



US007726115B2

(12) **United States Patent**
Murrow et al.

(10) **Patent No.:** **US 7,726,115 B2**
(45) **Date of Patent:** **Jun. 1, 2010**

(54) **AXIAL FLOW POSITIVE DISPLACEMENT WORM COMPRESSOR**

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(*) Notice: Subject to any disclaimer, the term of this patent is extended or adjusted under 35 U.S.C. 154(b) by 934 days.

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(21) Appl. No.: **11/345,760**

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(22) Filed: **Feb. 2, 2006**

(Continued)

(65) **Prior Publication Data**

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US 2007/0175202 A1 Aug. 2, 2007

(74) *Attorney, Agent, or Firm*—William Scott Andes; Steven J. Rosen

(51) **Int. Cl.**

F02C 3/055 (2006.01)

(57) **ABSTRACT**

(52) **U.S. Cl.** **60/39.45**; 418/48

(58) **Field of Classification Search** 60/39.45; 123/241, 249; 418/48, 51

See application file for complete search history.

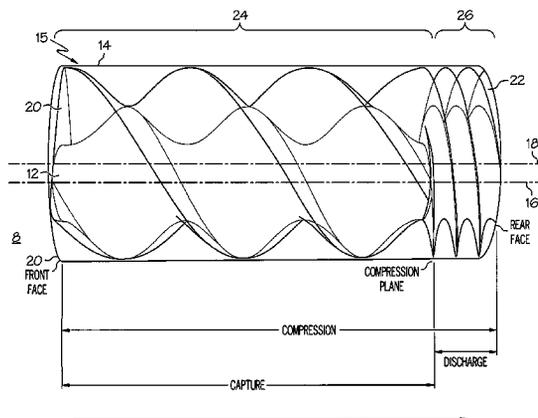
An axial flow positive displacement compressor has an inlet axially spaced apart and upstream from an outlet. Inner and outer bodies have offset inner and outer axes extend from the inlet to the outlet through first and second sections of a compressor assembly in serial downstream flow relationship. At least one of the bodies is rotatable about its axis. The inner and outer bodies have intermeshed inner and outer helical blades wound about the inner and outer axes respectively. The inner and outer helical blades extend radially outwardly and inwardly respectively. The helical blades have first and second twist slopes in the first and second sections respectively. The first twist slopes are less than the second twist slopes. An engine including the compressor has in downstream serial flow relationship from the compressor a combustor and a high pressure turbine drivingly connected to the compressor by a high pressure shaft.

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51 Claims, 9 Drawing Sheets



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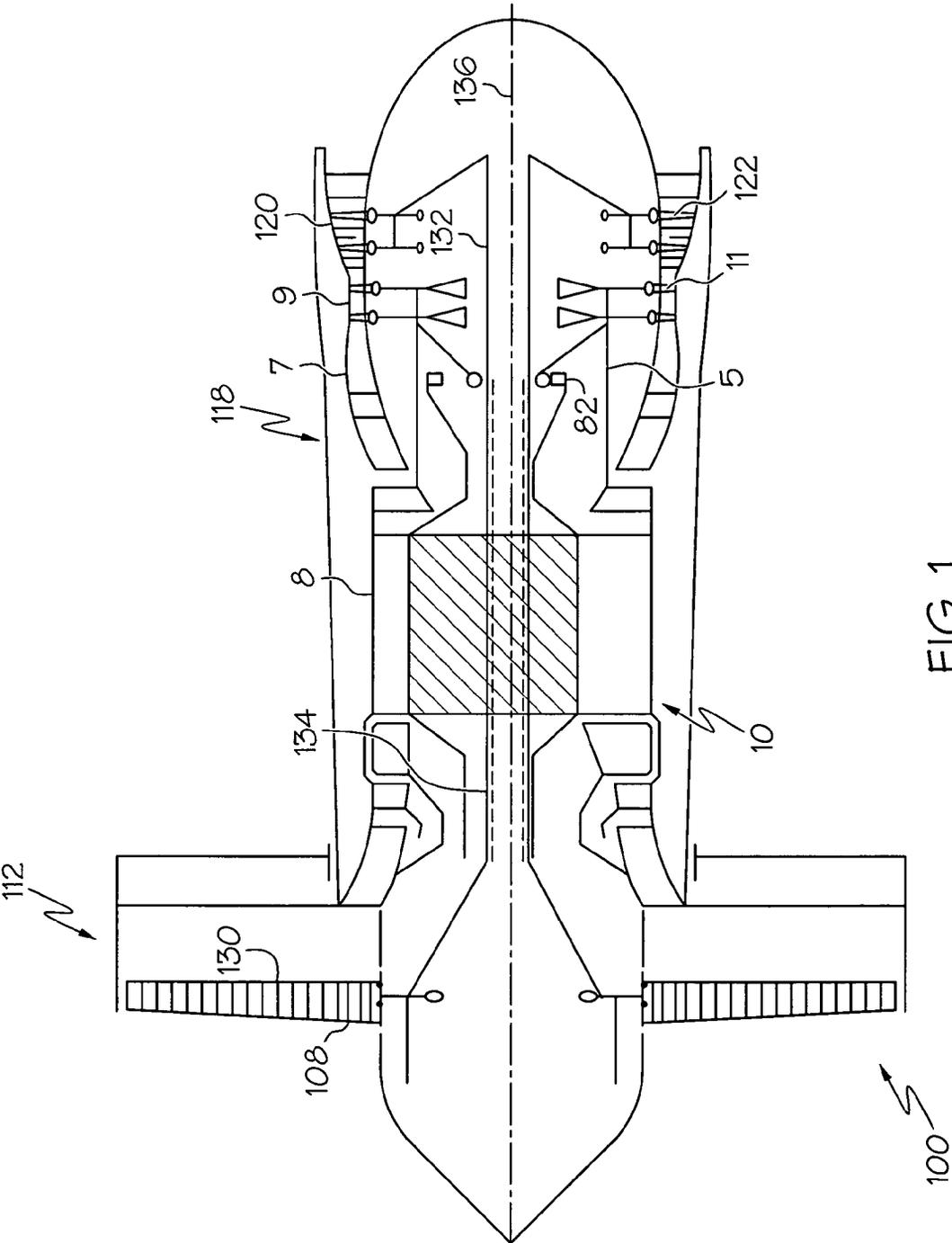


