An Object-Oriented Computer Code for Aircraft Engine Weight Estimation

Michael T. Tong
Glenn Research Center, Cleveland, Ohio

Bret A. Naylor
DB Consulting Group, Cleveland, Ohio

December 2009
NASA STI Program . . . in Profile

Since its founding, NASA has been dedicated to the advancement of aeronautics and space science. The NASA Scientific and Technical Information (STI) program plays a key part in helping NASA maintain this important role.

The NASA STI Program operates under the auspices of the Agency Chief Information Officer. It collects, organizes, provides for archiving, and disseminates NASA's STI. The NASA STI program provides access to the NASA Aeronautics and Space Database and its public interface, the NASA Technical Reports Server, thus providing one of the largest collections of aeronautical and space science STI in the world. Results are published in both non-NASA channels and by NASA in the NASA STI Report Series, which includes the following report types:

- **TECHNICAL PUBLICATION.** Reports of completed research or a major significant phase of research that present the results of NASA programs and include extensive data or theoretical analysis. Includes compilations of significant scientific and technical data and information deemed to be of continuing reference value. NASA counterpart of peer-reviewed formal professional papers but has less stringent limitations on manuscript length and extent of graphic presentations.

- **TECHNICAL MEMORANDUM.** Scientific and technical findings that are preliminary or of specialized interest, e.g., quick release reports, working papers, and bibliographies that contain minimal annotation. Does not contain extensive analysis.

- **CONTRACTOR REPORT.** Scientific and technical findings by NASA-sponsored contractors and grantees.

- **CONFERENCE PUBLICATION.** Collected papers from scientific and technical conferences, symposia, seminars, or other meetings sponsored or cosponsored by NASA.

- **SPECIAL PUBLICATION.** Scientific, technical, or historical information from NASA programs, projects, and missions, often concerned with subjects having substantial public interest.

- **TECHNICAL TRANSLATION.** English-language translations of foreign scientific and technical material pertinent to NASA's mission.

Specialized services also include creating custom thesauri, building customized databases, organizing and publishing research results.

For more information about the NASA STI program, see the following:

- Access the NASA STI program home page at [http://www.sti.nasa.gov](http://www.sti.nasa.gov)
- E-mail your question via the Internet to [help@sti.nasa.gov](mailto:help@sti.nasa.gov)
- Fax your question to the NASA STI Help Desk at 443–757–5803
- Telephone the NASA STI Help Desk at 443–757–5802
- Write to:
  NASA Center for AeroSpace Information (CASI)
  7115 Standard Drive
  Hanover, MD 21076–1320
An Object-Oriented Computer Code for Aircraft Engine Weight Estimation

Michael T. Tong
Glenn Research Center, Cleveland, Ohio

Bret A. Naylor
DB Consulting Group, Cleveland, Ohio

Prepared for the
Gas Turbine Technical Congress and Exposition (Turbo Expo 2008)
sponsored by the American Society of Mechanical Engineers
Berlin, Germany, June 9–13, 2008

National Aeronautics and
Space Administration

Glenn Research Center
Cleveland, Ohio 44135

December 2009
This work was sponsored by the Fundamental Aeronautics Program at the NASA Glenn Research Center.

Level of Review: This material has been technically reviewed by technical management.

Available from

NASA Center for Aerospace Information
7115 Standard Drive
Hanover, MD 21076–1320

National Technical Information Service
5285 Port Royal Road
Springfield, VA 22161

Available electronically at http://gltrs.grc.nasa.gov
An Object-Oriented Computer Code for Aircraft Engine Weight Estimation

Michael T. Tong
National Aeronautics and Space Administration
Glenn Research Center
Cleveland, Ohio 44135

