Smart Book Charts

<These charts feed Smart Book and all charts are required. Please review questions on the next chart to ensure proper data control>

<Only update these charts if there have been changes since the STI review was completed>
**TPSM: Heatshield for Extreme Entry Environment Technology (HEEET)**

**Problem / Need Being Addressed**

TPS for extreme entry environment is solely dependent on the vanishing Carbon Phenolic and SMD in-situ science missions and higher speed human return missions are at risk.

**Project Description/Approach**

- HEEET maturation has begun with architecture prioritized based on testing and analysis performed during the FY13 formulation activity.
- The HEEET project will deliver a TRL 6 technology via:
  - Developing a mid-fidelity thermal response model
  - Will conduct thermal testing in the arc jet and laser testing at as high a heat flux as feasible to determine the range over which the material has robust and predictable response
  - A material property database will be developed and structural testing will be performed to validate thermal/structural models
  - Infrastructure to scale up weaving to NTE 3” thickness and at least 24” width will be put in place
  - A ~1m Engineering Test Unit will be fabricated and limited testing conducted which will validate workmanship and the structural modeling

**Quantitative Impact**

- Use of Carbon Phenolic results in entry g'loads of > 200 g's for Venus and OP missions
- Either or some combination of mass reduction by ~40%, compared to CP and g'load reduction by ~50% for extreme entry environment missions

**Status Quo**

- Venus in-situ science missions, high-speed sample return missions, Outer Planet probe missions and Human Missions beyond Lunar return speeds have no choice but to use a yet-to-be developed variant of heritage Carbon Phenolic, an unsustainable and missions limiting approach.

**New Insights**

- 3-D Woven TPS has been shown to be viable
- Large design choices in tailorable WTPS provide us options to down select to a single weave/resin architecture that will have broad capability to fill the TPS gap
- Heat shield technology can be matured to TRL 5/6 by ‘16/’17 to enable SMD missions in the next round of NF/Discovery Proposal Cycle

- Mature the prioritized HEEET architecture to TRL 6 by FY17
- Work with the mission proposing organizations to successfully infuse HEEET into Discovery and New Frontiers proposals
HEEET is developing an efficient and innovative Thermal Protection System that can protect science payloads during entry where the heating is 2 orders of magnitude higher than for Space Shuttle or Mars missions. The ablative thermal protection system has already been recommended for use by SMD in Discovery AO and will enable missions to Saturn, Venus and the Outer Planets.

**Integration with Other Projects/Programs and Partnerships:**
- CIF, SBIR/STTR Commercialization Readiness Program (CRP), SMD, STMD
- Discovery and New Frontier competed mission proposals
- ESA M4/M5 proposals
- Industry partnerships: Bally Ribbon Mills (BRM), Fiber Materials Inc. (FMI)

**Technology Infusion Plan:**
- Mature WTPS architecture to TRL 6 by FY17 so that it can be adopted by proposal teams for the next round of NF and Discovery proposals – Near term target is Discovery 2014
- Potential Customers (PC) includes SMD, STMD, HEOMD and ESA

**Key Personnel:**
**Program Element Manager:** Wade May  
**Project Manager:** Ethiraj Venkatapathy  
**HEEET Element Manager:** Don Ellerby  
**Lead Center:** ARC  
**Supporting Centers:** LaRC, JSC, KSC  
**NASA NPR:** 7120.8  
**Guided or Competed:** Guided  
**Type of Technology:** Push and Pull

**Key Facts:**
**GCD Theme:** Advanced Entry, Descent and Landing  
**Execution Status:** Year 2 of 4  
**Technology State Date:** 10/1/2013  
**Technology End Date:** 9/30/2017  
**Technology TRL Start:** 3  
**Technology TRL End:** 6  
**Technology Current TRL:** 3+  
**Technology Lifecycle Phase:** Implementation
TPSM: Conformal Ablative TPS (CA-TPS)