FIG. 1

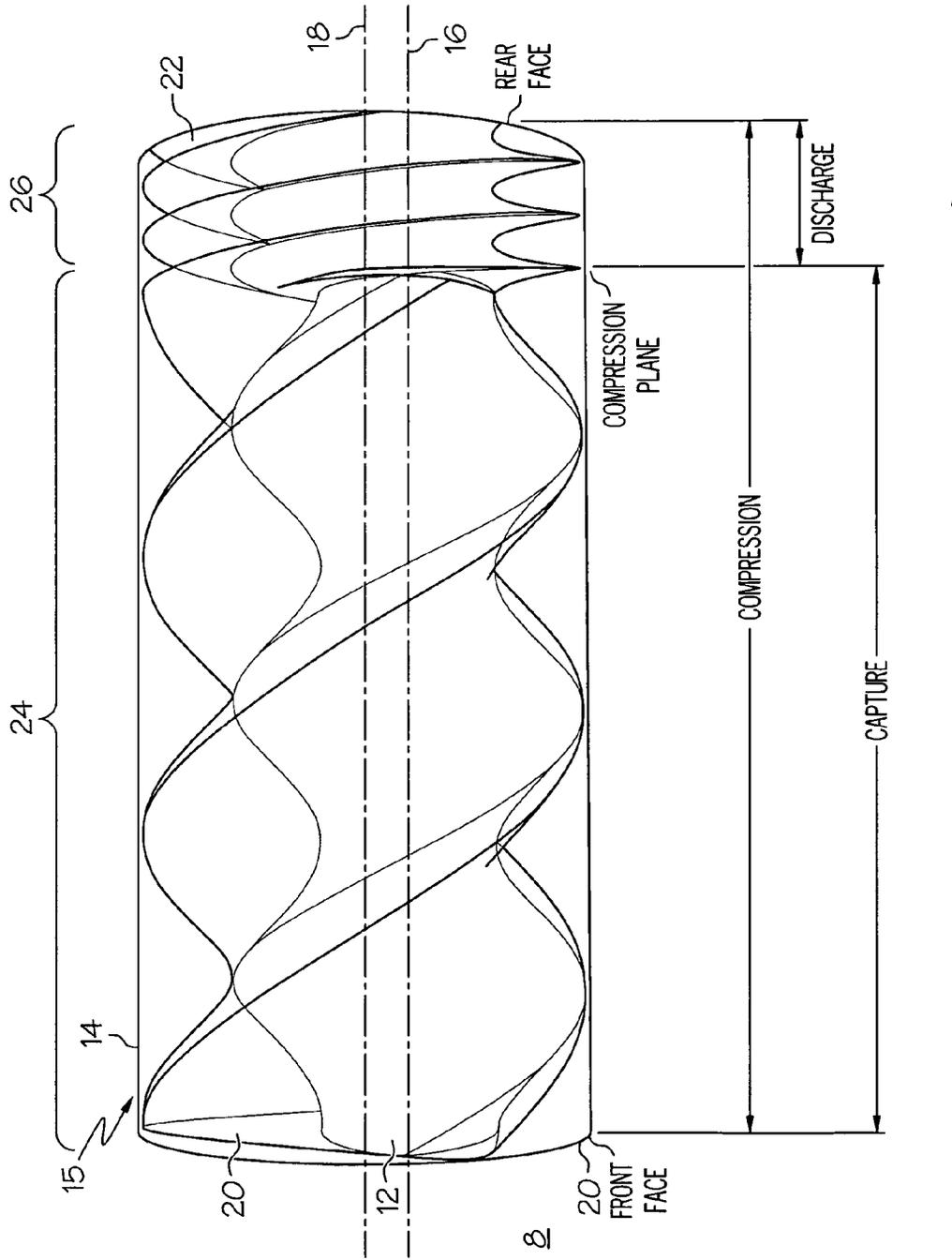


FIG. 2

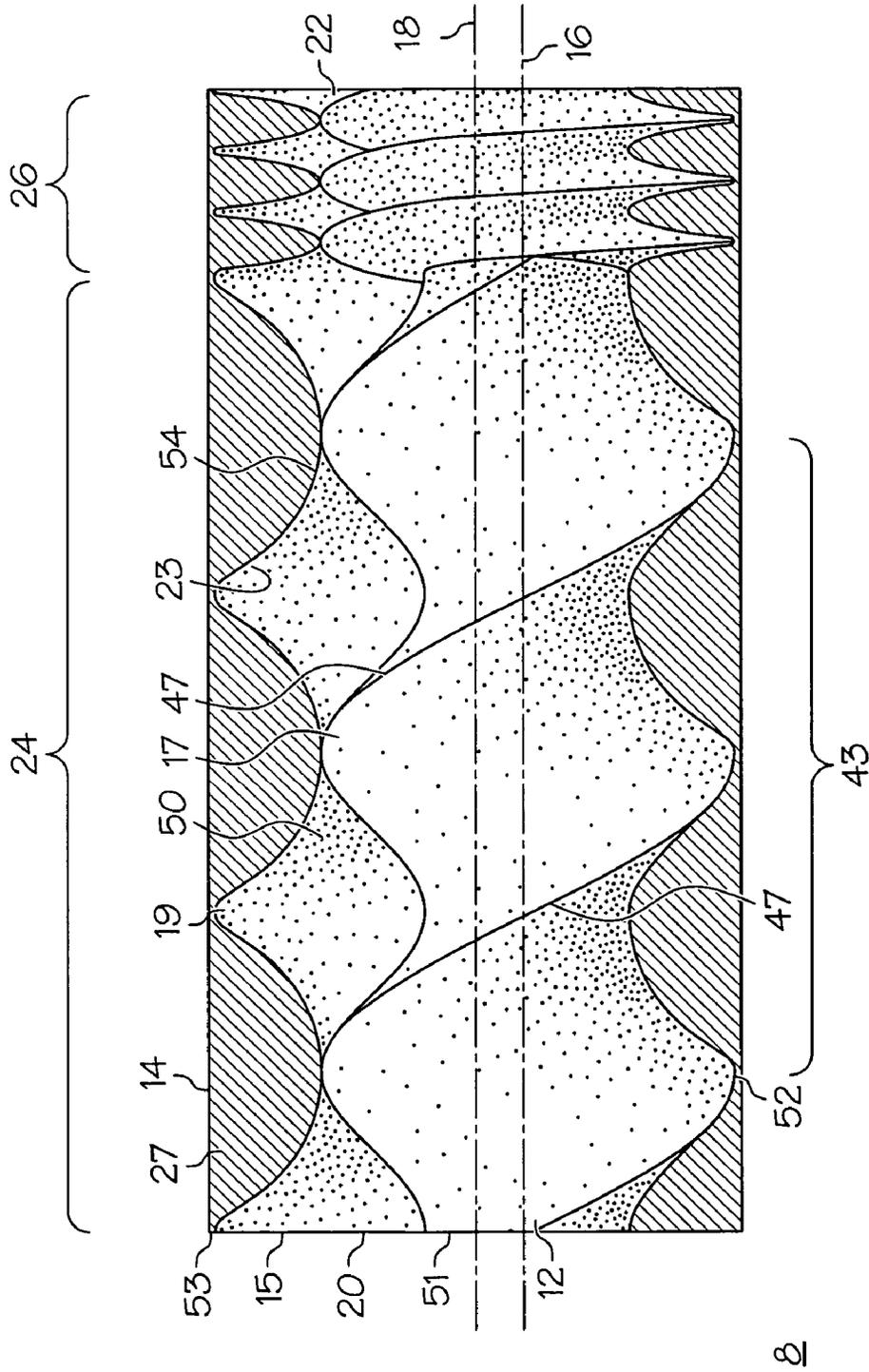


FIG. 3

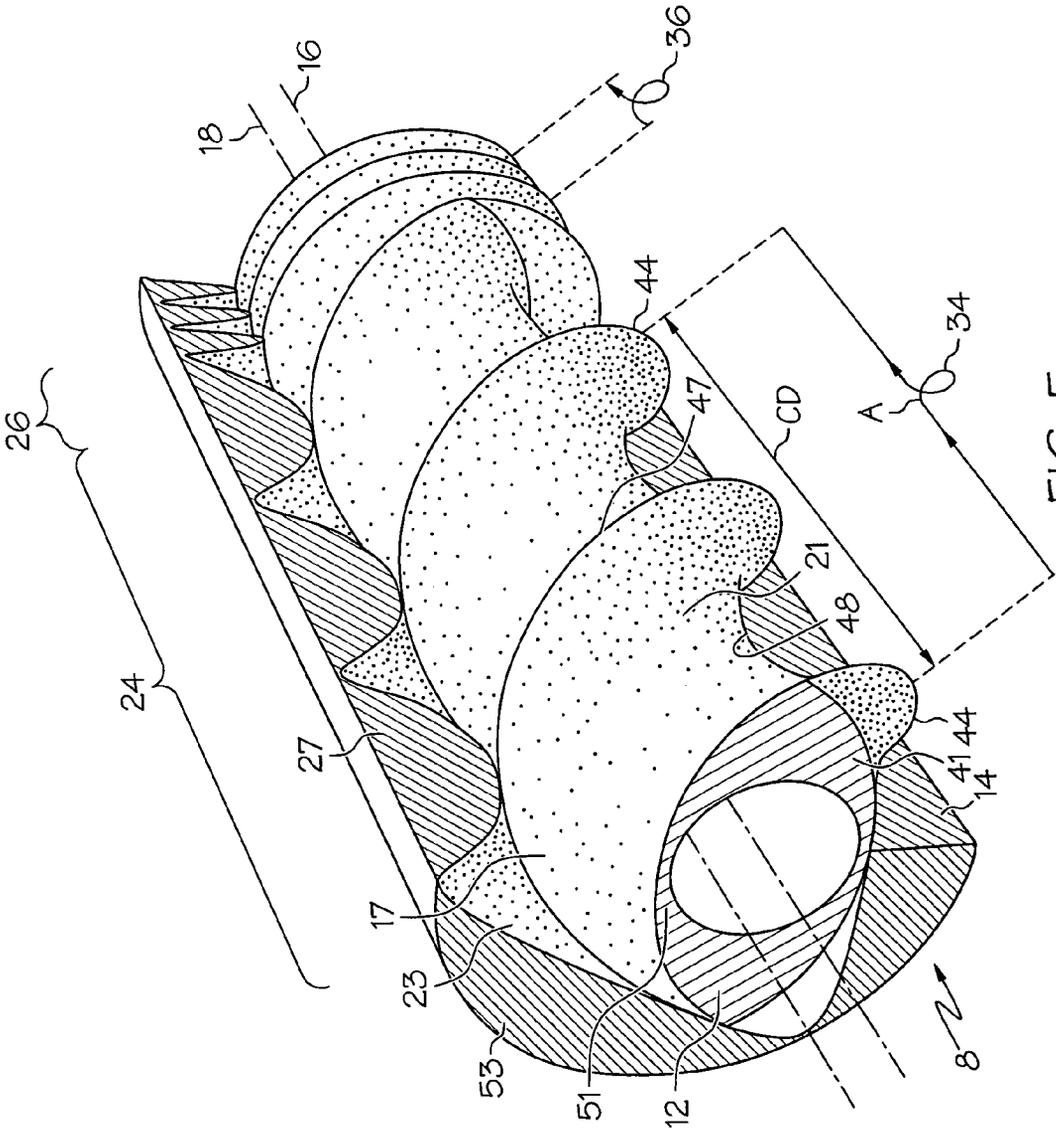


FIG. 5

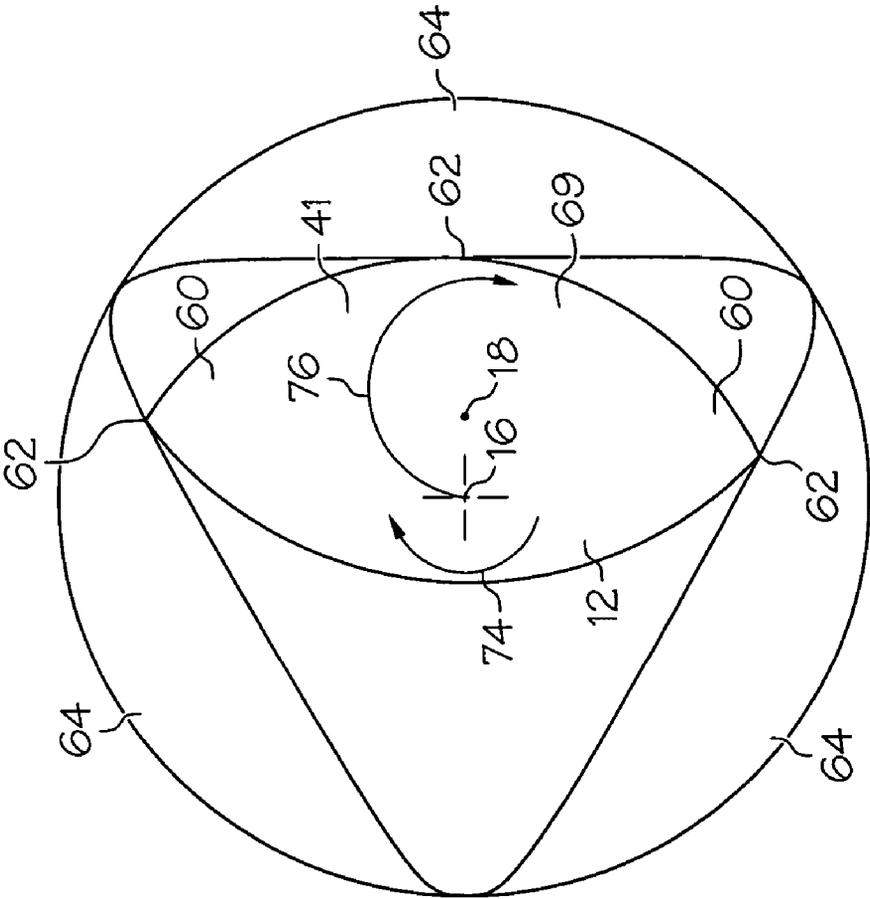


FIG. 6

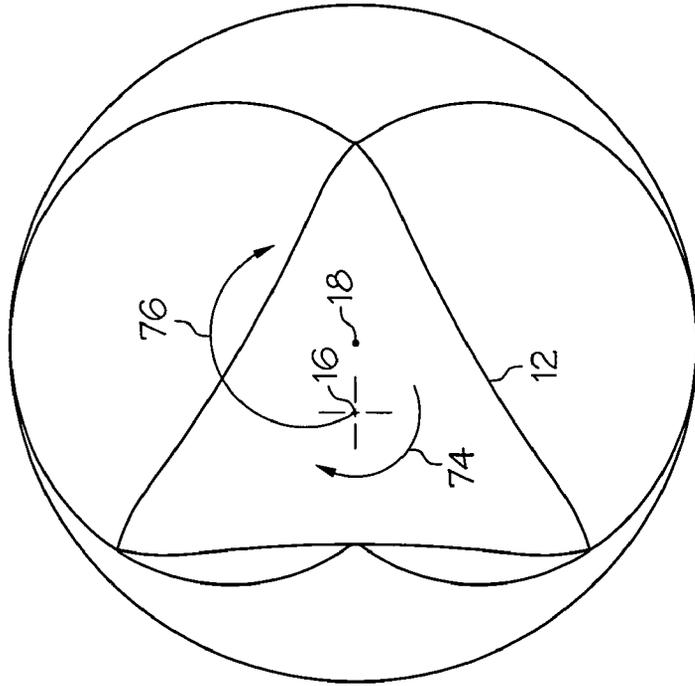


FIG. 8

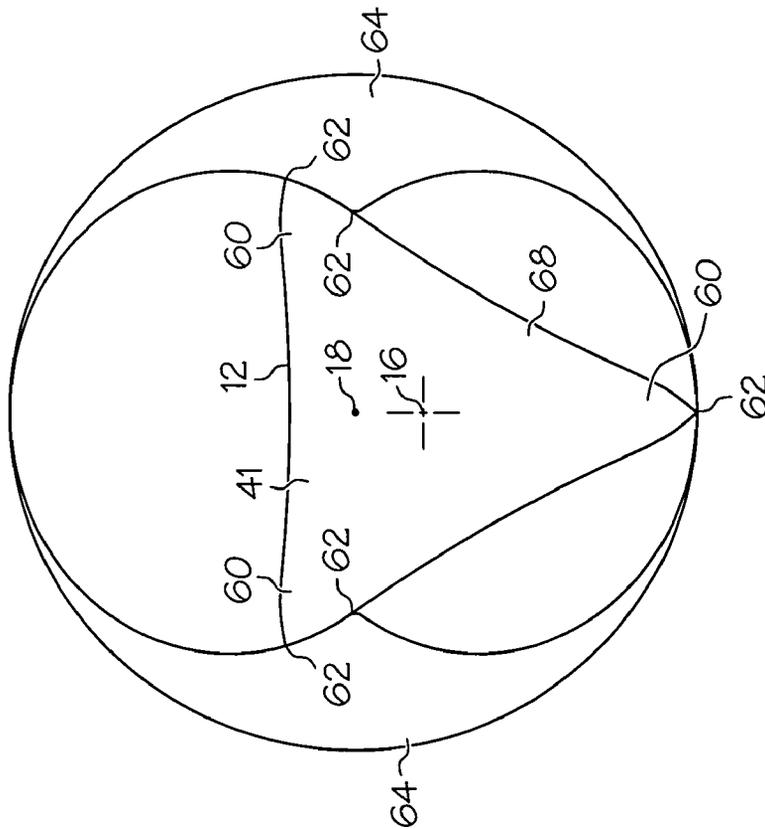


FIG. 7

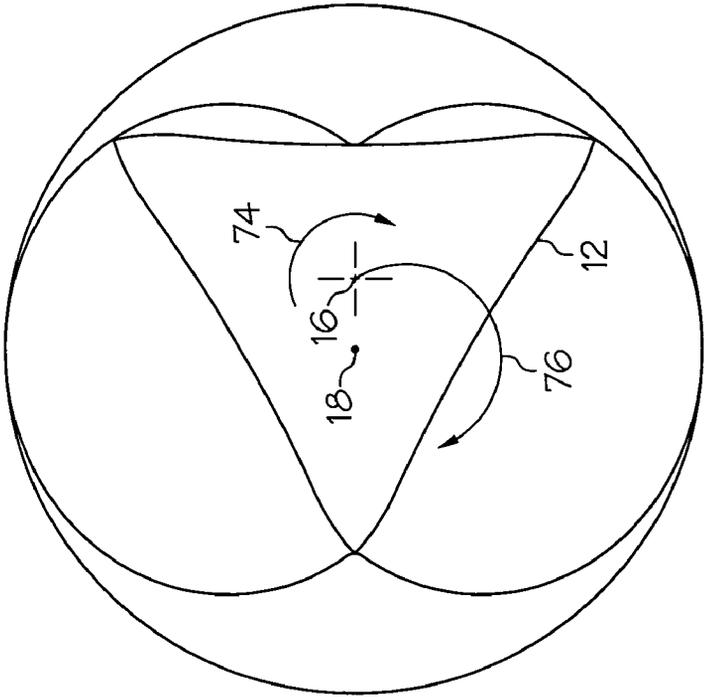


FIG. 10

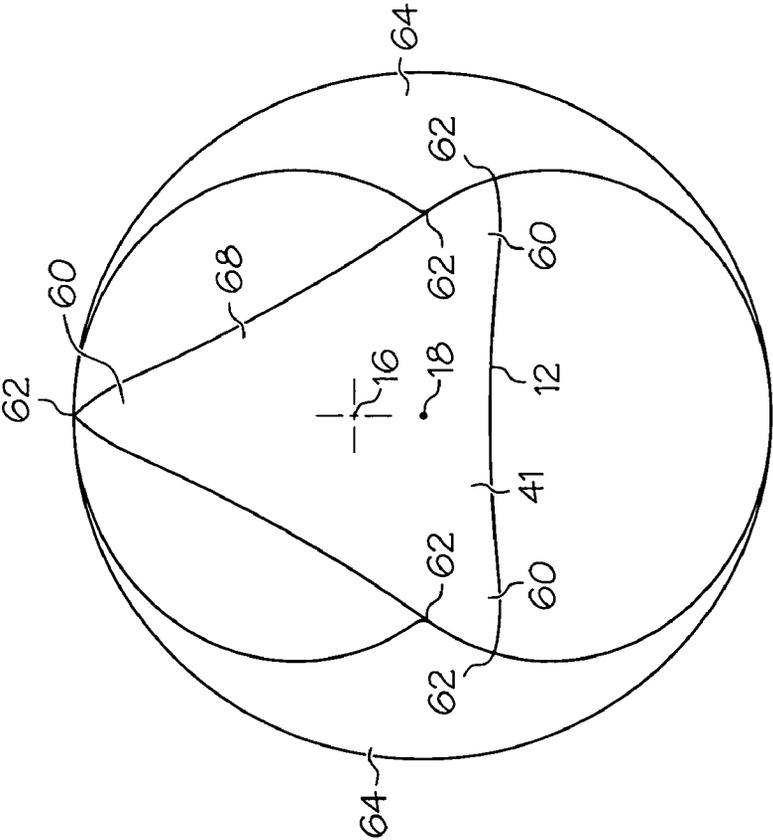


FIG. 9

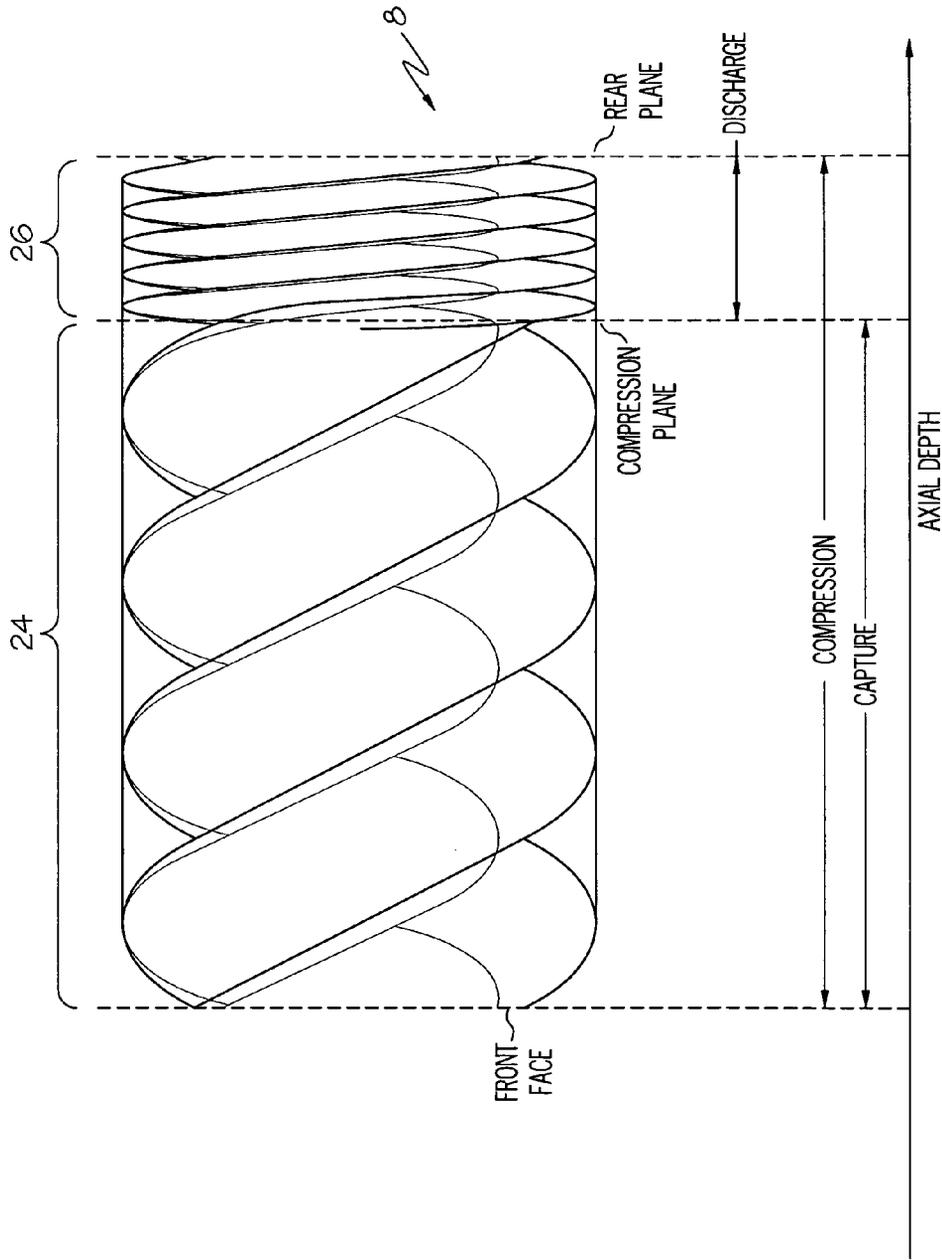


FIG. 11

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AXIAL FLOW POSITIVE DISPLACEMENT WORM COMPRESSOR

The Government has rights to this invention pursuant to
Contract No. NAS3-01135 awarded by the NASA.

BACKGROUND OF THE INVENTION

Field of the Invention

The present invention relates generally to continuous axial
flow compressors and, more particularly, to axial flow posi-
tive displacement compressors and worm and screw com-
pressors.

Compressors are widely used in many applications such as
