The invention is a method and system for monitoring and directly displaying the actual thrust produced by a jet aircraft engine under determined operating conditions and the available thrust and predicted (commanded) thrust of a functional model of an ideal engine under the same determined operating conditions. A first set of actual value output signals representative of a plurality of actual performance parameters of the engine under the determined operating conditions is generated and compared with a second set of predicted value output signals representative of the predicted value of corresponding performance parameters of a functional model of the engine under the determined operating conditions to produce a third set of difference value output signals within a range of normal, caution, or warning limit values. A thrust indicator displays when any one of the actual value output signals is in the warning range while shaping function means shape each of the respective difference output signals as each approaches the limit of the respective normal, caution, and warning range limits.

27 Claims, 3 Drawing Sheets
FIG. 3

shaped response

linear response \( y = x \)

input \( x \)

output \( y \)

output bar height

(Upward direction only shown)

\( h \)
METHOD AND SYSTEM FOR MONITORING AND DISPLAYING ENGINE PERFORMANCE PARAMETERS

ORIGIN OF INVENTION

The invention described herein was made by employees of the U.S. Government and may be manufactured and used by or for the Government for governmental purposes without the payment of any royalties thereon or therefor.

BACKGROUND OF THE INVENTION

1. Field of the Invention

The invention is a method and system for monitoring and displaying engine performance parameters and more particularly to a method and system for simultaneously monitoring and visually displaying, a plurality of the performance parameters of an aircraft engine during operation, including the predicted value of maximum available thrust or power, the predicated amount of thrust commanded, and the thrust then actually being produced within the critical limits of the predicted nominal and actual values of the plurality of monitored engine performance parameters.

2. Description of the Prior Art

In general conventional single and multiple engine aircraft control systems include individual instruments that provide operational data (engine performance parameter measurements) to the pilot based on the outputs of a plurality of individual sensors.

Some electronically generated, microprocessor driven displays in multiple engine jet aircraft, such as the Boeing 757/767 manufactured by Boeing Commercial Airplane Company, P.O. Box 3707, Seattle, Wash. 98124, use a similar approach, except the outputs from two or more sensors may be presented on a single electronic display.

Thus, conventional instrumentation can result in a cluttered control panel that increases the pilots workload in reading and interpreting the outputs of the various instruments. This in turn adds to the difficulty of readily and accurately identifying changes in critical and changing engine performance parameters that can lead to degrading off-nominal or off-limit operating conditions.

For instance, currently, before making a thrust or power adjustment during take off, pilots of multiple engine jet aircraft use charts to calculate the amount of thrust available from each engine and then using this reading to set either the engine pressure ratio (EPR) or low pressure compression rotational speed for each engine. In addition, the pilot must also cross-check the several engine performance parameters for each engine such as exhaust gas temperature (EGT), fuel flow (FF), oil pressure, temperature, and quantity to assure all are within the operational limits of each engine as provided by the manufacturer or based on the pilots experience and judgement before changing the thrust of the engines. This is an indirect, time consuming thrust control process.

SUMMARY OF THE INVENTION

An object of the invention is to provide a method and system for monitoring and displaying aircraft engine performance parameters permitting a pilot to make direct operational changes in thrust or power based solely on engine performance parameter that are simultaneously monitored and displayed.

A further object of the invention is to provide a method and system for monitoring and displaying aircraft engine performance parameters such as available thrust, the predicted amount of thrust commanded, and the actual amount of thrust being produced within the critical limits of predicted normal and actual engine performance parameters.

A further object of the invention is to provide a method and system for directly displaying the difference between predicted and actual performance parameters of a jet aircraft engine that are within determined limits, of a range of values.

Another object of the invention is to provide a method and system for simultaneously displaying the differences between the predicted and actual values of a plurality of engine performance parameters for one or more jet aircraft engines in a single display, the actual values being displayed in a first format and the difference values being displayed in a second format indicative of determined limits of said difference values.

Still another object of the invention is to provide a monitoring and display system and method in which difference signals are shaped as their values approach the limit of one or more ranges of limit values.

BRIEF DESCRIPTION OF THE DRAWINGS

The above and numerous other objects of the invention will become apparent from the following description when read in light of the appended drawings wherein:

FIG. 1 schematically illustrates a preferred embodiment of a monitoring and display system in accordance with the invention:

FIG. 2 illustrates the thrust indicator display and monitoring indicator display as shown in FIG. 1 in greater detail; and

FIG. 3 is a diagram illustrating the manner in which noise or jitter is reduced in the monitoring indicator display by using a shaping function.