**Problem / Need Being Addressed**

- Limited number of certified TPS
- PICA tile on a rigid heatshields is limited by small size billet manufacturing and low strain-to-failure resulting in high tile count and gaps with filler design
- Honeycombed concepts (AVCOAT) require extensive touch-labor, large curing ovens, and complicated NDE

**Project Goal**

NASA requires TPS ablator advances (TA14.3.1) to significantly lower the areal mass of TPS concepts, demonstrate extreme environment capability, demonstrate high reliability, demonstrate improved manufacturing consistency and lower cost

**Project Description/Approach**

- TPS Materials Development to TRL 5/6
  - Leverage NASA Ames TPS expertise, ETDD, and Fundamental Aero- Hypersonics investments
  - Perform evaluation of felt substrates, impregnants, material processing and thermal/ablative property optimization
  - Small scale tests to show aerothermal survivability at flight-like heat flux, pressure and shear including ground test instrumentation
  - Measure thermal and structural properties
  - Development of mid-fidelity thermal response models for design of mission TPS
  - Partner with industry to manufacture materials (both felts and composites) at flight-like scale
  - Deliver Conformal 1-m size Manufacturing Demonstration Unit (MDU)

**New Insights**

- Impregnate felt-based substrates with various polymers resulting in materials with high strain-to-failure that conform to conventional rigid aeroshells
- Concepts taken to TRL 2-3 under ARMD FAP Hypersonics (FY11) and ESMD EDL TDP showed survivability to stagnation heat fluxes approaching 500 W/cm²

**Status Quo**

- Concept taken to TRL 2-3 under ARMD FAP Hypersonics (FY11) and ESMD EDL TDP showed survivability to stagnation heat fluxes approaching 500 W/cm²

**Quanitative Impact**

- Deliver TRL 5/6 TPS material solution ready for mission implementation
  - Scale up demonstration
  - Tech transfer with industry partnership

**Low cost, robust TPS solutions for mission applications:**

- Mars 2018 class TPS
- COTS (e.g., Dragon)
- ADEPT ViTAL rigid nose cap
- Backshell for Saturn, Venus and higher speed sample return
- High speed entry into Titan

**4-year development effort (FY12-15)**

- 1-m length Mid L/D MDU
- Tech transfer to Small Probe Manufacturer TVA

2015 GCD 3rd Quarter Review

3.4
CA-TPS is enabling small businesses and universities with low cost access to space by using off-the-shelf broad goods and transferring the processing technology for a light weight TPS. Use of CA-TPS materials will allow future missions to Mars, Venus and Outer Planets to save both mass and cost.

Integration with Other Projects/Programs and Partnerships:
- STMD, SMD – New Frontier proposals (TBD)
- Potential: IM LLC TRV, Orion

Technology Infusion Plan:
- Establish mission development partnerships for conformal entry system architectures
- Work with Conformal TPS vendors on scale up approaches and develop tech-transfer approach
- Mission Infusion (MI) with TVA for small probe flight test
- Potential Customers (PC) includes SMD, STMD, HEOMD and commercial
- SMD interest in incentivizing CA-TPS use in next (FY16-17) Discovery proposals

Key Personnel:
Program Element Manager: Wade May
Project Manager: Ethiraj Venkatapathy
CA-TPS Element Lead: Robin Beck
Lead Center: ARC
Supporting Centers: JPL, LaRC
NASA NPR: 7120.8
Guided or Competed: Guided
Type of Technology: Push

Key Facts:
GCD Theme: Advanced Entry, Descent and Landing
Execution Status: Year 4 of 4
Technology State Date: 10/1/2011
Technology End Date: 9/30/2015
Technology TRL Start: 2
Technology TRL End: 5/6
Technology Current TRL: 4
Technology Lifecycle Phase: Implementation
TPSM: 3D Woven Multifunctional Ablative TPS (3D MAT)

PROBLEM / NEED BEING ADDRESSED

Address current Orion MPCV Compression Pad design challenge – provide beyond LEO 3D woven compression pad material solution