in gas generators in gas turbine engines. Continuous axial
flow compressors are utilized in a wide range of applications
owing to a combination of desirable attributes such as high
mass flow rate for a given frontal area, continuous near steady
fluid flow, reasonable adiabatic efficiency, and the ability to
operate free from aerodynamic stall and aeromechanical
instability over a wide range of conditions. It is a goal of
compressor and gas turbine manufacturers to have light-
weight, compact, and highly efficient axial flow compressors.
It is another goal to have as few parts as possible in the
compressor to reduce the costs of manufacturing, installing,
refurbishing, overhauling, and replacing the compressor.
Therefore, it is desirable to have a compressor that improves
on all of these characteristics.

BRIEF DESCRIPTION OF THE INVENTION

A continuous axial flow positive displacement compressor
also referred to as a worm compressor includes an inlet axi-
ally spaced apart and upstream from an outlet. The worm
compressor includes a compressor assembly including inner
and outer bodies extending from the inlet to the outlet. The
inner and outer bodies have offset inner and outer axes,
respectively. The compressor assembly has first and second
sections in serial downstream flow relationship. Either or
both bodies may be rotatable. In one embodiment of the
compressor, the inner body is rotatable about the inner axis
within the outer body. The outer body may be rotatably fixed
or rotatable about the outer axis. The inner and outer bodies
have intermeshed inner and outer helical blades wound about
inner and outer axes, respectively. The inner and outer helical
blades extend radially outwardly and inwardly, respectively.

The helical blades have first and second twist slopes in
the first and second sections of the compressor assembly, respec-
tively. A twist slope is defined as the amount of rotation of a
cross-section of the helical element per unit distance along an
axis. The first twist slopes are less than the second twist
slopes. The helical blades in the first section have a sufficient
number of turns to trap charges of gas in the first section
during the compressor's operation. In one embodiment of the
compressor, the number of turns is sufficient to mechanically
trap the charges of gas. In another embodiment of the com-
pressor, the number of turns is sufficient to dynamically trap
the charges of gas. The helical blades in the second section
have a sufficient number of turns to ensure that the leading
edge of the charge is not exposed to the conditions down-

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stream of the compressor until the trailing edge of the charge
has crossed the compression plane, thereby completing the
compression process.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 is a cross-sectional view illustration of an exemplary
aircraft gas turbine engine with a positive displacement con-
tinuous axial flow compressor.