BRIEF DESCRIPTION OF THE INVENTION

Referring to the drawings, FIG. 1 illustrates a preferred embodiment of the invention for use in single or multiple engine jet aircraft as generally comprising an engine monitoring system section 10 interconnected between an aircraft system section 11 and a display section 12.

Aircraft system section 11 includes a commercially available air data computer 13, such as a model number A.320 ADIRS air data computer manufactured by Honeywell responsive to conventional jet aircraft sensors 14 mounted and arranged to provide digital output signals 14, 16, and 17, representative respectively of the operating conditions of external air temperature, air pressure, and the mach or airspeed of the aircraft as a percentage of the speed of sound at ground level.

Aircraft system section 11 also provides parallel digital output signals 18 and 19 representative of the aircrafts throttle position as controlled by a pilot, the output signal 18 being directly connected to control for example a respective jet engine 20 such as a model JT8D-7 jet engine manufactured by Pratt and Whitney, (address). Conventional aircraft system sensors 15 responsive to the operation of the jet engine 20 provide a first group of digital output signals representative of the position of bleed valve 21 and the engine gas pressure
Thus, subsection 31 calculates and provides digital output signals 32-40 representative of the predicted engine performance parameters of a functional model of an ideal engine when operated under the same operating conditions as engine 20 as indicated by the digital output signals 14, 18, 22, 16, 17, and 21.

A simplified functional model for each of the ideal engine performance parameters represented by the digital output signals 32-40 is used to calculate the ideal thrust limits in subsection 42 as represented by output signals 45, 46, and 50, normalize the actual thrust in subsection 44 to generate output signal 47, normalize the predicted thrust in subsection 41 to generate output signal 59, and perform the differencing function in subsection 51.

To produce the required functional models it is assumed that for any given value of an engine performance parameter, $N_1$ for example, a reasonable approximation of the actual $N_1$ characteristics of the engine 20 can be obtained from: a third-order polynomial equation. To obtain the required coefficients, a set of data is obtained from the engine 20 for a wide range of flight conditions (altitude, mach, temperature, and throttle positions), from either the engine manufacturers data or flight test data. Applying a multiple polynomial regression analysis to this data provides the relevant coefficients.

Rejecting unnecessary terms, this regression serves to reduce the complexity of the final functional model. For instance, when this analysis is applied to a model JT8D-7 jet engine made by Pratt and Whitney Aircraft Group, 400 Main Street, Commercial Products Division, East Hartford, Conn. 06108, the functional model for the ideal engine performance parameter is of the following form:

$$N = a_0 + a_1X^2 + a_2Y^2 + a_3Z^2 - a_4X + a_5Y + a_6Z - a_7XYZ$$

Where $X$ is EPR, $Y$ is mach, $Z$ is altitude, and $a_0,...,a_7$ are the coefficients. Additionally, a standard atmospheric model was assumed. This means that altitude has temperature effect included. For a non-standard atmosphere, altitude and temperature would be two separate input variables instead of one. In practice, two $N_1$ models may be used, one for the bleed-valve, normally closed condition signal 21 or a bleed-valve open condition signal 21-0. A partially open bleed-valve condition signal may also be provided.

For example, with bleed-valve 21 closed, assume the predicted thrust of an ideal engine such as a model JT8D-7 jet engine is to be computed using the following inputs:

Output signal 18 = Throttle = 50.35 degrees
Output signal 22 = EPR = 1.855
Output signal 16 = Xalt = 0, where Xalt = altitude/1000
Output signal 17 = Mach = 0
Output signal 21 = BVOpen = Closed

Then, THRUST AVAILABLE = 13690.30 + 0.1033 * Xalt$^3$ - 7345.7260

For example, Mach 0.7, Xalt = -500.00, EPR = 1.855:

Xalt$^2$ = Mach - 0.1734 * Xalt - Mach$^2$ + 299.6407 * Xalt

Then, THRUST AVAILABLE = 13690.30 + 0.1033 * Xalt$^3$ - 7345.7260

Xalt$^2$ - Mach = -0.1734 * Xalt - Mach$^2$ + 299.6407 * Xalt

Mach = 0.7 - 360.9475 * Xalt + 2745.6020 * Mach - 57.9160 * Mach$^2$

Mach$^2$ = 5346.7260 - Mach = 13960
THRUST  =  665.083  -  0.665066  -  Mach$^2$ +
0.00164722  -  Xalt$^2$  -  20.1489  -  Thrust$^2$  +  1821.64  -  Mach$^2$  -
0.287707  -  Xalt$^2$  -  31.4498  -  Thrust$^2$  -  Mach  +  0.371256,
Xalt  +  0.082716.  Xalt$^2$.  Thrust  +  0.188178.  Xalt$^2$.  Mach  -
43.7439.  Thrust  -  6573.05.  Mach  +  30.9052.  Mach  +  760.728.
Thrust.  Mach  -  12.3659.  Thrust.  Xalt  +  127.065.  Mach  -
Xalt  =  7.97973.  Thrust.  Mach  -  Xalt  =  13680

When normalized (13680/13960) it is 98%.

