Outline

• The Orbital Debris (OD) Problem
• The NASA Orbital Debris Program Office (ODPO)
• DebriSat Project Overview
The Orbital Debris Problem

Growth of the Orbital Debris Populations

- Only objects in the US satellite catalog (~10 cm and larger) are shown
- Sizes of the dots are not to scale

Credit: NASA ODPO
How Much Junk is Currently Up There?

Baseball size or larger (≥10 cm): ~23,000
(tracked by the U.S. Air Force Space Surveillance Network)

Marble size or larger (≥1 cm): ~500,000

Dot or larger (≥1 mm): >100,000,000
(a grain of salt)

- Due to high impact speed in space (~10 km/sec in LEO), even sub-millimeter debris pose a realistic threat to human spaceflight and robotic missions
  - 10 km/sec = 36,000 km per hour (the speed of a bullet ~2,500 km per hour)
- Mission-ending threat is dominated by small (mm-to-cm sized) debris impacts
- Total mass: >7400 tons LEO-to-GEO (~2700 tons in LEO)

Credit: public domain images
Threat from Orbital Debris - Examples

- The gravity-gradient boom of an operational French satellite (CERISE) was cut in half by a tracked debris fragment in 1996
- The fully operational Iridium 33 was destroyed by a retired Russian satellite (Cosmos 2251) in 2009
- Near the end of the Space Shuttle Program, the Loss of Crew and Vehicle risks from MMOD impact damage were in the range of 1-in-250 to 1-in-300 per mission (OD to MM ~2:1 at ISS altitude)
- Satellite operators, including the International Space Station (ISS) Program, conduct collision avoidance maneuvers against the tracked objects on a regular basis
  - ISS has conducted 25 collision avoidance maneuvers and 5 “shelter-in-Soyuz” since 1999
- Impacts by small, untracked debris could be responsible for many satellite anomalies

Credit: NASA
The NASA Orbital Debris Program Office
The ODPO is the only organization in the U.S. Government conducting a full range of research on orbital debris

- This unique NASA capability was established at JSC in 1979 (D. Kessler, J. Loftus, B. Cour-Palais, etc.)
- ODPO’s roles and responsibilities are defined in NASA Procedural Requirements NPR 8715.6B

ODPO provides technical and policy level support to NASA HQ, OSTP, and other U.S. Government and commercial organizations

ODPO represents the U.S. Government in international fora (IADC, United Nations, etc.)

ODPO is recognized as a pioneer and leader in environment definition and modeling, and in mitigation policy development
ODPO’s End-to-End Mission Support

Mission Risk Assessments
OD risk assessment tools for NASA mission (ORDEM, ORSAT, DAS, etc.)

Measurements
Radar
Optical
In-situ
Laboratory

Modeling
Breakup
Engineering
Evolutionary
Reentry

Environment Management
Mitigation
Remediation
Policy
Mission Requirements

Coordination
U.S. Government
IADC, ISO
United Nations

Credits, left-to-right, top: Reprinted with permission Courtesy of MIT Lincoln Laboratory, Lexington, Massachusetts; Patrick Seitzer, University of Michigan; NASA (S125E007413). Left-to-right, bottom: NASA JPL, https://deepspace.jpl.nasa.gov/galleries/goldstone/#gallery; Ben Hanna, Arnold Engineering Development Complex/Air Force (2 images)
Space Situational Awareness Coverage in the U.S.

Particle Size (μm)

- 10 μm
- 100 μm
- 1 mm
- 1 cm
- 10 cm
- 1 m
- 10 m

Altitude (km)

- 100
- 1000
- 2000
- 36,000

Goldstone radars (>32.2°)
HUSIR (Haystack radar) (>30°)
USSTRATCOM Space Surveillance Network
HST-WFPC2 (580x610 km, 93-09)
STS (95-11), SDS (2018-2020)
Haystack Auxiliary (HAX) radar (>42.6°)
ES-MCAT (2018-)
MODEST (04-14)

Credit: NASA ODPO
Recent ODPO Highlights - Optical

• A NASA, Air Force, and Air Force Research Laboratory joint project
  – Ascension Island (7° 58' S, 14° 24' W)

• Two instruments at the facility
  – ES-MCAT: a double horse-shoe 1.3-m DFM telescope with a field-of-view of 41' × 41'
  – Benbrook: a 0.4-m telescope with a similar field-of-view

• Objectives for operations
  – Survey faint debris in GEO
  – Characterize low inclination objects in LEO
  – Provide rapid break-up response
  – Support other Space Situational Awareness applications

• Forward plan
  – Reach full autonomous operations for routine GEO debris survey by Sep 2018
Recent ODPO Highlights – In Situ Measurements

• The International Space Station (ISS) Program funded a DRAGONS technology demonstration mission launched in 2017
  – Objectives: (1) mature DRAGONS technologies and (2) characterize the sub-millimeter debris environment at the ISS altitude
  – To avoid confusion with the SpaceX Dragon, the mission was renamed Space Debris Sensor (SDS)

• NASA will continue to seek mission opportunities to deploy DRAGONS to characterize the millimeter-sized orbital debris populations at 600-1000 km altitude
Managing OD Risks for Space Assets

1. **Observe** events, objects/orbits, RCS, brightness, *etc.*

2. **Assess** population, debris size, mass, density, shape, velocity, lifetime, *etc.*

3. **Evaluate** impact consequences

4. **Mitigate** risks in a cost effective manner
Managing OD Risks for Space Assets