FIG. 2 is a diagrammatic cross-sectional view illustration
of the compressor illustrated in FIG. 1.

FIG. 3 is a diagrammatic partially cut-away perspective
view illustration of helical blade portions of inner and outer
bodies of the compressor illustrated in FIG. 2.

FIG. 4 is a diagrammatic cross-sectional view illustration
of gearing between inner and outer bodies of the compressor
illustrated in FIG. 3.

FIG. 5 is a diagrammatic cut-away perspective view illus-
tration of the helical blade portions of the inner and outer
bodies of the compressor illustrated in FIG. 3.

FIG. 6 is a diagrammatic cross-sectional view illustration
of the inner and outer bodies taken through 6-6 in FIG. 4.

FIGS. 7-10 are diagrammatic cross-sectional view illus-
trations of an alternate inner and outer body configura-
tion at different relative angular positions.

FIG. 11 is a diagrammatic cross-sectional view illustration
of the positive displacement continuous axial flow compres-
sor with the inner and outer bodies illustrated in FIG. 7.

DETAILED DESCRIPTION OF THE INVENTION

Illustrated in FIG. 1 is an exemplary embodiment of a
continuous axial flow positive displacement compressor also
referred to as a worm compressor **8** in a gas turbine engine
100. The worm compressor **8** is part of a gas generator **10** used
to power a low pressure turbine that produces work to drive a
fan **108** in a fan section of the engine **100**. The gas generator
10 may be used to directly drive power consuming devices
such as marine propulsion drives and electrical power gen-
erators or aircraft nozzles or fans. The exemplary embodi-
ment of the gas turbine engine **100** illustrated in FIG. 1 is an
aircraft gas turbine engine having a core engine **118** including
the worm compressor **8** and gas generator **10** downstream of
the fan section **112**.

The core engine **118** includes in downstream serial flow
relationship the worm compressor **8**, a combustor **7**, and a
high pressure turbine **9** (HPT) having high pressure turbine
blades **11** drivingly connected to the worm compressor **8** by a
high pressure shaft **5**. Combustion gases are discharged from
the core engine **118** into a low pressure turbine (LPT) **120**
having low pressure turbine rotor blades **122**. The low pres-
sure turbine rotor blades **122** are drivingly attached to a row of
circumferentially spaced apart fan rotor blades **130** of the fan
108 in the fan section **112** by a low pressure shaft **132** to form
a low pressure spool **134** circumscribing an engine centerline
136. The worm compressor **8** may be used in other applica-
tions including, but not limited to, ground based industrial
and marine gas turbine engines.

Referring to FIGS. 2-5, the worm compressor **8** includes a
compressor assembly **15** having inner and outer bodies **12**, **14**
extending from an inlet **20** to an outlet **22**. The inner body **12**
is disposed within a cavity **19** of the outer body **14**. The inner
and outer bodies **12**, **14** have inner and outer axes **16**, **18**,
respectively. The compressor assembly **15** has first and sec-
ond sections **24**, **26** in serial downstream flow relationship.
The compressor assembly **15** provides continuous flow
through the inlet **20** and the outlet **22** during operation of the

worm compressor 8. Individual charges of gas 50 are captured in and by the first section 24. Compression of the charges 50 occurs as the charges 50 pass from the first section 24 to the second section 26. Thus, an entire charge 50 undergoes compression while it is in both the first and second sections 24 and 26, respectively.

Either or both bodies may be rotatable and, if both bodies are rotatable, they rotate in the same circumferential direction, i.e. either clockwise or counterclockwise, but at different rotational speeds determined by a fixed relationship. If only one body is rotatable, then the other body is fixed. In one embodiment of the generator, the inner body 12 is rotatable about the inner axis 16 within the outer body 14 and the outer body 14 may be rotatably fixed or rotatable about the outer axis 18.

The inner and outer bodies 12, 14 have intermeshed inner and outer helical elements wound about the inner and outer axes 16, 18, respectively. The elements are inner and outer helical blades 17, 27 having inner and outer helical surfaces 21, 23, respectively. The inner helical blades 17 extend radially outwardly from a hollow inner hub 51 of the inner body 12 and the outer helical blades 27 extend radially inwardly from an outer shell 53 of the outer body 14. An inner helical edge 47 along the inner helical blade 17 sealingly engages the outer helical surface 23 of the outer helical blade 27 as they rotate relative to each other. An outer helical edge 48 along the outer helical blade 27 sealingly engages the inner helical surface 21 of the inner helical blade 17 as they rotate relative to each other.

Illustrated in FIG. 4 is a longitudinal cross-section taken through the inner and outer bodies 12, 14. The inner and outer bodies 12, 14 are illustrated in axial cross-section in FIG. 6. The inner body 12 is illustrated herein as having two inner body lobes 60 which correspond to two inner helical blades 17 that form a football or pointed oval-shaped inner body cross-section 69. The outer body 14 has three outer body lobes 64 which correspond to three outer helical blades 27 (illustrated in FIGS. 3 and 4). Note that three sealing points 62 between the inner and outer bodies 12, 14 are illustrated in FIG. 6 but that five periodically overlapping seals exist between the inner and outer helical blades 17, 27 along the entire length of the inner and outer bodies 12, 14.

An alternative configuration of the inner and outer bodies 12, 14 is illustrated in cross-section in FIGS. 7-10. The inner body 12 is illustrated therein as having three inner body lobes 60 which correspond to three inner helical blades 17 which result in a triangularly-shaped inner body cross-section 68 as illustrated in FIG. 7. The outer body 14 has two outer body lobes 64 which correspond to two outer helical blades 27. In general, if the inner body 12 has N number of lobes the outer body 14 will have N+1 or N-1 lobes. Note that five sealing points 62 between the inner and outer bodies 12, 14 are illustrated in FIG. 7, corresponding to five periodically overlapping seals between the inner and outer helical blades 17, 27 along the entire length of the inner and outer bodies 12, 14.

Referring to FIG. 5, the inner and outer helical blades 17, 27 have constant first and second twist slopes 34, 36 in the first and second sections 24, 26, respectively. A twist slope A is defined as the amount of rotation of a cross-section 41 of the helical element (such as the oval-shaped or triangularly-shaped inner body cross-sections 69, 68 illustrated in FIGS. 6 and 7, respectively) per distance along an axis such as the inner axis 16 as illustrated in FIG. 5. Illustrated in FIG. 5 is 360 degrees of rotation of the inner body cross-section 41. The twist slope A is also 360 degrees or 2π radians divided by an axial distance CD between two successive crests 44 along the same inner or outer helical edges 47, 48 of the helical

element such as the inner or outer helical blades 17, 27 as illustrated in FIG. 5. The axial distance CD is the distance required for one full turn 43 of the helix.

