircraft (1B) where the invention can be practiced. In FIG. 1A, an aircraft 11A is ascending, either after takeoff or in moving from a first flight altitude to a second flight altitude. The aircraft has an associated kinetic energy component \( KE(t,;\text{ref}) \), measured or estimated or otherwise provided, at each of a first sequence \{t\}, of two or more time values, and has an associated potential energy component \( PE(t,;\text{ref}) \), measured or estimated or otherwise provided, at the first sequence \{t\}, of time values. The aircraft kinetic energy and potential energy components are, respectively,

\[
KE(t) = \frac{1}{2}m(t)v(t)^2 + \frac{1}{2}I(t)v(t)^2 - mg(h(t) + h_{\text{bg}}),
\]

(1)

\[
PE(t) = \frac{1}{2}I(t)v(t)^2 - gh(t),
\]

(2)

where \( m(t) \) is the instantaneous mass (taking account of fuel consumption), \( I(t) \) is an instantaneous moment of inertia tensor for the aircraft, \( \alpha(t) \) is an aircraft rotation vector, computed with reference to a center of gravity or other selected location determined with reference to the aircraft. (optional), \( v(t)=dx/dt \) is the instantaneous aircraft velocity and \( h(t) \) is the instantaneous height of aircraft cg above local ground.

FIG. 2 is a flow chart of a procedure for practicing an embodiment of the invention. In step 21, an aircraft system measures or estimates or otherwise provides a value (referred to as an "estimated value" for convenience herein) \( E(t,i;\text{ref}) = d1 KE(t,i) + d2 PE(t,i) \) of an energy component of an aircraft during an ascent phase or descent phase of a flight, at each of a first sequence of times \( t = t_1, \ldots, t_N \), where \( d1 \) and \( d2 \) are selected real values, not both 0. In step 23, the system provides or computes a reference value \( E(t',;\text{ref}) \) of the energy component at a time, \( t = t_0 \), determined with reference to the time \( t_0 \), and \( t_0 \), or each time value \( t_i \) may be displaced by a calculable or measurable amount from the corresponding time value \( t_i \).

In step 25, the system computes an index of comparison value \( C1(E(t,i), E(t',;\text{ref})) \) of the estimated and reference energy components for at least one time value pair \( (t_i, t_0) \). When the comparison index value \( C1 \) lies outside a selected range, this index is interpreted as indicating that the estimated energy component is anomalous or non-normal or may lead to an unstable aircraft maneuver.

A variety of comparison indices \( C1 \) can be used here. Some examples are: (1) a first ratio \( \frac{E(t,i)}{E(t',;\text{ref})} \); (2) a second ratio \( \frac{E(t',;\text{ref})}{E(t,i)} \); (3) a difference \( E(t,i) - E(t',;\text{ref}) \); (4) an absolute difference \( |E(t,i) - E(t',;\text{ref})| \); (5) a normalized difference \( \frac{|E(t,i) - E(t',;\text{ref})|}{\max{E(t,i), E(t',;\text{ref})}} \); or (6) a weighted average of the differences \( \sum w_i \left( E(t,i) - E(t',;\text{ref}) \right) \).

The analysis may be extended to consider time rates of change of energy, \( dE(t)/dt \) and/or \( dPE(t)/dt \), of the energy components at a sequence of one or more time values \( t = t_1, \ldots, t_N \), plus corresponding reference time rates of change, \( dE(t',;\text{ref})/dt \), at a sequence of times \( t' \), determined with reference to the time sequence \( t \). Another comparison index, \( C2(\text{dE}(t,/dt), \text{dE}(t',;\text{ref})/dt) \), which may be the same or different from the comparison index \( C1 \), is computed and compared with a second selected range to determine if the aircraft flight is anomalous or non-normal or is within a normal range. Again, the comparison index \( C2 \) may use point values or a weighted average of the values \( dE(t)/dt \) and/or \( dPE(t)/dt \).