Abstract

Reliable engine-weight estimation at the conceptual design stage is critical to the development of new aircraft engines. It helps to identify the best engine concept amongst several candidates. At NASA Glenn Research Center (GRC), the Weight Analysis of Turbine Engines (WATE) computer code, originally developed by Boeing Aircraft, has been used to estimate the engine weight of various conceptual engine designs. The code, written in FORTRAN, was originally developed for NASA in 1979. Since then, substantial improvements have been made to the code to improve the weight calculations for most of the engine components. Most recently, to improve the maintainability and extensibility of WATE, the FORTRAN code has been converted into an object-oriented version. The conversion was done within the NASA’s NPSS (Numerical Propulsion System Simulation) framework. This enables WATE to interact seamlessly with the thermodynamic cycle model which provides component flow data such as corrected airflows, temperatures, and pressures, etc. that are required for sizing the components and weight calculations. The tighter integration between the NPSS and WATE would greatly enhance system-level analysis and optimization capabilities. It also would facilitate the enhancement of the WATE code for next-generation aircraft and space propulsion systems. In this paper, the architecture of the object-oriented WATE code (or WATE++) is described. Both the FORTRAN and object-oriented versions of the code are employed to compute the dimensions and weight of a 300-passenger aircraft engine (GE90 class). Both versions of the code produce essentially identical results as should be the case.

Introduction

Engine weight is a key design parameter for any new aircraft. It affects aircraft range and is a key element in fuel burn. Weight is also considered an indicator of engine cost. Reliable engine-weight estimation at the conceptual design stage is critical to the development of new aircraft engines. It helps to identify the best engine concept amongst several candidates.

At the NASA Glenn Research Center (GRC), the Weight Analysis of Turbine Engines (WATE) computer code (Ref. 1), originally developed by Boeing Aircraft, has been used to estimate the engine weight of various conceptual engine designs. The code, written in FORTRAN, was originally developed for NASA in 1979. It calculated the weight and dimension of each major engine component, such as compressor, burner, turbines and frames, primarily using a semi-empirical method augmented with analytical calculations for specific component elements. A database of 29 engines was used to develop empirical relationships used to calculate component weight and dimensions. This method provided an accuracy of approximately ±10 percent of engine weight for the database engine designs.

Since 1979, substantial improvements have been made to the computer code by NASA (Ref. 2), and McDonnell Douglas Corporation, to enhance the capability of WATE and to improve its accuracy. Many of the empirical relationships have been replaced with analytical weight and dimension calculations. The primary method used to calculate weight throughout the code is to calculate material volume and multiply by material density. An approach is used where the stress level, maximum temperature and pressure, material, geometry, stage loading, hub-tip ratio, blade/vane counts, and shaft speed are used to determine the component weight. Flow properties such as corrected mass flow, temperatures, pressures, etc. on each component are obtained from the thermodynamic cycle analysis. This method accounts for more of the individual parts that make up an engine component than an empirical method. A material database, consisting of the material data of most of the commonly-used aerospace materials, was also incorporated into WATE. In addition to engine component weight calculation improvements, a large number of changes were made to the WATE code to improve the graphical output of the program. Component graphic representation has been greatly enhanced to provide a more detailed picture of the flowpath to assist the user in determining the correctness of the flowpath. The code was also modified to allow the user to use controls and limiters when developing the flowpath. This capability saves the user time in assuring the adjacent components will connect correctly. An optimizing capability was added as well to allow optimization of the flowpath on either weight or dimensions.

Recently, to improve the maintainability and extensibility of WATE, GRC analysts converted the FORTRAN code into an object-oriented version. The conversion was done within the NASA’s NPSS (Numerical Propulsion System Simulation) (Ref. 3) framework. This enables WATE to interact seamlessly with the thermodynamic cycle model which provides component flow data such as corrected airflows,
temperatures, and pressures, etc., that are required for sizing the components and weight calculations.

The object-oriented programming is already an established software development method and its advantages over procedure-oriented programming like FORTRAN have been widely recognized (Refs. 4 and 5). The main ideas behind object-orientation are data abstraction, inheritance, polymorphism and dynamic binding. This paper describes the approach to the conversion of the FORTRAN engine-weight estimation code into an object-oriented code using NPSS interpreted language. The engine-weight modeling system and its object-oriented architecture are described. Validation of the code is presented.

