National Aeronautics and Space Administration

Overview of NASA Power Technologies for Space and Aero Applications

Presented to

IEEE Cleveland Power and Energy Society

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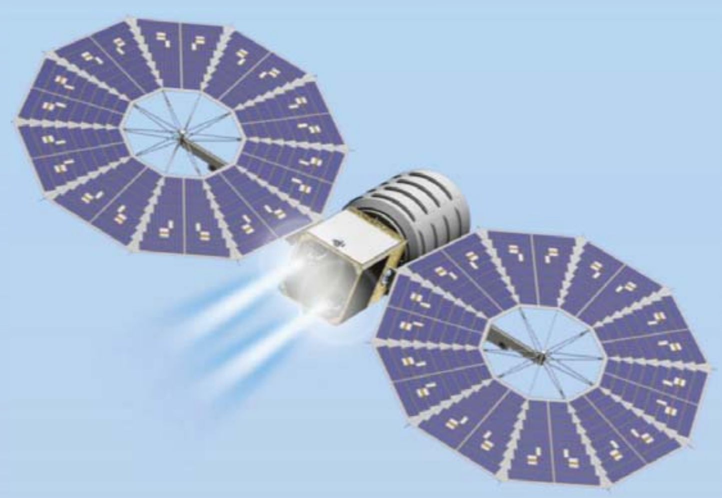
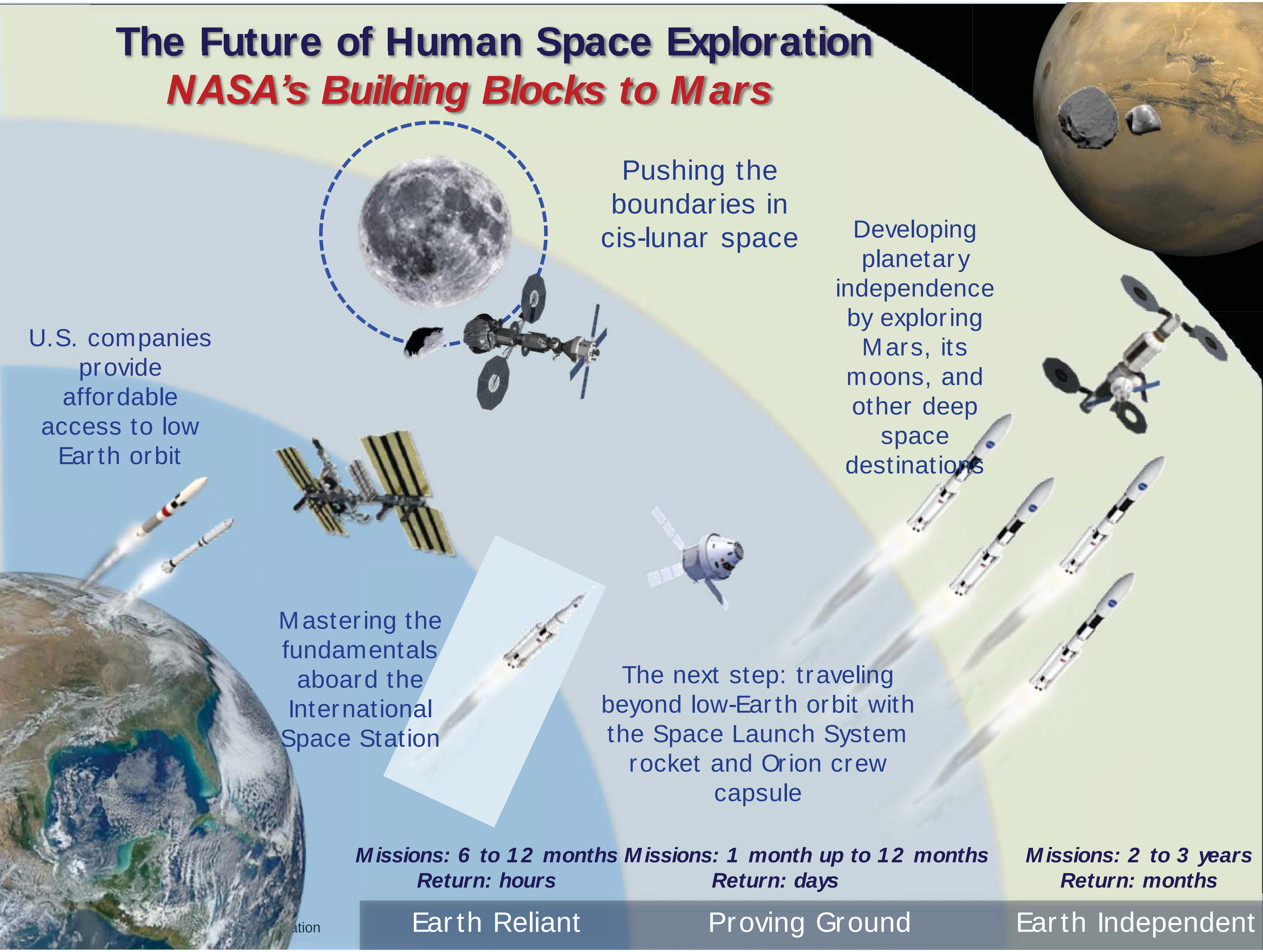
October 16, 2014