The twist slope A of the inner element in each of the sections is different from the twist slope A of the outer element. The ratio of the twist slope A of the outer body 14 to the twist slope A of the inner body 12 is equal to the ratio of the number of inner helical blades 17 blades on the inner body 12 to the number of outer helical blades 27 on the outer body 14. The first twist slopes 34 in the first section 24 are less than the second twist slopes 36 in the second section 26. The helical elements may also be described in terms of helical angle. The helical elements have constant first and second helical angles corresponding to the constant first and second twist slopes 34, 36, in the first and second sections 24, 26, respectively.

Referring again to FIGS. 3-5, the inner helical blade 17 in the first section 24 has a sufficient number of turns 43 to trap the charges of gas 50 in the first section 24 without allowing the compression process to affect the inlet flow field during the compressor's operation. The trapped charges of gas 50 allow positive displacement compression so that higher pressures developed downstream cannot force gas back out the inlet 20. In one embodiment of the gas generator, the number of turns 43 in the first section 24 is sufficient to mechanically trap the charges of gas 50. In another embodiment of the gas generator 10, the number of turns 43 in the first section 24 is sufficient to dynamically trap the charges of gas 50. Mechanically trapped means that the charge 50 is trapped by being closed off from the inlet 20 at an upstream end 52 of the charge 50 before it passes into the second section 26 at a downstream end 54 of the charge 50. Dynamically trapped means that though the downstream end 54 of the trapped charge may have passed into the second section 26, the upstream end 52 of the charge has not yet completely closed. However, at its downstream end 54 by the time a pressure wave from the second section travels to the inlet 20, relative rotation between the bodies will have closed off the charge of gas 50 at its upstream end 52.

For the fixed outer body 14 embodiment, the inner body 12 is cranked relative to the outer axis 18 so that as it rotates about the inner axis 16, the inner axis 16 orbits about the outer axis 18 as illustrated in FIGS. 7-10. The inner body 12 is illustrated as having been rotated about the inner axis 16 from its position in FIG. 7 to its position in FIG. 8 and the inner axis 16 is illustrated as having orbited about the outer axis 18 about 90 degrees. The inner and outer bodies 12, 14 are geared together so that they always rotate relative to each other at a fixed ratio as illustrated by gearing in gearbox 82 in FIGS. 1 and 4.

The inner body 12 rotates about the inner axis 16 with an inner body rotational speed 74 equal to its orbital speed 76 divided by the number of inner body lobes. The number of inner lobes are equal the number of blades. If the inner body 12 rotates in the same direction as its orbital direction, a 2 lobed outer body configuration is used. If the inner body 12 rotates in an opposite orbital direction, a 4 lobed outer body configuration is used. In a first embodiment the inner and outer bodies 12, 14 are both rotatable and the outer body 14 rotates about the outer axis 18 at 1.5 times the rotational speed that the inner body 12 rotates about the inner axis 16. The outer body 14 rotates at a speed equal to the rotational speed of the inner body 12 times the number of lobes on the inner body divided by the number of lobes on the outer body 14.

The twist slopes of the outer body 14 are equal to the twist slopes of the inner body 12 times the number of inner body lobes N divided by the number of outer body lobes M. For the configuration illustrated in FIGS. 7-10 having three inner

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lobes or inner helical blades **17** and two outer lobes or outer helical blades **27**, it takes 900 degrees of rotation of the outer body **14** and 600 degrees of rotation of the inner body **12** to mechanically capture one of the charges of gas **50**. The inner body twist slope is substantially increased going from the first section **24** to the second section **26** at an axial location designated a compression plane as indicated in FIG. **2**. A fairing section between the first and second sections may replace the compression plane if undesirable stress distributions are present. A further comparison between the configuration illustrated in FIGS. **7-10** having three inner lobes and two outer lobes to the configuration illustrated in FIG. **6** having two inner body lobes **60** and three outer body lobes **64** may be had by comparing FIG. **11** to FIG. **2**. Note the number of turns and degrees of rotation of the outer body **14** and the number of turns and degrees of rotation of the inner body **12** needed to capture one of the charges of gas **50** between the upstream and downstream ends **52**, **54** of the charge **50**. Also note the difference in twist slopes of the first and second sections **24**, **26**.

The continuous axial flow positive displacement compressor, referred to herein as a worm compressor **8**, may be used in a wide range of applications and provides high mass flow rate for a given frontal area, continuous near steady fluid flow, and reasonable efficiency over a wide range of operating conditions. It is light-weight and highly efficient and has far fewer parts as compared to other axial compressors which in turn reduces the costs of manufacturing, installing, refurbishing, overhauling, and replacing the compressor. The first embodiment provides a first mode of the compressor's operation disclosed herein in which the inner and outer bodies **12**, **14** both rotate about the inner and outer axes **16**, **18**, respectively. The first mode avoids introducing a centrifugal rotor whirl effect on a support of the compressor and core engine. In a second embodiment the outer body **14** remains static and the inner body **12** simultaneously orbits the outer body's geometric center which is the outer axis **18** and spins about the instantaneous inner body's geometric center which is the inner axis **16**. The second embodiment provides a second mode of the compressor's operation disclosed in which there is only a single rotor rotating potentially simplifying the mechanical design process.

The continuous axial flow positive displacement compressor, referred to herein as a worm compressor **8**, may be used in a wide range of applications and provides reasonably high mass flow rate for a given frontal area, continuous near steady fluid flow, and is expected to provide reasonable efficiency over a wide range of operating conditions. Because the worm compressor operates in a positive displacement mode, it will provide compression levels that are nearly independent of rotor speed over a wide operating range. In thermal engines and other applications, this feature provides a distinct advantage over conventional axial flow compressors, for which compression ratios are directly related to rotor speed. Positive displacement operation also reduces or eliminates aerodynamic stall effects which allows the compressor to be run off-design at compression ratios well above a conventional stall line with the only ill effect being degradation of adiabatic efficiency. The worm compressor is expected to be light-weight, highly efficient, and have far fewer parts than conventional axial compressors which in turn reduces the costs of manufacturing, installing, refurbishing, overhauling, and replacing the compressor.

While there have been described herein what are considered to be preferred and exemplary embodiments of the present invention, other modifications of the invention shall be apparent to those skilled in the art from the teachings

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herein and, it is therefore, desired to be secured in the appended claims all such modifications as fall within the true spirit and scope of the invention. Accordingly, what is desired to be secured by Letters Patent of the United States is the invention as defined and differentiated in the following claims.

What is claimed is:

1. An axial flow positive displacement compressor comprising:

an inlet axially spaced apart and upstream from an outlet, a compressor assembly including an inner body disposed within an outer body and the inner and outer bodies extending from the inlet to the outlet,

the inner and outer bodies having offset inner and outer axes respectively,

at least one of the inner and outer bodies being rotatable about a corresponding one of the inner and outer axes,

the inner and outer bodies having intermeshed inner and outer helical blades wound about the inner and outer axes respectively,

the inner and outer helical blades extending radially outwardly and inwardly respectively,

the inner helical blades extending radially outwardly from an inner hub of the inner body,

the compressor assembly having first and second sections in serial downstream flow relationship extending between the inlet and the outlet,

the inner and outer helical blades having first and second twist slopes in the first and second sections respectively, and

the first twist slopes being less than the second twist slopes.