WATE++ Architecture

The object-oriented WATE (or WATE++) calculates the weight and dimension of each major engine component, relative to a reference point, usually the design point. The WATE++ architecture is intended to be flexible and extensible. It exploits the capabilities of object-oriented programming (inheritance, polymorphism, and encapsulation), as well as modern object-oriented concepts including framework and component objects. Inheritance is used to concentrate code common to multiple component types in abstract component classes, preventing code duplication and enhancing code maintainability. For example, the 'Axial Compressor class' in Figure 1 represents an abstract ancestor incorporating all functionality common to fan, low-pressure compressor, and high-pressure compressor. Encapsulation enhances code maintainability and readability by concentrating all data declarations and procedures in a single code unit. Polymorphism is the ability of parameters to represent different object classes and is extensively applied in WATE++. It allows the framework to get the correct behavior of each WATE++ element without knowing what the specific type of each one is. For example, the WATE++ has an abstract identifier 'calcGeometry' to calculate the geometry of any component in the model. During simulation, 'calcGeometry' subsequently represents all WATE++ components and runs their geometry calculations so that the engine flowpath can be drawn.

There are three primary object types that are used in the WATE++ calculations: elements, ports, and subelements.

Elements

WATE++ elements perform high level component calculations. Elements may have many plug locations termed sockets, into which computational blocks termed subelements may be attached. The compressor element, for example, contains a socket into which the subelement that does the disk sizing calculation is attached. It also has sockets into which the user may plug subelements for computing the gearbox and frame weights.

Ports

Ports provide data connectivity between elements. In WATE++, ports are used to transfer geometry information (such as radius and axial position) between elements. It also provides mechanical links from one element to another. For example, a port is used to connect the shaft with the compressor and turbine.

Subelements

Subelements perform specific, detailed computations. They generally only work when connected to sockets. WATE++ supports multiple types of subelements that can plug into the element sockets. Each subelement performs detailed computations that impact the element’s overall calculations. The variable-area-nozzle subelement, for example, can be plugged into the nozzle element socket for variable-area nozzle computation. Sockets on an element need not be filled; default values will be used if left empty.

Element and Subelement Classes

The abstract class ‘WATEelement’ encapsulates the common structural components of a gas turbine engine. These component classes and their inheritance hierarchy are shown in Figure 1. In the entire code, the interaction between different classes is only through messages.

Interaction With the Thermodynamic Cycle Model

WATE++ is designed to function with the thermodynamic cycle model within the NPSS framework, as shown in Figure 2. The thermo design point case of the thermodynamic cycle model can be used to provide engine cycle data required for sizing the engine components, or additional off-design points can be run and the output data will be scanned for maximum conditions of airflows, pressures, or temperatures for each component. In order to produce the most accurate weight estimate, the off-design cases should encompass the maximum performance level required for each engine component. All components that contribute weight must also be included in the thermodynamic cycle model.

The development of NPSS was a cooperative effort between NASA and other government agencies, industry, and universities to integrate propulsion technologies with high-performance computing and communication technologies into a complete system for performing detailed full-engine simulations (Fig. 6). It consists of three main elements: (1) the engineering application models, (2) the system software for the simulation environment, and (3) the high-performance computing platform. To facilitate the timely and cost-effective capture of complex physical processes, NPSS uses object-oriented technologies such as C++ objects to encapsulate
individual engine components and Object Request Brokers (ORB’s) from the Common Object Request Broker Architecture for object communication and deployment across heterogeneous computing platforms.

The ultimate goal of NPSS is to create a “numerical test cell” that enables engineers to examine various design options numerically and minimize the number of costly and time-consuming real life tests.

![Diagram](image-url)

Figure 1.—WATE++ component inheritance architecture.
Material Database

The material database in WATE++ consists of material data of most of the commonly-used aerospace materials. The list of materials is shown in Table 1. The object \textit{WATEmaterial} allows the user to enter the material name to specify which material property table to use. Material properties will be automatically set using the material property table unless the user chooses to override them. The user can override a given property by defining a function or table with the same material name within the given \textit{WATEmaterial} object.

WATE++ Input and Output

The basic format of a WATE++ input file is text-based. The input procedures can be broken down into three main steps: (1) creation of objects, (2) assignment of values to variables, and (3) commands. The input file is read sequentially so normally an object will be created then values assigned to its variables and the process repeated with the next object. Once all the objects have been declared then commands are made to the code. An example of WATE++ input for an engine fan is shown in Appendix A.