Topics

- Space Power Development Objectives and Roadmap
- Aircraft Power Development Objectives and Roadmap
- Component Technology Development

Space Power Development Objectives and Roadmap

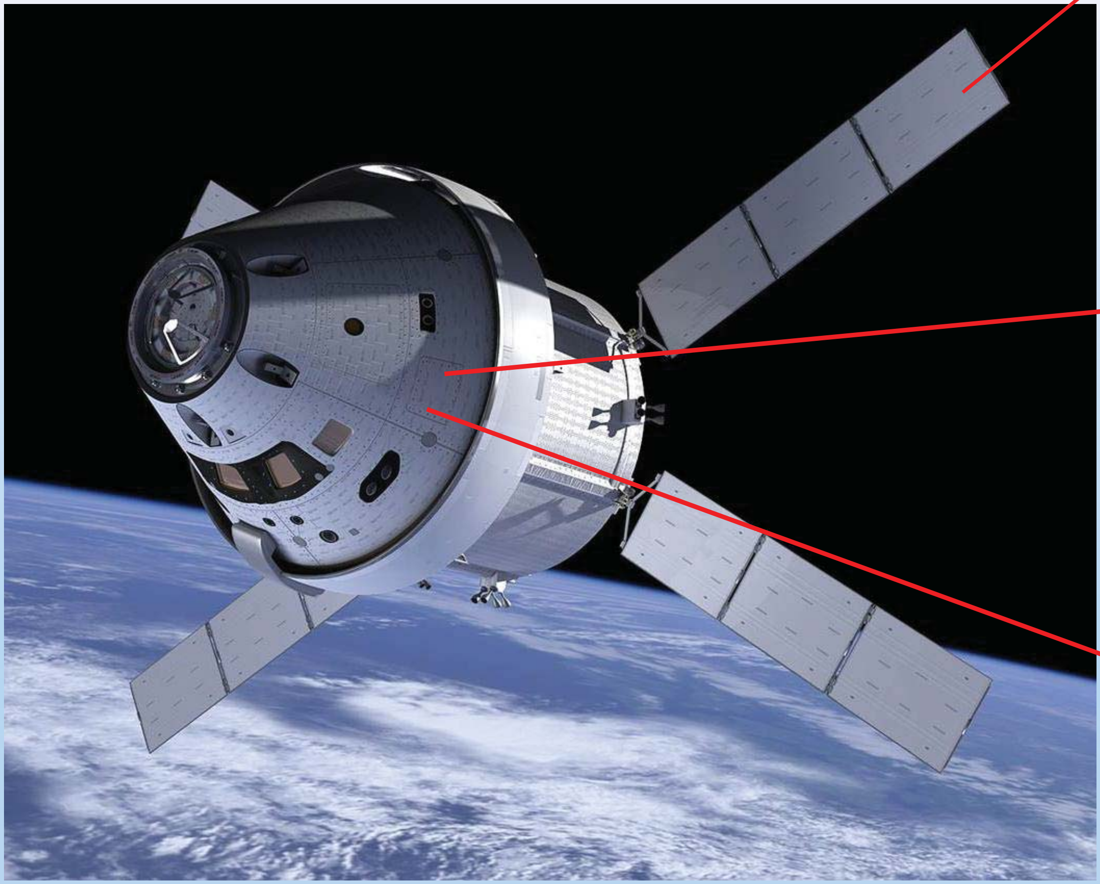

The Future of Human Space Exploration
NASA's Building Blocks to Mars
Pushing the boundaries in cis-lunar space
Developing planetary independence by exploring Mars, its moons, and other deep space destinations
U.S. companies provide affordable access to low Earth orbit
Mastering the fundamentals aboard the International Space Station
The next step: traveling beyond low-Earth orbit with the Space Launch System rocket and Orion crew capsule
Missions: 6 to 12 months
Return: hours
Missions: 1 month up to 12 months
Return: days
Missions: 2 to 3 years
Return: months
Earth Reliant
Proving Ground
Earth Independent

The Space Launch System (SLS)

- Designed to carry the Orion spacecraft, cargo, equipment and science experiments to Earth's orbit and destinations beyond.
- The SLS will have an initial lift capacity of 70 metric tons and will be evolvable to 130 metric tons.
- It will use a liquid hydrogen and liquid oxygen propulsion system, which will include the RS-25 from the Space Shuttle Program for the core stage and the J-2X engine for the upper stage.
- SLS will use solid rocket boosters for the initial development flights, follow-on boosters will be competed based on performance requirements and affordability considerations.

Initial 70 metric ton configuration

Orion MPCV Electrical Power System



Solar Array Wings

- 4 wings with 3 deployable panels
- Triple junction solar cells for high conversion efficiency
- Two axis articulation for sun tracking
- 11.1 kW total power for user loads and battery recharge

Battery Energy Storage

- 4 batteries of ≈ 30 A-hr each
- Li ion chemistry for high energy density
- High voltage for direct connection to power distribution
- Cell balancing for high charge/discharge cycle life