FIG. 3 is a flow chart of a procedure for practicing an embodiment using time derivatives of the energy component \( E(t,i) \). In step 31, an aircraft system measures or estimates or otherwise provides an "estimated value", \( \text{dE}(t);\text{est}) \), calculated with reference to a center of gravity or other selected location determined with reference to the aircraft. (optional), \( v(t) \) is the instantaneous aircraft velocity and \( h(t) \) is the instantaneous height of aircraft cg above local ground.

FIG. 3 is a flow chart of a procedure for practicing an embodiment of the invention. In step 21, an aircraft system measures or estimates or otherwise provides a value (referred to as an "estimated value" for convenience herein) \( E(t,i;\text{est}) = d3 KE(t,i;\text{est}) + d4 PE(t,i;\text{est}) \) of an energy component of an aircraft during an ascent phase or descent phase of a flight, at each of a first sequence of times \( n=1, \ldots, N1 \), where \( d3 \) and \( d4 \) are selected real values, not both 0. In step 33, the system provides or computes a reference value \( \text{dE}(t',;\text{ref}) \) of the energy component at a time, \( t = t_0 \), determined with reference to the time \( t_0 \), and \( t_0 \), or each time value \( t_i \) may be displaced by a calculable or measurable amount from the corresponding time value \( t_i \).
interprets this condition reference energy component time derivatives for at least one or measurable amount from the corresponding time value \( t_{r} \). When the comparison index value \( C_{2} \) lies outside a selected range for this index, the system interprets this condition as indicating that the estimated energy component time derivative is anomalous or non-normal or may lead to an unstable aircraft maneuver (step 37).

The analysis may be further extended to consider a third comparison index, \( C_{3}(E(t_{1}), E(t_{2};\text{ref}), (d/dt)E(t_{1}), (d/dt)E(t_{2};\text{ref}) \), that depends upon some or all of the estimated values and time rates of change of the estimated values of the energy components. Again, the comparison index \( C_{3} \) may use point values or a weighted average of the values \( E(t_{1}) \) and/or \( E(t_{2};\text{ref}) \) and/or \( (d/dt)E(t_{1}) \) and/or \( (d/dt)E(t_{2};\text{ref}) \).

A formulation of, and use of, the equations of motion of an aircraft, including the effects of gravity, variable wind speeds, drag and lift forces on various control surfaces, variation of aircraft mass due to fuel consumption, and variable thrust, is set forth in an Appendix. A thrust vector is determined, as a function of the location coordinates, that will move the aircraft from an initial velocity \( v_{i}(x_{0}, y_{0}, z_{0}) \) to a desired final velocity \( v_{f}(x_{f}, y_{f}, z_{f}) \) as part of a takeoff phase or as part of an approach phase for a flight. The aircraft kinetic energy is a sum

\[
KE = \frac{1}{2}m(v_{x})^{2} + \frac{1}{2}m(v_{y})^{2} + \frac{1}{2}m(v_{z})^{2},
\]

where \( I(t) \) is an instantaneous moment of inertia tensor for the aircraft and \( c(t) \) is an aircraft rotation vector, computed with reference to a center of gravity or other selected location determined with reference to the aircraft. The rotational component of kinetic energy may be negligible or may be ignored for other reasons. The aircraft potential energy may be taken to be \( mgh \), as in Eq. (2), where \( h \) is the aircraft altitude above local ground level.

Appendix: A Formulation of Aircraft Equations of Motion

Let \( v(x,y,z)=(v_{x}, v_{y}, v_{z}) \) by the vector velocity for an aircraft configuration \( C \), let \( m=m(t) \) be the mass of the aircraft (with or without fuel consumption accounted for), and let \( F(x,y,z)=\left(F_{x}, F_{y}, F_{z}\right) \) for the total force vector acting upon the aircraft, including a thrust vector. One vector equation of motion for the aircraft is

\[
dm/dt = F
\]