A \textit{data-viewer} object, which uniquely determines which variables to output as well as its format, has been created to generate WATE++ output file. An example of WATE++ output is shown in Appendix B.
WATE++ Execution

WATE++ components are contained within WATEassembly. Execution of the WATE++ components happens in two phases. The first phase is the "COLLECT" phase, where the WATE++ components query the thermodynamic cycle model as it runs to determine maximum values of temperatures and pressures and other parameters that WATE++ will later use to calculate weights and lengths. At some point, the WATE++ components will be placed in the "CALCULATE" state. The WATEassembly can then be executed directly, by calling the run() function on it, and this will cause all of the WATE++ components to perform their calculations. An engine flowpath plot can be obtained by instantiating a WATEsvgViewer object inside the WATEassembly.

Validation of WATE++ Code

The validation of WATE++ has been done by using it to perform an engine sizing and weight calculation for a 300-passenger aircraft engine (GE90 class). The results are compared with those calculated by the FORTRAN WATE in Table 2. Both versions of the code produce essentially identical results. The flowpath of the engine is shown in Figure 3.

![Figure 3.—300-passenger engine flowpath model generated by WATE++.](image)

| TABLE 2.—A COMPARISON OF ENGINE WEIGHT AND DIMENSIONS OUTPUT BETWEEN WATE++ AND WATE (FORTRAN) |
|------------------------------------------------------------|------|----------------|
| Bare engine, wt./lb                                       | 16430| 16428          | 0.01 |
| Total engine, wt./lb                                      | 18156| 18154          | 0.01 |
| Inlet/nacelle, wt./lb                                     | 1891 | 1892           | 0.05 |
| Engine pod, wt./lb                                        | 20047| 20046          | 0.04 |
| Length, in.                                               | 202.8| 203.0          | 0.10 |
| Pod length, in.                                           | 283.1| 283.3          | 0.08 |
| Fan cowl length, in.                                      | 207.6| 207.6          | ---- |
| Engine maximum diameter, in.                              | 122.7| 122.7          | ---- |
| Nacelle maximum diameter, in.                             | 150.8| 150.9          | 0.07 |
| Engine pod c.g. location, in.                             | 79.4 | 78.6           | 0.25 |

NASA/TM—2009-215656
Summary

In this paper, the architecture of WATE++, an object-oriented engine-weight estimation code, has been described. The code was converted from the structured FORTRAN version of the code, to improve its maintainability and extensibility. It was accomplished by carefully designing the object classes and choosing the exact data type to suit the application. The code has been validated by comparing the results that it calculated with those calculated by the FORTRAN version. Both versions of the code give essentially identical results, as should be the case.