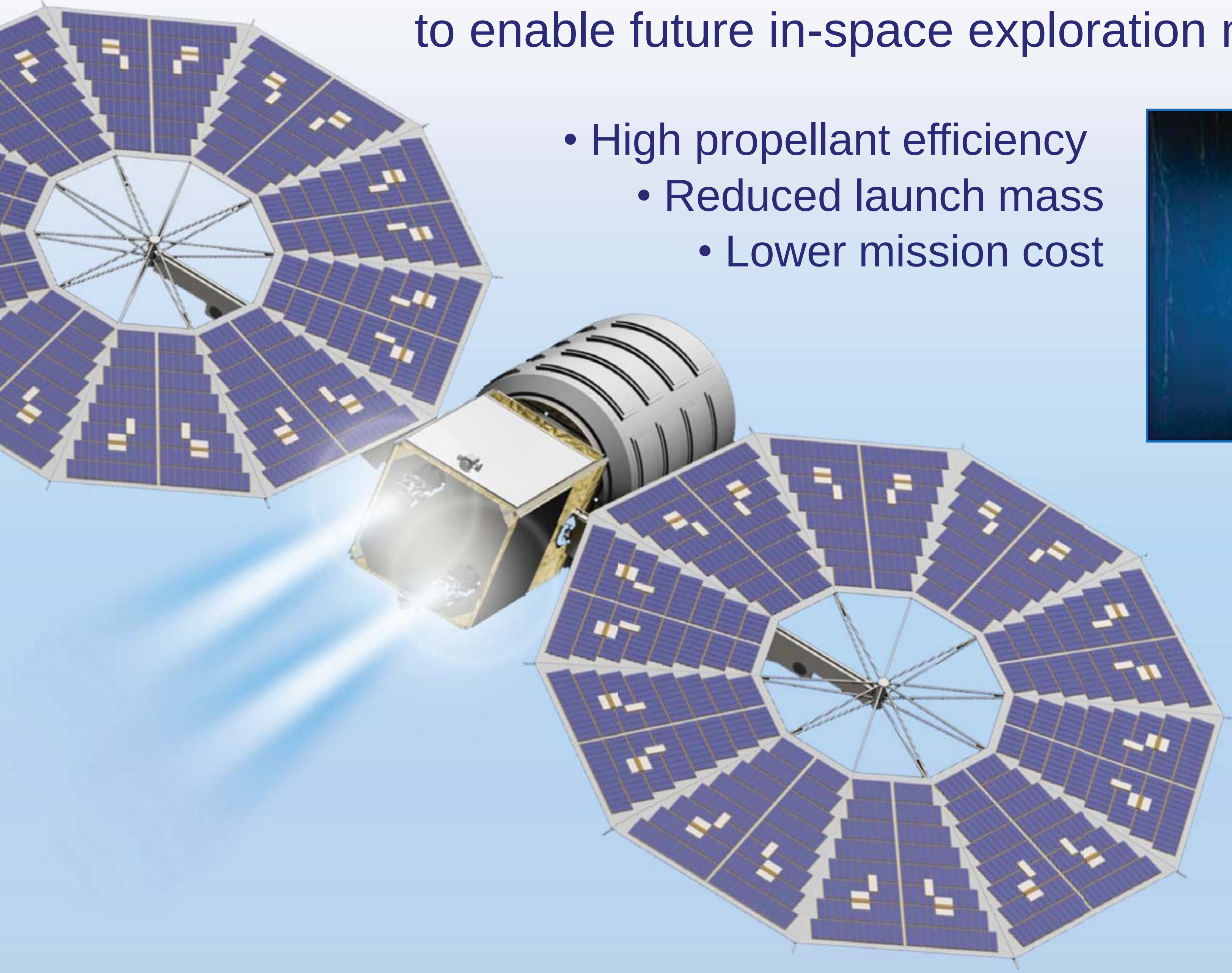
Power Distribution Equipment

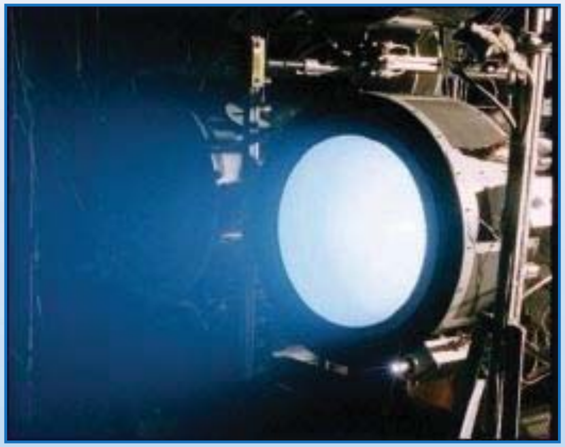
- 4 power distribution channels
- High voltage (120 VDC) distribution for reduced weight
- Current-limiting SiC switchgear for fault protection
- Transient protection for lightning strikes (on ground)

Solar Electric Propulsion (SEP)

NASA is developing high-performance SEP capability to enable future in-space exploration missions.