PROJECT DESCRIPTION/APPROACH

• Phase I: ~5 months and addresses key risk areas of WTPS for Compression Pad (Phase II approval depends on Phase I success)
  • Requirements for Lunar return
  • Resin Transfer Molding up to 3" thick woven preform
  • Alternative resins to enhance manufacturing options
  • Weave architecture – composition tailored to meet both mechanical & thermal loads

• Phase II: FY’13 through FY’15 development based on FY’12 efforts
  • Mature to TRL ~4
  • Build, test & demonstrate performance meets requirement
  • Deliver a prototype compression pad material along with documents and test reports
  • Technology transfer to MPCV prime contractor

QUANTITATIVE IMPACT

• Early insertion of WTPS is feasible and highly advantageous for both GCD & MPCV
• With GCD investment early success = WTPS material being adopted by MPCV
• MPCV adoption of 3D MAT for the compression pad:
  • Results in near-term flight test data for a class of 3D Woven TPS
  • Compression Pad is both a structure and a TPS – the flight validated material will attract use by many programs including COTS, HEOMD and SMD

NEW INSIGHTS

• Orion EFT-1 compression pad design is complex and limited to LEO Return
• EFT-1 compression pad material was 2D laminate (similar to Apollo)
• Apollo capsule was smaller; loads on compression pads were different
• EFT-1 design required metallic insert to mitigate failure in 2D material
• Melting of metallic insert was predicted and observed
• EFT-1 pad cracking was predicted & observed
• Metallic insert design not suitable for beyond EFT-1 / LEO

• 3D Woven material can meet mechanical and thermal loads with minimal redesign
  • Leverage OCT funded WTPS development
  • Inter-layer tensile properties of 3D WTPS are much greater than 2D material
• 3D Woven will eliminate need for metallic insert, simplify the design, & reduce heating augmentation

STATUS QUO

• Deliver prototype 3D woven compression pad material at TRL ~4 by June of 2015 to meet summer 2015 CDR for MPCV EM-1 Lunar Return uncrewed flight test in FY’18
3D MAT Overview

3D MAT is developing a robust multi-functional material for use in spacecraft heatshields. The 3D-woven composite has pushed state of the art manufacturing to new levels yielding tailored materials that can be both structure and thermal protection system. The unique structural heat shield material is enabling future Orion missions to take humans far into the solar system.

<table>
<thead>
<tr>
<th>Integration with Other Projects/Programs and Partnerships:</th>
<th>Technology Infusion Plan:</th>
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<tbody>
<tr>
<td>• SBIR/STTR Commercialization Readiness Program (CRP), SMD, STMD</td>
<td>• Technology transfer to Lockheed Martin / Orion MPCV for further development and use as compression pads on EM-1 flight (2018) completed</td>
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<td>• Orion (mission infused for EM1)</td>
<td>• Jay Feldman is supporting Boeing’s assessment of 3DMAT for potential use on their Commercial Crew vehicle (CST-100) compression pad</td>
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<td>• Potential: Under evaluation by Boeing (CST 100)</td>
<td>• Boeing’s CCtCap funding covers this work</td>
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<tr>
<td>• Industry Partnerships: San Diego Composites (SDC), Bally Ribbon Mills (BRM)</td>
<td>• Potential Customers include STMD, SMD, HEOMD</td>
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**Key Personnel:**
- **Program Element Manager:** Wade May
- **Project Manager:** Ethiraj Venkatapathy
- **3D MAT Element Lead:** Jay Feldman
- **Lead Center:** ARC
- **Supporting Centers:** JSC, LaRC
- **NASA NPR:** 7120.8
- **Guided or Competed:** Guided
- **Type of Technology:** Pull

**Key Facts:**
- **GCD Theme:** Advanced Entry, Descent and Landing
- **Execution Status:** Year 3 of 3
- **Technology State Date:** 7/1/2012
- **Technology End Date:** 6/30/2015
- **Technology TRL Start:** 2
- **Technology TRL End:** 4
- **Technology Current TRL:** 4
- **Technology Lifecycle Phase:** Implementation
Robotic and human missions are limited by decades-old rigid, high ballistic coefficient aeroshells
- Geometry and packaging constraints within launch shroud or carrier spacecraft (secondary payload)
- Very high TPS certification and development costs and timeline
- Current EDL capability limits delivery of payloads to Mars surface at 1mT
- Probe missions to Venus experience 100s g’s deceleration loads

Large payload (> 1Mt) delivery at Mars and next generation science missions require deployable, low ballistic coefficient entry systems not achievable with conventional aeroshell technology.