References


Appendix A
Wate++ Input File For a Gas Turbine Engine Fan—An Example

//=======================================
//	 FAN
//=======================================
WATEhiBypassFan WATE_Fan {
    componentRef = "Fan";
    numContainedStages = 1;
    MNin = 0.630;
    MNout = 0.460;
    stg1MaxPR = 1.7;
    hubTipRatioIn = 0.30;
    numBlades_in = 22;
    calcStatorWt = FALSE;
    bladeMaterial.type = "Ti_17";
    statorMaterial.type = "Ti_6Al_4v";
    real bladeMaterial.rho(real) { return 0.096; }
    real statorMaterial.rho(real) { return 0.1; }
    real containmentMaterial.rho(real) { return 0.051; }
    contRingRadialThickness = 2;
    bladeSolidity = 1.5;
    bladeVolumeFactor = 0.024;
    stg1BladeAR = 2.045;
    lastStgBladeAR = 2.045;
    bladeTaperRatio = 1.40;
    maxSpdRatio_in = 1.000;
    s_Nmech = 1.;
    geometryType = "ConstTipRadius";
    radiusChangeStg = 1;
    numStatorBlades_in = 54;
    statorVolumeFactor = 0.14;
    stg1StatorAR = 3.754;
    lastStgStatorAR = 3.754;
    RSlenRatioPreSplit = 0.65;
    RSlenRatioPostSplit = 1.85;
    RSlenRatioBypass = 0.78;
    real caseMaterial.rho(real) { return 0.1; }
    real igvMaterial.rho(real) { return 0.1; }
    ductlenInnerRR = 0.0;
    statorSolidity = 0.0;
    stg1StatorRotorLR = 1.0;
    lastStgStatorRotorLR = 1.0;
    Table TB_PRvsTipSpd(real pratio) {
        pratio = {1., 1.18, 1.36, 1.43, 1.503, 1.581, 1.667,
                  1.775, 1.9}
        utip = {600., 885., 1100., 1200., 1300., 1400.,
                1500., 1600., 1700.}
    }
}
WATEdiskMTC S_Disk {
    shape1 = "OPTIMUM";
    material.type = "Ti_17";
    shaftRef = "WATE_LP_Shaft";
}
WATEframeCustom S_RearFrame {
    isFrontFrame = FALSE;
    isStator = TRUE;
    real material.rho(real) { return 0.16; }
    real supportMaterial.rho(real) { return 0.1; }
    volumeFactor = 0.05;
    aspectRatio_in = 2.90;
    supportThickness = 0.1;
    gapFrameLengthRatio = 0.35;
    numBlades_in = 54;
    passThruComp = "WATE_Duct6";
    connectPoint = "REAR";
    rearBearingRef = "WATE_LP_Shaft.bearing1";
}
WATEtowerShaft S_TowerShaft {
    HPX = 500;
    diamRatio = 0.90;
}

NASA/TM—2009-215656 7
Appendix B

Wate++ Engine Weight Output Summary—An Example

<table>
<thead>
<tr>
<th>Component</th>
<th>WT</th>
<th>COMP LEN</th>
<th>ACCU LEN</th>
<th>UPSTR R11</th>
<th>UPSTR R01</th>
<th>UPSTR R12</th>
<th>UPSTR R02</th>
<th>DOWNSTR R11</th>
<th>DOWNSTR R01</th>
<th>DOWNSTR R12</th>
<th>DOWNSTR R02</th>
</tr>
</thead>
<tbody>
<tr>
<td>WATEa_WATE_Fan</td>
<td>4026</td>
<td>24.498</td>
<td>24.498</td>
<td>18.403</td>
<td>61.343</td>
<td>0.000</td>
<td>0.000</td>
<td>27.554</td>
<td>61.343</td>
<td>27.554</td>
<td>61.343</td>
</tr>
<tr>
<td>WATEa_WATE_Splitter</td>
<td>0</td>
<td>0.000</td>
<td>24.498</td>
<td>27.554</td>
<td>61.343</td>
<td>27.554</td>
<td>61.343</td>
<td>32.791</td>
<td>32.791</td>
<td>61.343</td>
<td></td>
</tr>
<tr>
<td>WATEa_WATE_Duct4</td>
<td>12</td>
<td>4.000</td>
<td>28.498</td>
<td>27.351</td>
<td>32.993</td>
<td>27.554</td>
<td>32.791</td>
<td>27.351</td>
<td>32.993</td>
<td>19.941</td>
<td>26.038</td>
</tr>
<tr>
<td>WATEa_WATE_Core_Nozz</td>
<td>323</td>
<td>15.506</td>
<td>202.785</td>
<td>24.748</td>
<td>36.920</td>
<td>24.748</td>
<td>36.920</td>
<td>27.551</td>
<td>35.074</td>
<td>0.000</td>
<td>0.000</td>
</tr>
<tr>
<td>WATEa_WATE_Duct15</td>
<td>94</td>
<td>32.043</td>
<td>93.854</td>
<td>32.791</td>
<td>61.343</td>
<td>32.791</td>
<td>61.343</td>
<td>32.791</td>
<td>61.343</td>
<td>61.343</td>
<td>61.343</td>
</tr>
<tr>
<td>WATEa_WATE_Byp_Nozz</td>
<td>1665</td>
<td>53.368</td>
<td>147.222</td>
<td>32.791</td>
<td>61.343</td>
<td>32.791</td>
<td>61.343</td>
<td>41.857</td>
<td>61.343</td>
<td>0.000</td>
<td>0.000</td>
</tr>
<tr>
<td>WATEa_WATE_Inlet</td>
<td>1564</td>
<td>60.342</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>18.403</td>
<td>61.343</td>
<td>0.000</td>
</tr>
<tr>
<td>WATEa_WATE_LP_Shift</td>
<td>1126</td>
<td>146.007</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
</tr>
<tr>
<td>WATEa_WATE_HP_Shift</td>
<td>168</td>
<td>18.803</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
<td>0.000</td>
</tr>
</tbody>
</table>