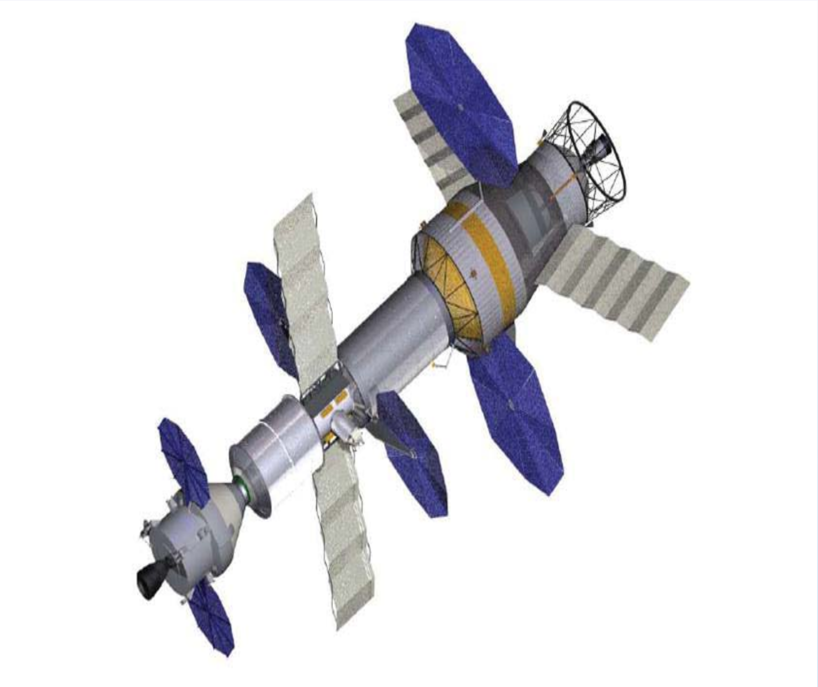
- High propellant efficiency
- Reduced launch mass
- Lower mission cost





Potential Deep Space Vehicle Power System Characteristics

- Power 10 kW average
- Two independent power channels with multi-level cross-strapping
- Solar array power
  - 24+ kW Multi-junction arrays
- Lithium Ion battery storage
  - 200+ amp*hrs
  - Sized for deep space or low lunar orbit operation
- Distribution
  - 120 V secondary (SAE AS 5698)
  - 2 kW power transfer between vehicles



Deep space vehicle concept

Aero Power Development Objectives and Roadmap

Aircraft Turboelectric Propulsion

Projected Timeframe for Achieving Technology Readiness Level (TRL) 6

Spinoff Technologies Benefit More/All Electric Architectures:

- High-power density electric motors replacing hydraulic actuation
- Electrical component and transmission system weight reduction

Power Level for Electrical Propulsion System

>10 MW

- Turboelectric and hybrid electric distributed propulsion 300 PAX

5 to 10 MW

- Hybrid electric 737–150 PAX
- Turboelectric 737–150 PAX

2 to 5 MW class

- Hybrid electric 100 PAX regional
- Turboelectric distributed propulsion 150 PAX

1 to 2 MW class

- Hybrid electric 50 PAX regional
- Turboelectric distributed propulsion 100 PAX regional

kW class

- All-electric and hybrid-electric general aviation

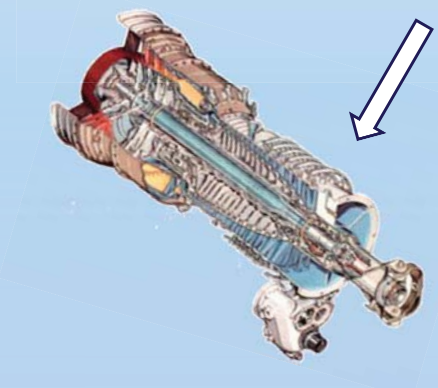
(Power level for single engine)

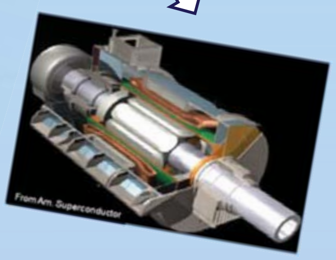
Today | 10 Year | 20 Year | 30 Year | 40 Year