FUEL  F  =  -5585.67700  +  186.4989.  Y.H  EPR$^3$  +  307.2636.
1.8745.  Xalt$^2$.  EPR  +  6165.1390.  EPR  -  62.2521.  Mach  +
225.2723.  Xalt  -  271.1962.  EPR  -  Xalt  =
12.6817.  Mach  -  Xalt  =  6976

0.10144  -  EPR$^2$.  Xalt  +  0.81913.  Mach$^2$.  EPR  -  0.92738E-03.
EPR.  Xalt  +  0.05170.  EPR.  Mach  -  Xalt  =  89.78

N$_2$  =  35.89122  -  1.89779.  EPR$^3$  -  0.70771E-05.  Y.H  Xalt$^2$  +
EPR  +  36.39701.  EPR  +  3.84387.  Mach  -  0.37757.  EPR.
Xalt  -  0.02129.  EPR  -  Mach  -  Xalt  =  91.3

EGT  =  37.1654  +  5.1897.  EPR$^3$  -  0.4468E-03.  Xalt$^3$  +
60.2309.  Mach$^3$  +  45.7712.  Mach$^2$.  EPR  -  0.7257.  Mach$^2$.
Xalt  +  0.0291.  Xalt$^2$.  Mach  +  216.1152.  EPR  -  2.4248.
Xalt  +  4.5583.  EPR  -  Mach  -  1.1548.  EPR  -  Xalt  =  0.6929.
EPR  -  Mach  -  Xalt  =  467.

Because the oil system of the engineer 20 is operated at
mid-range values, the ideal oil system values used are
the mid-range value for the normal operating range for
each value. That is,  
OILP=45.0
OILT=80.0
OILQ=1.15

If desired, the oil system may be modelled in the same
manner as the rest of the engine system.

The differencing function subsection 51 is pro-
grammed to produce any one of a set of digital output
signals 62-68 corresponding to the difference be-
tween the like set of actual digital output signals 23-29
and the set of ideal output signals 34-40. The scaling of these
output signals 62-68 is such that

normal:  ±(0 to 10%)
caution:  ±(10 to 15%), and
warning:  ±(15 to 20%), limited to ±20%.

All values of differencing signals 62-68 are in percent (%).
The differences for engine 20 are computed as follows:

For N$_1$,
difference N$_1$=actual N$_1$- ideal N$_1$

For N$_2$,
difference N$_2$=actual N$_2$- ideal N$_2$

For EGT, where 600° C. is considered maximum.
different EGT=(actual EGT - ideal EGT)/600 * 100.

For Fuel Flow, where 12000 lbs. per hour is considered maximum.
difference Fuel Flow=(actual fuel flow - ideal fuel flow)/12000 * 100.

For oil pressure, where the maximum is considered to be 100 psi,
difference oil pressure=(actual oil pressure - ideal oil pressure)/100 * 100.

For oil temperature, where the maximum is considered to be 80° C.,
difference oil temperature=(actual oil temperature - ideal oil pressure)/180 * 100.

For oil quantity, where the output signal 40 is limited to a maximum of 1.3 gal,
difference oil quantity=(actual oil quantity - ideal oil quantity)/1.5 * 100.

The caution/warning limits subsection 49 is pro-
grammed to generate any one of a set of limit digital
output signals 52-58 corresponding to a like one of the
first set of actual digital output signals 23-29 when the
value of any one of the actual digital output signals 23-29 as in a caution or warning range of values. The
35 corresponding one of any one of limited output signals
52-58 in a caution or warning range is scaled to match
the corresponding one of the difference output signals
62-68 of the differencing function subsection 51, where
difference output signals 62-68 of the differencing function
subsection 51, where

normal:  ±(0 to 10%)
caution:  ±(10 to 15%), and
warning:  ±(15 to 20%), limited to ±20%.

All values of limit signals 52-58 are in percent (%).
For example, when N$_1$ has a normal range of 0% to
94%: a caution range from 94% to 100.1%, and a warn-
ing range of 100.1% or greater the following is used:
If N$_1$ is less than 89.0%, then
Caution...Warning...N$_1$=0.
If $N_1$ is equal to or greater than 89.0% but less than 94.0%, then

Caution _ Warning $N_1 = 2 \times (N_1 - 89.0)$. 