2. A compressor as claimed in claim **1** further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

3. A compressor as claimed in claim **2** further comprising the number of turns being sufficient to mechanically trap the charges of gas.

4. A compressor as claimed in claim **2** further comprising the number of turns being sufficient to dynamically trap the charges of gas.

5. A compressor as claimed in claim **1** further comprising the outer body being rotatable about the outer axis and the inner body and being rotatable about the inner axis.

6. A compressor as claimed in claim **5** further comprising the inner and outer bodies being geared together in a fixed gear ratio.

7. A compressor as claimed in claim **6** further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

8. A compressor as claimed in claim **7** further comprising the number of turns being sufficient to mechanically trap the charges of gas.

9. A compressor as claimed in claim **7** further comprising the number of turns being sufficient to dynamically trap the charges of gas.

10. A compressor as claimed in claim **1** further comprising the outer body being rotatably fixed about the outer axis and the inner body being orbital about the outer axis.

11. A compressor as claimed in claim **10** further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

12. A compressor as claimed in claim **11** further comprising the number of turns being sufficient to mechanically trap the charges of gas.

13. A compressor as claimed in claim 12 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

14. A engine comprising:

in downstream serial flow relationship an axial flow positive displacement compressor, a combustor, and a high pressure turbine drivingly connected to the compressor by a high pressure shaft,

the compressor having an inlet axially spaced apart and upstream from an outlet,

a compressor assembly including an inner body disposed within an outer body and the inner and outer bodies extending from the inlet to the outlet,

the inner and outer bodies having offset inner and outer axes respectively,

at least one of the inner and outer bodies being rotatable about a corresponding one of the inner and outer axes,

the inner and outer bodies having intermeshed inner and outer helical blades wound about the inner and outer axes respectively,

the inner and outer helical blades extending radially outwardly and inwardly respectively,

the inner helical blades extending radially outwardly from an inner hub of the inner body,

the compressor assembly having first and second sections in serial downstream flow relationship extending between the inlet and the outlet,

the inner and outer helical blades having first and second twist slopes in the first and second sections respectively, and

the first twist slopes being less than the second twist slopes.

15. An engine as claimed in claim 14 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

16. An engine as claimed in claim 15 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

17. An engine as claimed in claim 15 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

18. An engine as claimed in claim 14 further comprising the outer body being rotatable about the outer axis and the inner body and being rotatable about the inner axis.

19. An engine as claimed in claim 18 further comprising the inner and outer bodies being geared together in a fixed gear ratio.

20. An engine as claimed in claim 19 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

21. An engine as claimed in claim 20 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

22. An engine as claimed in claim 20 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

23. An engine as claimed in claim 14 further comprising the outer body being rotatably fixed about the outer axis and the inner body being orbital about the outer axis.

24. An engine as claimed in claim 23 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

25. An engine as claimed in claim 24 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

26. An engine as claimed in claim 25 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

27. A gas turbine engine comprising:

a gas generator connected in work producing relationship to a power consuming device,

a core engine including in downstream serial flow relationship an axial flow positive displacement compressor, a combustor, and a high pressure turbine drivingly connected to the compressor by a high pressure shaft,

the compressor having an inlet axially spaced apart and upstream from an outlet,

a compressor assembly including an inner body disposed within an outer body and the inner and outer bodies extending from the inlet to the outlet,

the inner and outer bodies having offset inner and outer axes respectively,

at least one of the inner and outer bodies being rotatable about a corresponding one of the inner and outer axes,

the inner and outer bodies having intermeshed inner and outer helical blades wound about the inner and outer axes respectively,

the inner and outer helical blades extending radially outwardly and inwardly respectively,

the inner helical blades extending radially outwardly from an inner hub of the inner body,

the compressor assembly having first and second sections in serial downstream flow relationship extending between the inlet and the outlet,

the inner and outer helical blades having first and second twist slopes in the first and second sections respectively, and

the first twist slopes being less than the second twist slopes.

28. An engine as claimed in claim 27 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

29. An engine as claimed in claim 28 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

30. An engine as claimed in claim 28 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

31. An engine as claimed in claim 27 further comprising the outer body being rotatable about the outer axis and the inner body and being rotatable about the inner axis.

32. An engine as claimed in claim 31 further comprising the inner and outer bodies being geared together in a fixed gear ratio.

33. An engine as claimed in claim 32 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

34. An engine as claimed in claim 33 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

35. An engine as claimed in claim 33 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

36. An engine as claimed in claim 27 further comprising the outer body being rotatably fixed about the outer axis and the inner body being orbital about the outer axis.

37. An engine as claimed in claim 36 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

38. An engine as claimed in claim 37 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

39. An engine as claimed in claim 38 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

40. An aircraft gas turbine engine comprising:

a fan section and a core engine including a gas generator downstream of the fan section,

a low pressure turbine having at least one row of turbine rotor blades downstream of the gas generator,

the low pressure turbine drivingly attached to at least one row of circumferentially spaced apart fan rotor blades in the fan section by a low pressure shaft,

the core engine including in downstream serial flow relationship an axial flow positive displacement compressor, a combustor, and a high pressure turbine drivingly connected to the compressor by a high pressure shaft,

the compressor having an inlet axially spaced apart and upstream from an outlet,

a compressor assembly including an inner body disposed within an outer body and the inner and outer bodies extending from the inlet to the outlet,

the inner and outer bodies having offset inner and outer axes respectively,

at least one of the inner and outer bodies being rotatable about a corresponding one of the inner and outer axes,

the inner and outer bodies having intermeshed inner and outer helical blades wound about the inner and outer axes respectively,

the inner and outer helical blades extending radially outwardly and inwardly respectively,

the inner helical blades extending radially outwardly from an inner hub of the inner body,

the compressor assembly having first and second sections in serial downstream flow relationship extending between the inlet and the outlet,

the inner and outer helical blades having first and second twist slopes in the first and second sections respectively, and

the first twist slopes being less than the second twist slopes.

41. An engine as claimed in claim 40 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

42. An engine as claimed in claim 41 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

43. An engine as claimed in claim 41 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

44. An engine as claimed in claim 40 further comprising the inner and outer bodies being geared together in a fixed gear ratio.

45. An engine as claimed in claim 44 further comprising the outer body being rotatable about the outer axis and the inner body and being rotatable about the inner axis.

46. An engine as claimed in claim 45 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

47. An engine as claimed in claim 46 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

48. An engine as claimed in claim 44 further comprising the outer body being rotatably fixed about the outer axis and the inner body being orbital about the outer axis.

49. An engine as claimed in claim 48 further comprising the helical blades in the first section having a sufficient number of turns to trap charges of gas in the first section during the compressor's operation.

50. An engine as claimed in claim 49 further comprising the number of turns being sufficient to mechanically trap the charges of gas.

51. An engine as claimed in claim 49 further comprising the number of turns being sufficient to dynamically trap the charges of gas.

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