The time variation of the aircraft mass is likely close to linear \( (m(t)=m_{0}-m_{1}t) \). When the thrust vector \( F(\text{thrust}) \) is known as a function of the location coordinates \((x,y,z)\), Equation (A-1) can be re-expressed as

\[
dm/dt = m\left[\delta\dot{v}_{x}/\delta x, \delta\dot{v}_{y}/\delta y, \delta\dot{v}_{z}/\delta z\right] + \dot{\delta}(mv)\delta y + \delta(\dot{mv})/\delta y
dm/dt = \left[\delta(mv)/\delta x, \delta(mv)/\delta y, \delta(mv)/\delta z\right] + \delta(lv)(mv) = F
\]

The force vector may be approximated as

\[
F(x,y,z)=\text{mg} + F(\text{drag/wind}) + F(\text{lift}) + F(\text{thrust})
\]
the aircraft from an initial condition \(v(x_0, y_0, z_0)\) to a desired final condition \(v'(x_0, y_0, z_0)\) as part of an ascent phase or descent phase of the flight. As a first approach, parameters and associated forces (drag, lift, thrust, etc.) may be measured at a sequence of times for a moving aircraft, and Eq. (A-8) may be solved for a sequence of corresponding location coordinates \((x, y, z)\) to determine or estimate an aircraft velocity vector \(v\) and altitude \(h\) to determine aircraft kinetic energy \(KE\) and/or potential energy \(PE\) for this particular flight movement. Here, the kinetic energy and/or potential energy of the aircraft are determined or estimated off-line, after this portion of the flight is completed.

As a second approach, a proposed aircraft maneuver to move from an initial velocity condition \(v(x_0, y_0, z_0)\) and initial altitude to a final velocity condition \(v(x_0, y_0, z_0)\) and final altitude can be posited and a thrust field \(F(\text{thrust})\) can be determined that will accomplish this can be determined, through solution of Eq. (A-8), before or during execution of the aircraft maneuver (on-line).

What is claimed is:

1. A method of monitoring energy components of an aircraft in flight, the method comprising:
   - providing an estimate or measurement of a value (referred to as an "estimated value") of an energy component, \(E(t_0)\), of a combination of a kinetic energy component \(KE(t_0)\) and a potential energy component \(PE(t_0)\) of an aircraft during an ascent phase of a flight, at each of a first sequence of times \(n=1, \ldots, N\); and
   - providing a reference value \(E(t_0, \text{ref})\) of said reference energy component, over said respective sequences of times \(n=1, \ldots, N\); and
   - providing an index of comparison \(C1\{E(t_0), E(t_0, \text{ref})\}\) of the estimated and reference energy components for at least one time number \(n\); and
   - when a value of the comparison index \(C1\) lies outside a selected range, interpreting this condition as indicating that the estimated energy component is anomalous.

2. The method of claim 1, further comprising:
   - providing reference values, \(KE(t_0, \text{ref})\) and \(PE(t_0, \text{ref})\), of kinetic energy and potential energy components for said estimated values; and
   - choosing said comparison index from the group of indices consisting of: (1) a first ratio \(E(t_0)/E(t_0, \text{ref})\); (2) a second ratio \(E(t_0, \text{ref})/E(t_0)\); (3) a difference \(E(t_0) - E(t_0, \text{ref})\); (4) an absolute difference \(|E(t_0) - E(t_0, \text{ref})|\); (5) a normalized difference \(|E(t_0) - E(t_0, \text{ref})/(\alpha E(t_0) + (1-\alpha)E(t_0, \text{ref}))|\), where \(\alpha\) is a selected value in a range \(0\leq\alpha\leq1\); and
   - (6) weighted averages, \(\alpha KE(t_0) + (1-\alpha)PE(t_0, \text{ref})\) and \(\alpha PE(t_0, \text{ref}) + (1-\alpha)PE(t_0)\), of differences of kinetic energy terms and of potential energy terms, where \(\alpha\) is a selected positive number.

3. The method of claim 1, further comprising choosing said comparison index to include at least one of said values \(E(t_0)\) for said estimated energy component and at least one of said values \(E(t_0, \text{ref})\) for said reference energy component.