If $N_1$ is equal to or greater than 94.0% but less than 100.1%, then

Caution _ Warning $N_1 = 10 + 5 \times \left( \frac{N_1 - 94.0}{100.1 - 94.0} \right)$. 

If $N_1$ is equal to or greater than 100.1% but less than 115.0%, then

Caution _ Warning $N_1 = 15 + 5 \times \left( \frac{N_1 - 100.0}{115.0 - 100.0} \right)$. 

If $N_1$ is equal to or greater than 115.0%, then

Caution _ Warning $N_1 = 20$. 

For $N_2$, where
the normal region is 0% to 94%,
the caution region is from 94% to 100.0%,
and the warning region is 100.0% or greater, the following is used:

If $N_2$ is less than 89.0%, then

Caution _ Warning $N_2 = 0$. 

If $N_2$ is equal to or greater than 89.0% but less than 94.0%, then

Caution _ Warning $N_2 = 2 \times (N_2 - 89.0)$. 

If $N_2$ is equal to or greater than 94.0% but less than 100.0%, then

Caution _ Warning $N_2 = 10 + 5 \times \left( \frac{N_2 - 94.0}{100.0 - 94.0} \right)$. 

If $N_2$ is equal to or greater than 100.0% but less than 115.0%, then

Caution _ Warning $N_2 = 15 + 5 \times \left( \frac{N_2 - 100.0}{115.0 - 100.0} \right)$. 

If $N_2$ is equal to or greater than 115.0%, then Caution _ Warning $N_2 = 20$. 

Fuel Flow has neither a caution nor a warning region. Therefore,

Caution _ Warning Fuel Flow = 0. 

For EGT, where
the normal region is 0°C to 535°C,
the caution region is from 535°C to 570°C,
and the warning region is 570°C or greater, the following is used:

If EGT is less than 500°C, then

Caution _ Warning EGT = 0. 

If EGT is equal to or greater than 500°C but less than 535°C, then

Caution _ Warning EGT = (EGT - 500) / 3.5 

If EGT is equal to or greater than 535°C but less than 570°C, then

Caution _ Warning EGT = (EGT - 535) / (570 - 535). 

If EGT is equal to or greater than 570°C but less than 600°C, then

Caution _ Warning EGT = 15 + 5 \times \left( \frac{EGT - 570}{600 - 570} \right)$. 

If EGT is equal to or greater than 600°C, then

Caution _ Warning EGT = 20. 

For oil pressure, when
the low warning is 35 psi or less,
the low caution is 35 to 40 psi,
and the upper warning region is 55 psi or greater, the following is used:

If oil pressure is less than 35 psi, then

Caution _ Warning Oil Pressure = 25 \times (oil Pressure - 35). 

If oil pressure is equal to or greater than 35 psi but less than 40 psi, then

Caution _ Warning Oil Pressure = 20. 

If oil pressure is equal to or greater than 40 psi but less than 45 psi then Caution _ Warning Oil Pressure = 15 \times (40 - oil Pressure). 

If oil pressure is equal to or greater than 45 psi but less than 55 psi, then

Caution _ Warning Oil Pressure = (oil Pressure - 45). 

If oil pressure is equal to or greater than 55 psi but less than 70 psi, then

Caution _ Warning Oil Pressure = 15 \times (oil Pressure - 55) / 3.0. 

If oil pressure is equal to or greater than 70 psi, then

Caution _ Warning Oil Pressure = 20. 

For oil temperature, where
the low warning is 40°C or less,
the normal region is 40° to 120°C,
the upper caution region is 120° to 157°C,
and the upper warning region is 157°C or greater, the following is used:

If oil temperature is less than 40°C, then

Caution _ Warning Oil Temperature = 25 \times (oil Temperature - 40). 

If oil temperature is equal to or greater than 40°C but less than 120°C, then

Caution _ Warning Oil Temperature = (oil Temperature - 40) / 80. 

If oil temperature is equal to or greater than 120°C but less than 157°C, then

Caution _ Warning Oil Temperature = 10 + 5 \times \left( \frac{oil Temperature - 120}{157 - 120} \right)$. 