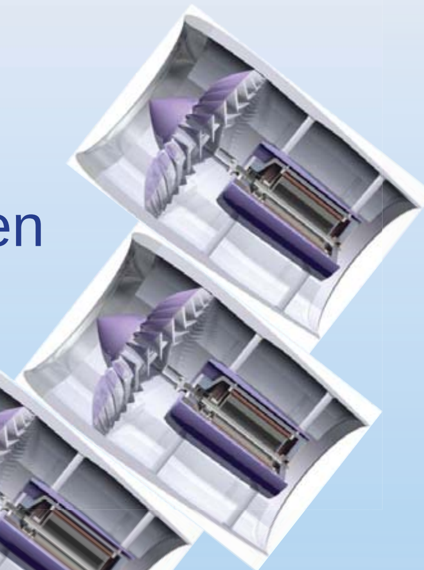
Aircraft Turboelectric Propulsion

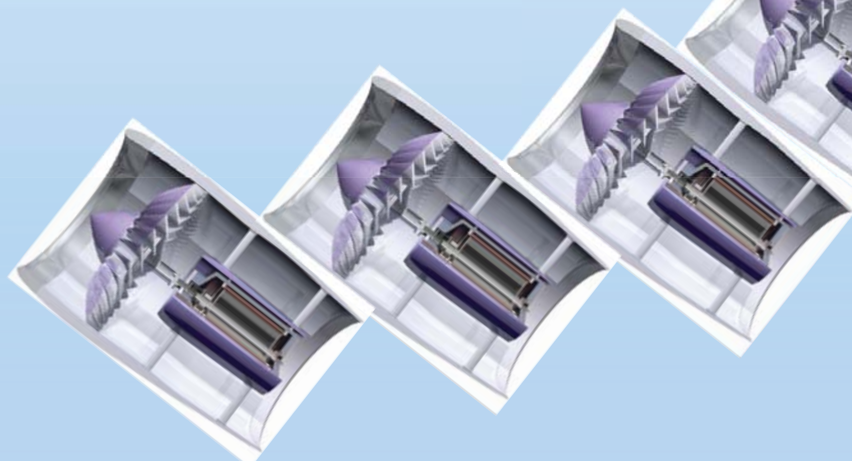
Wingtip mounted superconducting turbogenerators