ADEPT develops a low ballistic coefficient aeroshell system that consists of a 3-D woven carbon cloth skin stretched over a mechanically deployable ribbed structure (similar to an umbrella)
- Design provides predictable structural loads via skin/ribs/struts behaving as a rigid system
- ADEPT entry conditions are well within the current arc jet capabilities – testable and certifiable
- Demonstration of aerodynamic stability and carbon fabric robust performance under entry loads
- Characterization of 3D woven carbon fabric for thermal & structural behavior
  - Carbon fabric testing under relevant heating conditions while under sustained bi-directional tension loads
- Design, integrate and test a 1m diameter class test article with flight-like carbon fabric design and structure attachment. Perform systems level testing in both ground and sounding rocket flight environments.

Breakthrough capability to deliver entry system payloads to the most challenging mission destinations
What would you say to a Senator that you meet in the elevator?

• The ADEPT project is a new, advanced heatshield design to protect payloads and landers delivered to planetary bodies with atmospheres. ADEPT is a mechanically deployable heatshield, like an umbrella, that can open up at planet arrival to diameters 2-5 times greater than current rigid heatshields. This large size overcomes the current limitations of NASA’s heatshields to enable delivery of 10s of metric tons to Mars’ surface – essential for human exploration.

• Tweet: “ADEPT is a mechanically deployable low-ballistic coefficient entry system (think: umbrella) which overcomes current limitations of rigid heatshields.”

Integration with Other Projects/Programs and Partnerships

• CIF for Nano-ADEPT and wind tunnel testing
• UP Aerospace - Sounding Rocket (SR) flight test
• AVA avionics box baselined for SR
• SMD, STMD support
• Industry Partnerships: Bally Ribbon Mills (BRM), Thin Red Line Aerospace (TRLA)

Technology Infusion Plan:

• Near term: PC, Low Ballistic Coefficient EDL (1 m class), SMD, Potential secondary payload or TDO for Discovery 2017
• Long term: PC, Low Ballistic Coefficient EDL (10+ m class), HEOMD, Potential use on large payload delivery (10s of MT) to surface of Mars.
• Targeting SMD Discovery 2017 AO and supporting EDL Pathfinder

Key Personnel:

Project Manager: Paul Wercinski
Lead Center: Ames Research Center
Supporting Centers: LaRC, GSFC, AFRC, JPL
NASA NPR: 7120.8
Guided or Competed: Guided
Type of Technology: Push and Pull

Key Facts:
GCD Theme: AEDL
Execution Status: Year 1 of 2 (1m scale development)
Technology State Date: Oct 1, 2014
Technology End Date: Sep 30, 2016
Technology TRL Start: 3
Technology TRL End: 5-6
Technology Current TRL: 3
Technology Lifecycle Phase: Implementation
ADEPT Organization and Key Members

NASA ARC
- Project Management
- Systems Engineering
- Thermal Mechanical Design, Fab, Testing
- Aero/FSI Analysis and Testing

CFDRC, Huntsville, AL and TRLA, Chilliwack, British Columbia
Carbon Fabric Stitching (SBIR Øll-e)

Bally Ribbon Mills, Bally, PA
Woven Fabric Manufacturer

NASA GSFC
- Mission Design Support
- Mechanisms Design
- Systems Engineering Support

NASA AFRC
Focused Thermal Testing Development using Armstrong Combined Loads Facility

C-CAT, Fort Worth TX
Advanced Carbon-Carbon (ACC) Supplier

Southern Research Institute, Birmingham, AL
Material Property Testing

NASA LaRC
- Focused Thermal Testing using Radiant Test Facility
- Thermal/Structural Modeling