If oil temperature is equal to or greater than 157°C, then

Caution _ Warning Oil Temperature = 20.
If oil temperature is equal to or greater than 157° C., but less than 180° C., then

\[ \text{Caution-Warning-Oil-Temperature} = \frac{156}{180} \times 100\% \]

For oil quantity, where the low warning is 1 gallon or less and the normal region is greater than 1 gallon, the following is used:

If oil quantity is less than 1 gallon, then

\[ \text{Caution-Warning-Oil-Quantity} = \frac{1}{10 \times 1.05} \]

If oil quantity is equal to or greater than 1 gallon but less than 1.05 gallons, then

\[ \text{Caution-Warning-Oil-Quantity} = 100 \times \frac{1}{1.05} \]

If oil quantity is equal to or greater than 1.05 gallons, then

\[ \text{Caution-Warning-Oil-Quantity} = (1 - 1.05) \]

For the following inputs from the Worst Case Selector Function subsection 71, the following outputs are produced by the Shaping Function subsection 81:

- input = 0%, output = 0% of \( 3h \)
- input = \( -b_1 \), output = \( -31\% \) of \( 3h \)
- input = \( -b_2 \), output = \( -66\% \) of \( 3h \)
- input = \( -b_3 \), output = \( -100\% \) of \( 3h \)
- input = \( b_1 \), output = \( 31\% \) of \( 3h \)
- input = \( b_2 \), output = \( 66\% \) of \( 3h \)
- input = \( b_3 \), output = \( 100\% \) of \( 3h \)

Continuing the description of display section 12, FIG. 2 illustrates both the thrust indicator 48 and monitoring indicator 91 in greater detail. The thrust indicator 48 provides aircraft control information directly to the pilot based on the actual and ideal functional model performance parameters of two engines 20 designated as left (L) and right (R) respectively. The pilot may use this control information directly in making power adjustments without any additional “look-up” of engine performance data.

The monitoring indicator 91 simultaneously provides comparison information relative to determined limits of the actual and “ideal” engine performance parameters in both digital and graphic form. In addition the monitoring indicator 91 may provide an output signal 92 that preempts the thrust indicator 48 display when anyone of the monitored engine performance parameters exceeds a determined value as previously described as an added safety feature.

Thrust indicator 48 includes two vertical, horizontally spaced thrust scales 93 and 94 that represent the maximum available thrust from an “ideal” functional model of actual engines 20L and 20R respectively. The thrust scales 93 and 94 are scaled in percentage increments of 10% from 10% to 110% percent, the maximum available thrust of an “ideal” functional model of engines 20L and 20R. On each thrust scale 93 and 94, 100% equals the maximum available thrust without exceeding any one of the performance parameters of engines 20L and 20R, which are predicted for the “ideal” functional model as previously explained. The actual available thrust value is displayed digitally in pounds at the top of each scale thrust 93 and 94, in this instance 13,960 for both engines 20L and 20R as previously calculated.

Each thrust scale 93 and 94 includes a colored (red) range-thrust warning limit 96 that begins at the 100% mark and extends through 110% and a predicted, colored (yellow) range thrust caution limit 97 that begins for instance at 90% in this example and extends through 100%. The yellow range thrust caution limit 97 is based on a computed maximum continuous thrust obtained from the “ideal” functional model.

Under normal operating conditions, the system is arranged so that no other engine performance parameter (N₁, N₂, or EGT) will be within the yellow caution range 106 unless actual thrust value is in the yellow caution range 97. Also, under normal operations, no other engine parameter (N₁, N₂, or EGT) will be within the red warning range 107 unless the actual thrust value.
(output signal 45) for the respective engine 20L or 20R is in the red warning range 96.
A colored (yellow) thrust reference pointer 101 provided for each scale 93 and 94 that displays a reference value selected by the pilot for each engine 20L and 20R. This selected value, in percent of maximum available thrust, is presented in digital form (98% for example) at reference point 101 for a determined time interval (5 seconds for instance) following a change in the selected reference value.

The thrust indicator 48 also includes a colored (white) vertical thrust predictor bar 102 which overlays a current or actual thrust bar 103, both of which are positioned between and parallel to the respective scales 93 and 94. The thrust predictor bar 102 is responsive to and displays the calculated and pilot commanded thrust of the "ideal" functional model independent of the actual engines 20L and 20R but based on their then current operating conditions as determined by digital output signals 14, 16, 17, 18, and 21. The predicted commanded thrust is presented both as the colored (white) bar 102 and as a predictor pointer 104 that also includes a digital readout in the percent of maximum available thrust represented by the predicted thrust.

Each pair of actual thrust bars 103 is responsible to and displays the value of the actual thrust represented by a digital output signal 47 for the respective engine 20L or 20R with which associated. The actual thrust indicator bars 103 are multi-colored, reflecting their respective operating values by color-green when outside both the caution range 97 and warning range 96, of scales 93 and 94, yellow in the caution range 97 of the scales 93 and 94, and red in the warning range 96 of the scales 93 and 94. Under normal steady-state operating conditions the position of the predictor bars 102 and actual thrust bars 103 relative to the respective scales 93 and 94 should be generally in agreement.