Superconducting motor-driven fans in a continuous nacelle





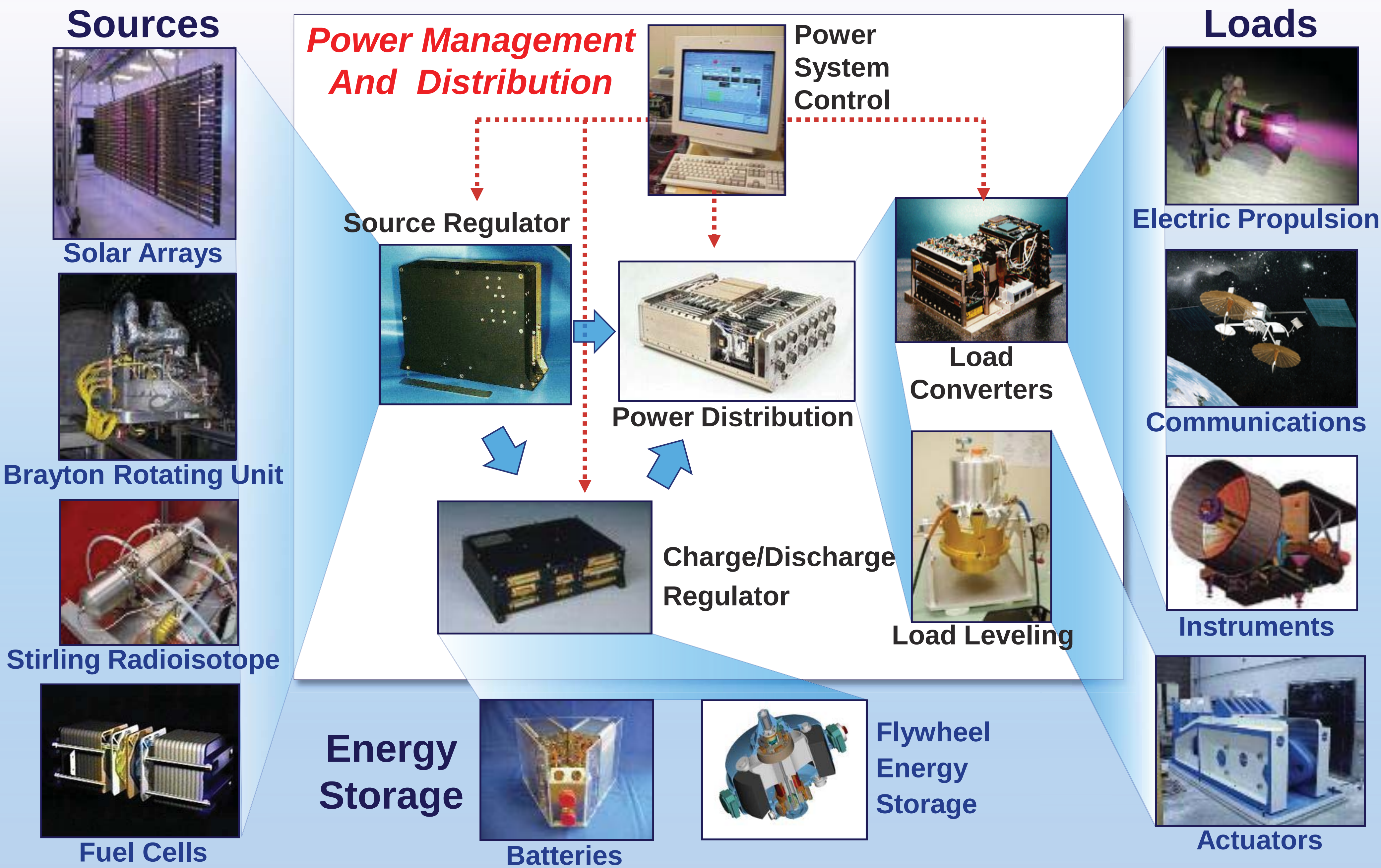




Power is distributed electrically from turbine-driven generators to motors that drive the propulsive fans.

Advanced Power Technologies Development Needs and Directions

Power System Taxonomy



Photovoltaic Arrays

Current State

- Solar Cell Efficiency approx. 30%
- 6 mil thick, non-flexible cells
- Relatively high cost with only limited automation

- Honey-comb panels @ 10-15 kW power levels
- Stowed volume limits power levels available

Drivers

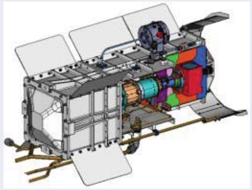
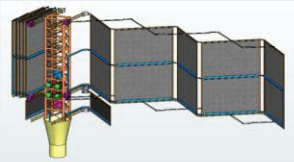