4. The method of claim 1, further comprising choosing said comparison index to include at least one weighted average of said values \(E(t_0)\) for said estimated energy component and at least one weighted average of said values \(E(t_0, \text{ref})\) for said reference energy component, over said respective sequences of times \(t_n\) and \(t_n^\text{ref}\).

5. The method of claim 1, further comprising:
   - providing an estimated value \((d/dt)E(t_n)\) for said estimated energy component, where \(d3\) and \(d4\) are selected real values, not both 0; and
   - providing a reference value \((d/dt)E(t_n, \text{ref})\) of said estimated energy component and \((d/dt)E(t_n, \text{ref})\) of said reference energy component, where \(t_n^\text{ref}\) is a time determined with reference to said time \(t_n\).

6. The method of claim 5, further comprising:
   - providing an index of comparison \(C2\{(d/dt)E(t_n); (d/dt)E(t_n, \text{ref})\}\) of said estimated values and time rates of change of said estimated energy component and said reference energy component for at least one time number \(n\).

7. The method of claim 6, further comprising choosing said comparison index to include at least one of said values \((d/dt)E(t_n)\) for said estimated energy component and at least one of said values \((d/dt)E(t_n, \text{ref})\) for said reference energy component.

8. The method of claim 6, further comprising choosing said comparison index to include at least one weighted average of said values \((d/dt)E(t_n)\) for said estimated energy component and at least one weighted average of said values \((d/dt)E(t_n, \text{ref})\) for said reference energy component, over said respective sequences of times \(t_n\) and \(t_n^\text{ref}\).

9. The method of claim 6, further comprising:
   - when a value of said comparison index \(C2\) lies outside a selected range, interpreting this condition as indicating that said time rate of change of said estimated energy component is anomalous.

10. The method of claim 5, further comprising:
    - providing an index of comparison \(C3\{E(t_n); E(t_n, \text{ref})\}\) of said estimated values and time rates of change of said estimated energy component and said reference energy component for at least one time number \(n\); and
    - when a value of the comparison index \(C3\) lies outside a selected range, interpreting this condition as indicating that at least one of said value and said time rate of change of said estimated kinetic energy component is anomalous.

11. The method of claim 1, further comprising choosing said real numbers \(d1\) and \(d2\) to be one of the following pairs of real numbers: \((d1,d2)=(1,0)\), \((d1,d2)=(1,1)\), \((d1,d2)=(0,1)\), and \((d1,d2)=(d,1-d)\) with \(0<d<1\).

12. The method of claim 1, further comprising determining said energy component by a process comprising:
    - measuring aircraft thrust vector components and at least one of extraneous force vector components, comprising wind vector components, drag force vector components, lift force vector components and gravity force components, at said sequence of times \(t_n\); and
    - determining or estimating at least one velocity vector \(v\) of said aircraft and altitude of said aircraft at location coordinates \((x_n, y_n, z_n)\) corresponding to said time \(t_n\), from a solution of an equation \((v\cdot\Delta v)=a\) sum of extraneous vector forces \(F(\text{extraneous})\) on said aircraft and thrust vector force \(F(\text{thrust})\) provided by one or more engines of said aircraft; and
    - determining or estimating at least one of an aircraft velocity vector \(v(x_n, y_n, z_n)\) and an aircraft altitude for at least one of said times \(t_n\); and
    - computing said estimated or measured value of said energy component \(E(t_n)\) for at least one of said times \(t_n\), using the determined or estimated aircraft velocity vector and aircraft altitude.