Referring to FIGS. 2 and 3, monitoring indicator 91 includes a plurality of side-by-side pairs of vertical bar deviation indicators 111-117 for the performance parameters N1, EGT, N2, fuel flow (FF), oil pressure, oil temperature, fuel quantity respectively for the engines 20L and 20R. Each bar indicator 111-117 displays the difference between the actual and estimated value of each performance parameter for each engine 20L and 20R as represented by the output signals 82-88 respectively from the shaping function subsection 81 as previously described. To reduce visual noise or jitter on the display 91 caused by the deviation bars 111-117 growing slightly around the O valve, a shaping function as shown in FIG. 3 is applied to the input deviation to the display. The shaping function diminishes this jitter around the O point, in the center of the caution range CR, and at the limit of the warning range WR. The shaping function also serves to increase the movement of the bars 111-117 around the transition areas between the normal and caution ranges NR and CR and between the caution and warning ranges CR and WR. This shaping function is expressed as follows:

\[
for \, b_1 \leq |X| \leq b_2: \quad y = h - h \left( \frac{X - b_1}{b_2 - b_1} \right)
\]

\[
for \, b_3 \leq |X|: \quad y = \frac{b_3}{2} - \frac{X}{2} - \frac{b_3}{2} \sin \left( \frac{X - b_2}{b_2 - b_1} \right)
\]

The actual value of each performance parameter of each engine 20L and 20R is digitally displayed above each bar deviation indicator 111-117.

The face of the monitoring indicator 91 is also divided into normal, caution, and warning ranges (NR, CR, and WR respectively) both above and below a centerline CL that denotes the estimated or predicted value for each of the performance parameters of engines 20L and 20R as shown in FIG. 2. The normal ranges are demarked by the centerline CL which may be green in color and upper and lower range lines -106 which may be yellow in color and demark the beginning of the yellow caution range CR, the limits of which are defined by spaced colored lines 107 (red) demarking the warning range WR. The green, yellow, and red range bars 103 of the thrust indicator 48 correspond to the like green, yellow, and red ranges NR, GR, and WR of the monitor indicator 91.

The normal, caution, and warning ranges associated with the respective engine performance parameters may be -10% to 10%, -15% to -10%, and 10% to 15%, and less than -15% and greater than 15% respectively.

The limitations produced by the caution/warning limits subsection 49 are merged with the deviation values of subsection 51 by the worst case selector subsection 71 as each engine performance parameter approaches each limit. For example, the N1 caution limit which begins at 94% is merged with the N1 deviation value beginning at 89% N1. This merging permits the corresponding N1 deviation bar 111 to just begin transit coming into the yellow caution range CR as N1 reaches 94%. To make the monitor indicator 91 even more quickly and accurately readable by a pilot, both the digital readout and deviation bar 111 are the same color as the associated range CR.

When the deviation of any one of the performance parameters of either engine 20L or 20R reaches the red warning range WR, this condition will be indicated by both a red deviation bar 111-117 and a corresponding red digital readout. A digital warning signal 92 input to the thrust indicator 48 actuates the coloring of bar 103 to red warning for the associated engine 20L or 20R preempting all other engine parameter indicators in the display.

To produce the actual display elements, as shown in FIG. 2 and which are generated by section 12, assume that the display is a conventional cathode ray tube having a face size that is 1000 units across and 575 units high and with the coordinate origin (x=0, y=0) in the lower left corner of the display face.

Given this, the following lines are drawn from (output signal 45) to x=975:

- The CL line, in color green, at y=338.
- The upper WR line 107, in color red, at Y=438.
- The upper CR line 106, in color yellow, at Y=388.
- The lower WR line 107, in color red, at Y=238.
- The lower CR line 106, in color yellow, at Y=288.
It should be noted that the 3h value of FIG. 3 is equivalent to 150 units with the CL line at Y = 338 units. The monitoring vertical bar deviation indicators 111-117 are all centered at Y = 338 units. The horizontal position of each of the centers of each of the plurality of side-by-side pairs of these vertical bar deviation indicators 111-117 are as follows:

<table>
<thead>
<tr>
<th>Number</th>
<th>Left Position</th>
<th>Right Position</th>
</tr>
</thead>
<tbody>
<tr>
<td>111</td>
<td>x = 288</td>
<td>x = 338</td>
</tr>
<tr>
<td>112</td>
<td>x = 390</td>
<td>x = 440</td>
</tr>
<tr>
<td>113</td>
<td>x = 492</td>
<td>x = 542</td>
</tr>
<tr>
<td>114</td>
<td>x = 594</td>
<td>x = 644</td>
</tr>
<tr>
<td>115</td>
<td>x = 696</td>
<td>x = 746</td>
</tr>
<tr>
<td>116</td>
<td>x = 798</td>
<td>x = 848</td>
</tr>
<tr>
<td>117</td>
<td>x = 900</td>
<td>x = 950</td>
</tr>
</tbody>
</table>

The width of each bar is 50 units. The bar is normally centered at x = 95 and at x = 153 respectively. The lower end of each bar begins at Y = 127 units. The relative height of each bar is computed as follows:

height = 320 units * normalize predicted thrust (59)/100. The width of each bar is 20 units. The predictor pointer 104 is also positioned at this height. The digital value displayed as part of the predictor pointer 104 is the predicted thrust value 59 expressed in percent. The bar and pointer are white in color. In addition to the predicted thrust pointers, reference pointers 101 may be displayed. These pointers are yellow in color and may be set anywhere vertically along the thrust scales 93 and 94 respectively. The position is set directly by the pilot.

The invention thus continuously and simultaneously displays to a pilot the maximum amount of thrust or power an engine or engines can produce under the current operating conditions, the amount of thrust commanded; and the actual thrust the engine or engines are producing. Additionally, all normal engine thrust limiting parameters (N1, N2, and N3) are continuously and simultaneously used to compute the thrust range markings displayed on thrust indicator 48, eliminating the need for cross-checking other engine settings when setting engine thrust or power.

The display of the predicted thrust of an “ideal” engine under the current operating conditions is independent of and permits a direct comparison check between the performance of the operating engine 20 or engines 20L and 20R and the “ideal” or functional model. Thus, the invention determines what the thrust performance parameters of an “ideal” functional model engine should be and compares them simultaneously and continuously against the corresponding actual engine thrust performance parameters.

Under normal operating conditions, the differences between the actual and “ideal” engine performance parameters are continuously and simultaneously displayed by deviation bars 111-117 in the monitor indicator 91. Under near-critical operating conditions, critical limits are factored into the displayed information providing a pilot with the most severe indications whether it is an operating limit or an off-ideal operating condition.

Presenting this information in the form of the column-deviation bars 111-117 permits holistic viewing as the time required to acquire the displayed engine performance parameters is constant relative to the number of displayed engine performance parameters. That is, it requires no more time to process and visually display one set of engine performance parameters than two in accordance with the invention. Thus, the method and systems in accordance with the invention becomes more effective as either the number of engines or engine performance parameters for each engine increases.

By providing a fixed position on the thrust scales 93 and 94 for the maximum available thrust which is computed from the “ideal” engine as explained, a pilot is provided with a fixed, visual reference location thereby reducing visual scan time. Also, by normalizing the thrust (digital output signal 60) with the actual maximum thrust available (digital output signal 50) at current engine operating conditions, the customary use of take off power setting charts is no longer required. The predicted thrust (digital output signal 59) simultaneously provides the pilot an independent visual check between commanded thrust (throttle position) and the actual engine thrust.

By using a simplified, functional model of the monitored engines 20L and 20R, a pilot may compare visually and simultaneously the displayed operational per-
formance of the monitored engines and their respective engine performance parameters. This comparative data provides a means for a pilot to identify degraded or degrading operational conditions. Thus, the thrust and monitor indicators 48 and 91 respectively allow the pilot to determine the status of all engine performance parameters at a single glance and to directly control engine thrust based on the displayed information thereby reducing pilot work load with increased operational safety.

While preferred embodiments of the invention have been described in detail, numerous revisions and modifications can be made within the principles of the invention which is limited only by the appended claims.

What is claimed is:

1. A method of monitoring and displaying the actual, predicted, and available thrust of an aircraft engine under determined operating conditions comprising the steps of:
   - calculating the actual thrust of said aircraft engine under said determined operating conditions and generating an output signal representative thereof;
   - calculating the predicted thrust of a functional model of said aircraft engine under said determined operating conditions and generating an output signal representative thereof;
   - calculating the available thrust of said functional model of said aircraft engine under said determined operating conditions; and
   - simultaneously displaying the representative values of said actual, predicted, and available thrusts.

2. The invention as defined in claim 1 wherein said actual, predicted, and available thrusts are displayed in digital and graphic form.