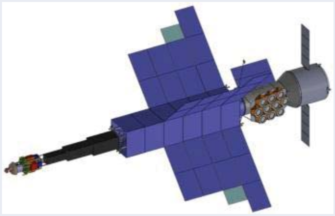
- High Power Scalability
- Higher efficiency
- Lower Cost
- Lower Mass
- Improved Radiation Resistance
- Survive Space Environments
- High bus voltage capability
- Increased Reliability
- Improved stowed volume and deployability
- High temperature/high intensity and low temperature/low intensity operation

Missions

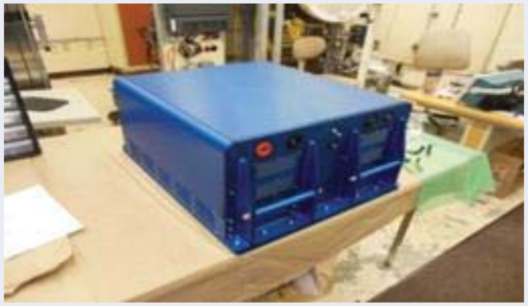
- Low cost, low mass blanket technology using automated manufacturing methods

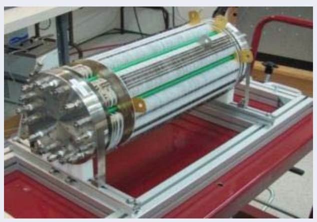
- Large multi-hundred kilowatt solar arrays w/ improved stowed volume and deployability.
- Arrays tailored for low intensity / low light operation