13. The method of claim 1, further comprising determining said energy component by a process comprising:
    - providing at least one of extraneous force vector components \(F(\text{extraneous})\), comprising wind vector compo-
nents, drag force vector components, lift force vector components and gravity force components, at said sequence of times \( t_n \);

- determining or estimating at least one of thrust vector components \( F(\text{thrust}) \) required to transport an aircraft from a selected initial velocity condition \( \{x_n,y_n,z_n\} \) to a selected final velocity condition \( \{x_f,y_f,z_f\} \) under influence of the at least one extraneous force components; determining or estimating at least one of velocity vector \( v \) of said aircraft and altitude of said aircraft at coordinates \( \{x_n,y_n,z_n\} \) corresponding to said time \( t_n \), from a solution of an equation \( (v.A)(mv)=a \sum \) of vector forces \( F(\text{extraneous}) \) acting on said aircraft; the thrust vector force \( F(\text{thrust}) \), at said sequence of times \( \{t_n\} \);

- providing an estimate of thrust vector components and at least one weighted average of said values \( \{\text{d}E(t_n),\{\text{d}KE(t_n),\{\text{d}PE(t_n)\} \) for said reference energy component, over said respective sequences of times \( \{t_n\} \) and \( \{t'_n\} \);

16. The method of claim 14, further comprising choosing said comparison index to include at least one weighted average of said values \( \{\text{d}E(t_n)\} \) for said estimated energy component and at least one weighted average of said values \( \{\text{d}E(t'_n)\} \) for said reference energy component, over said respective sequences of times \( \{t_n\} \) and \( \{t'_n\} \);

17. The method of claim 14, further comprising choosing said real numbers \( d_1 \) and \( d_2 \) to be one of the following pairs of real numbers: \((d_3,d_4)=(0,1)\), \((d_3,d_4)=(1,1)\), \((d_3,d_4)=(0,1)\), and \((d_3,d_4)=(d_1-d) \) with \( 0<d_1<d_1 \);

18. The method of claim 14, further comprising determining said energy component by a process comprising:

- measuring aircraft thrust vector components and at least one of extraneous force vector components, comprising wind vector components, drag force vector components, lift force vector components and gravity force components, at said sequence of times \( \{t_n\} \);

- determining or estimating at least one of velocity vector \( v \) of said aircraft and altitude of said aircraft at location coordinates \( \{x_n,y_n,z_n\} \) corresponding to said time \( t_n \), from a solution of an equation \( (v.A)(mv)=a \sum \) of vector forces \( F(\text{extraneous}) \) acting on said aircraft; the thrust vector force \( F(\text{thrust}) \) provided by one or more engines of said aircraft;

- determining or estimating at least one of an aircraft velocity vector \( v(x_n,y_n,z_n) \) and an aircraft altitude for at least one of said times \( t_n \); and

- computing said estimated or measured value of said energy component \( E(t_n) \) for at least one of said times \( t_n \), using the determined or estimated aircraft velocity vector and aircraft altitude.

19. The method of claim 14, further comprising determining said energy component by a process comprising:

- providing at least one of extraneous force vector components \( F(\text{extraneous}) \), comprising wind vector components, drag force vector components, lift force vector components and gravity force components, at said sequence of times \( \{t_n\} \);

- providing an estimate of thrust vector components \( F(\text{thrust}) \) required to transport an aircraft from a selected initial velocity condition \( \{x_n,y_n,z_n\} \) to a selected final velocity condition \( \{x_f,y_f,z_f\} \) under influence of the at least one extraneous force components;

- determining or estimating at least one of velocity vector \( v \) of said aircraft and altitude of said aircraft at location coordinates \( \{x_n,y_n,z_n\} \) corresponding to said time \( t_n \), from a solution of an equation \( (v.A)(mv)=a \sum \) of vector forces \( F(\text{extraneous}) \) acting on said aircraft; the thrust vector force \( F(\text{thrust}) \), at said sequence of times \( \{t_n\} \);

- determining or estimating at least one of an aircraft velocity vector \( v(x_n,y_n,z_n) \) and an aircraft altitude for at least one of said times \( t_n \); and

- computing said estimated or measured value of said energy component \( \{\text{d}E(t_n)\} \) for at least one of said times \( t_n \), using the determined or estimated aircraft velocity vector and aircraft altitude.

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