ENGINE MOUNT WEIGHT = 326.46
BASE ENGINE WEIGHT = 16429.8
ACCESSORIES WEIGHT = 1726
TOTAL ENGINE WEIGHT = 18155.8
INLET/NACELLE WEIGHT = 1891.01
TOTAL ENGINE POD WEIGHT = 20046.8

ENGINE LENGTH = 202.785
TOTAL ENGINE POD LENGTH = 263.127
FAN COWL LENGTH = 207.565
ENGINE MAX DIAMETER = 122.685
NACELLE MAX DIAMETER = 150.825
ENGINE POD C.G. LOCATION = 79.4397

<table>
<thead>
<tr>
<th>Component</th>
<th>TOT WT</th>
<th>ROTOR</th>
<th>STATOR</th>
<th>FRAME</th>
<th>LT</th>
<th>LTM</th>
</tr>
</thead>
<tbody>
<tr>
<td>WATEa_WATE_Fan</td>
<td>4026.61</td>
<td>1252.351</td>
<td>1486.884</td>
<td>1287.379</td>
<td>61.812</td>
<td>51.966</td>
</tr>
<tr>
<td>WATEa_WATE_LPC</td>
<td>668.79</td>
<td>363.867</td>
<td>304.922</td>
<td>0.000</td>
<td>19.855</td>
<td>19.855</td>
</tr>
<tr>
<td>WATEa_WATE_HPC</td>
<td>2022.57</td>
<td>1050.818</td>
<td>971.751</td>
<td>0.000</td>
<td>50.900</td>
<td>50.900</td>
</tr>
<tr>
<td>WATEa_WATE_HPT</td>
<td>1744.70</td>
<td>966.672</td>
<td>446.962</td>
<td>331.070</td>
<td>9.503</td>
<td>9.503</td>
</tr>
<tr>
<td>WATEa_WATE_LPT</td>
<td>3544.30</td>
<td>2061.493</td>
<td>939.405</td>
<td>543.404</td>
<td>40.322</td>
<td>30.249</td>
</tr>
</tbody>
</table>
**Title and Subtitle:**
An Object-Oriented Computer Code for Aircraft Engine Weight Estimation

**Authors:**
Tong, Michael, T.; Naylor, Bret, A.

**Abstract:**
Reliable engine-weight estimation at the conceptual design stage is critical to the development of new aircraft engines. It helps to identify the best engine concept amongst several candidates. At NASA Glenn Research Center (GRC), the Weight Analysis of Turbine Engines (WATE) computer code, originally developed by Boeing Aircraft, has been used to estimate the engine weight of various conceptual engine designs. The code, written in FORTRAN, was originally developed for NASA in 1979. Since then, substantial improvements have been made to the code to improve the weight calculations for most of the engine components. Most recently, to improve the maintainability and extensibility of WATE, the FORTRAN code has been converted into an object-oriented version. The conversion was done within the NASA’s NPSS (Numerical Propulsion System Simulation) framework. This enables WATE to interact seamlessly with the thermodynamic cycle model which provides component flow data such as airflows, temperatures, and pressures, etc., that are required for sizing the components and weight calculations. The tighter integration between the NPSS and WATE would greatly enhance system-level analysis and optimization capabilities. It also would facilitate the enhancement of the WATE code for next-generation aircraft and space propulsion systems. In this paper, the architecture of the object-oriented WATE code (or WATE++) is described. Both the FORTRAN and object-oriented versions of the code are employed to compute the dimensions and weight of a 300-passenger aircraft engine (GE90 class). Both versions of the code produce essentially identical results as should be the case.