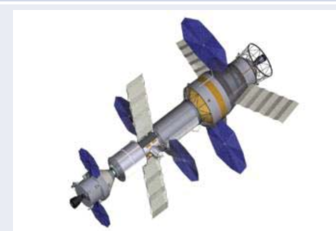
Nuclear Power Generation

Current State	Drivers	Future State
MMRTG • 110 W modules • Low efficiency 	• Long duration deep space missions • Greater distance from sun • Planet surface ops • Large power generations for nuclear electric propulsion • 100sW – MW needs	• Advanced Stirling Generation > 20% Conversion Efficiency  • Nuclear surface power  • Large fission for NEP 

Batteries

Current State	Drivers	Future State
• Rechargeable: $Ni\text{-}H_2$ (45Wh/kg, > 10 years); Li-Ion (100 Wh/kg, > 5 years life) • Primary: Ag-Zn (100 Wh/kg; 20 cycles); $Li\text{-}SO_2$ (200 Wh/kg; 5 years life) • Heavy, Bulky • Safety Concerns 	• Very high specific energy Rechargeable batteries to enable longer operation • Emphasis on safety • Longer cycle life • Extreme temperature environments	• "Beyond Li ion" Rechargeable Batteries: > 500 Wh/kg, 5 yrs • Rechargeable Li ion Long cycle life batteries:> 220 Wh/kg, 5 yrs • Primary: 1000 Wh/kg, > 20 yrs 

Current State	Drivers	Future State
Regenerative Fuel Cells		
• Power rating 2-10 kW • 35-50% Efficient • Life: 50 Cycles • Heavy, Bulky, Complex, Safety Concerns 	• Longer missions – days / weeks • High Efficiency • “Passive” management of fluids and gasses • High Power Rating and energy storage capability • Long Life, high reliability, safe • Operate with flexible fuels	• Power Rating: 10-30 kW >8 hrs. • Operable with reactants at > 2000 psi to reduce tank volume • Life: 10,000 hours • 70% Efficient, Reliable & Safe • Solid oxide fuel cells capable of CO_2 processing and oxygen production
Flywheels		
• Specific Energy 50Whr/kg 	• High power • Long life • High Energy Density • High Strength Fibers • Low Loss Bearings • Reliability • Mass	• Carbon fiber or Graphene specific power >200+ W-hr/kg. • Cycle life >150,000 cycles • Operating temperature • -150C to +150C

Current State	Drivers	Future State
Power Conversion and Distribution Systems		
• Power converters 94% efficient • Power Distribution: 170V and 120 V • Switchgear – Solid State, Electromechanical Relays	• Need for unique vehicle configurations • Extreme Space environments • Maximize efficiency, power density, safety, reliability • Minimize mass/volume, DDT&E costs, integration and operations cost	• Modular PMAD • Power Converter >97% • Voltage >300V • Novel Switching Devices • Superconductors • High radiation tolerance 
Intelligent Power Management Systems		
• Spacecraft power managed by ground controllers 	• Long term autonomous operations • Load and energy management under constrained capacity • Failure diagnostics and prognostics • Integration with Mission Manager	 Autonomous Vehicle Management with Ground Oversight

Electric Machines for Commercial Aircraft Propulsion

Current State	Drivers	Future State
• Commercial aircraft use turbofans or turbo props. Electric aircraft propulsion only implemented on small experimental planes. • Motors, generators, power distribution, and energy storage to heavy and inefficient to exceed performance of baseline system	• High Specific Power Electric Machines (>8HP/lb) • High Efficiency Electric Machines • High reliability/ redundancy • High Specific Energy batteries for some configurations	• 10-100MW aircraft propulsion electric system for regional, single isle and larger commercial aircraft. • Reduced aircraft fuel burn, NOx emissions, and noise • Electric propulsion power system able to meet or exceed current safety standards (engine out, redundancy, others).