3. The invention as defined in claim 2 wherein the value of said actual thrust is displayed in different colors representing determined value ranges.

4. The method of claim 1 including the steps of:
   - generating a first set of digital output signals representative of one or more performance parameters of said aircraft engine under said determined operating conditions;
   - calculating the value of one or more performance parameters of said functional model of said aircraft engine under said determined operating conditions and corresponding to the performance parameters of said aircraft engine;
   - generating a second set of output signals representative of the values of said performance parameters of said functional model and simultaneously displaying the values of said first and second sets of output signals.

5. The method of claim 4 wherein the values of said first set of output signals are displayed digitally and the values of said second set of output signal are displayed graphically in different colors representing determined value ranges.

6. The method of claim 5 wherein the colored value ranges of the actual thrust and said second set of output signal are identical.

7. A method of monitoring and displaying one or more selected performance parameters of an aircraft engine comprising the steps of:
   - sensing the value of said one or more selected performance parameters of said aircraft engine under determined operating conditions;
   - generating a first set of actual value output signals representative of the actual value of each of said one or more selected performance parameters; and
   - predicting the value of corresponding performance parameters of an "ideal" functional model of each of said one or more performance parameters of said aircraft engine under said determined operating conditions;

8. The method of claim 7 wherein said determined operating parameters include air temperature, air pressure, mach, and bleed valve position.

9. The method of claim 8 wherein each of said performance parameters of said functional model equal and are of the following form:

   \[ A_0 + A_1x + A_2x^2 + A_3x^3 + A_4x^4 + A_5x^5 + A_6x^6 \]

10. The method of claim 7, wherein said one or more actual and functional model's ideal engine performance parameters include:
   - low pressure compressor rotational speed (N1), high pressure compressor rotational speed (N2), exhaust gas temperature (EGT), fuel flow (FF), oil pressure; oil temperature; and oil quantity.

11. The method of claim 7 including the step of visually displaying when any one of said output signals in said first set of output signals exceeds the limits of a determined value range.

12. The method as defined in claim 7 including the step of visually displaying when any one of said signals in said third set of output signals exceeds the limits of any one of or more determined value ranges.

13. The method of claim 12 including the step of establishing normal, caution, and warning ranges having determined value limits for each of said signals in said third set of output signals.

14. The method of claim 13 including the step of applying a shaping function to said output signals in said third set of output signals as each approaches a determined value of 0, b1, or b2 at the beginning of each of said normal, caution, and warning difference value ranges respectively, said shaping function at each of said difference values 0, b1, and b2 being derived as follows:

   \[ f(x) = 1 - \cos\left(\frac{\pi}{2} \left(\frac{x}{b_1}\right)\right) \]
21. The invention of claim 16 including warning limit means for determining when any one of said first set of actual value output signals exceeds a determined warning limit value; and

means on said monitoring display indicator for indicating when such condition occurs.

22. The invention as defined in claim 4 wherein said first group of output signals includes an EPR output signal representative of the engine pressure ratio of said jet engine under said determined operating conditions and said second means includes means responsive to said determined operating conditions and said EPR output signal for calculating the actual thrust of said aircraft engine and generating an output signal representative thereof;

a display panel; and

thrust indicator display means for directly displaying the value of said calculated actual thrust on said display panel.

23. The invention as defined in claim 22 wherein said computer means includes means responsive to said determined aircraft operating conditions for calculating and generating output signals representative of the values of the predicted and available thrusts of said functional model of said aircraft when operated under said determined operating conditions; and

thrust indicator display means for directly displaying the value of said respective available, predicted, and actual thrusts.

24. The invention as defined in claim 23 wherein said respective display means include a common display panel and said values of said actual, predicted, and available thrusts are simultaneously displayed.

25. The invention as defined in claim 24 including circuit means interconnecting said monitoring indicator means and said thrust indicator means for displaying on said thrust indicator means when any one of said actual engine performance parameters as represented by any one of said one or more output signals in said first set of output signals exceeds a determined value.

26. A system for monitoring and displaying the actual, predicted and available thrust of a jet aircraft engine under determined operating conditions comprising a first computer means responsive to said determined operating conditions for calculating the actual thrust of said aircraft engine based on said determined conditions and generating an actual thrust output signal representative thereof; said computer means including a second computer means responsive to said determined operating conditions for calculating the predicted and available thrust of a functional model of said aircraft engine under said determined operating conditions and generating signals representative of the values of said predicted and available thrusts: and

thrust indicator display means for simultaneously displaying the values of said actual, predicted, and available thrusts.

27. The system as defined in claim 26 including means for displaying the respective thrust values digitally and graphically. * * * *