1. **Observe** events, objects/orbits, RCS, brightness, *etc.*

2. **Assess** population, debris size, mass, density, shape, velocity, lifetime, *etc.*

3. **Evaluate** impact consequences

   - Damage @ 7 km/sec
   - 4 mm Al sphere
   - 1.2 cm thick Al wall

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**Need lab-based satellite impact test data for**

- RCS-to-size conversion
- Optical magnitude-to-size conversion
- Fragment area-to-mass ratio distribution (to ~mm)
- Population characterization (including MLI and solar panel pieces)
- Fragment size and mass distributions (to ~mm)
- Fragment density and shape distributions (to ~mm)
DebriSat Project Overview

Credit: University of Florida
The purpose of the DebriSat project is to replicate a hyper-velocity fragmentation event using modern-day spacecraft materials and construction techniques to better improve the existing DoD and NASA breakup models.
The need for laboratory-based impact tests was recognized by DoD and NASA decades ago.

Key impact test series, Satellite Orbital Debris Characterization Impact Test (SOCIT) was conducted by the Department of Defense (DOD) and NASA at AEDC in 1992 to support the development of satellite breakup models.

Breakup models based on SOCIT have supported many applications over the years.
# Previous Laboratory Impact Test
## SOCIT vs DebriSat

<table>
<thead>
<tr>
<th></th>
<th>SOCIT</th>
<th>DebriSat</th>
</tr>
</thead>
<tbody>
<tr>
<td><strong>Target body</strong></td>
<td>U.S. Navy Transit 1960’s era satellite</td>
<td>Representative modern spacecraft in LEO</td>
</tr>
<tr>
<td><strong>dimensions</strong></td>
<td>46 cm (dia) × 30 cm (ht)</td>
<td>60 cm (dia) × 50 cm (ht)</td>
</tr>
<tr>
<td><strong>Target mass</strong></td>
<td>34.5 kg</td>
<td>56 kg</td>
</tr>
<tr>
<td><strong>MLI and solar</strong></td>
<td>No</td>
<td>Yes</td>
</tr>
<tr>
<td><strong>panel</strong></td>
<td></td>
<td></td>
</tr>
<tr>
<td><strong>Projectile material</strong></td>
<td>Al sphere</td>
<td>Hollow Al cylinder with attached nylon bore-rider</td>
</tr>
<tr>
<td><strong>Projectile</strong></td>
<td>4.7 cm diameter, 150 g</td>
<td>8.6 cm × 9 cm, 570 g</td>
</tr>
<tr>
<td><strong>dimension/mass</strong></td>
<td></td>
<td></td>
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<tr>
<td><strong>Impact speed</strong></td>
<td>6.1 km/sec</td>
<td>6.8 km/sec</td>
</tr>
<tr>
<td><strong>Impact Energy to Target Mass ratio (EMR)</strong></td>
<td>81 J/g (2.8 MJ total)</td>
<td>235 J/g (13.2 MJ total)</td>
</tr>
</tbody>
</table>

See Ausay, E., “A Comparison of the SOCIT and DebriSat Experiments”
DebriSat Project Motivation (2/2)

- As new materials and construction techniques are developed for modern satellites, there is a need for new laboratory-based tests to acquire data to improve the existing DoD and NASA breakup models and support space situational awareness (SSA) applications.

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NASA model predictions match well fragments from the breakups of old satellites, but are noticeably different from fragments generated by modern satellites, such as FY-1C (left) and Iridium (right).
DebriSat Project Goals

- Design and fabricate a 60-cm/56-kg class spacecraft ("DebriSat"), including MLI and solar panels, to be representative of modern payloads in LEO
- Conduct a hypervelocity laboratory impact test to simulate a catastrophic fragmentation event
- Collect, measure, and characterize all fragments down to 2 mm in size
- Use the data to improve space situational awareness applications and satellite breakup models for better orbital debris environment definition
DebriSat Project Team

- NASA Orbital Debris Program Office (ODPO)
  - Co-sponsor, project and technical oversight, data collection, data analyses, NASA model improvements

- AF Space and Missile Systems Center (SMC)
  - Co-sponsor, technical oversight

- The Aerospace Corporation (Aerospace)
  - DebriSat design support, DebrisLV design & fabrication, data collection, data analyses, DoD model improvements

- University of Florida (UF)
  - DebriSat design & fabrication, data collection, fragment processing and characterization

- AF Arnold Engineering Development Complex (AEDC)
  - Hypervelocity impact tests
# DebriSat Project Milestones
## Prep-Test

<table>
<thead>
<tr>
<th>Year</th>
<th>Milestone</th>
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<tbody>
<tr>
<td>2009-2011</td>
<td>• Project preparation</td>
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<tr>
<td></td>
<td>• First ODPO visit to SMC in May 2010</td>
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<tr>
<td></td>
<td>• Col. Swanson visited ODPO at NASA/JSC in August 2010</td>
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<tr>
<td>Sep 2011</td>
<td>• Project kickoff</td>
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<tr>
<td>Jan 2013</td>
<td>• Final DebriSat design</td>
</tr>
<tr>
<td>Jan 2014</td>
<td>• Complete DebriSat fabrication</td>
</tr>
<tr>
<td>April 2014</td>
<td>• Hypervelocity impacts at AEDC</td>
</tr>
</tbody>
</table>
DebrISat Project Milestones
Collection-Characterization-Application

2014-present
- Fragment processing and characterization

Sep 2016
- Delivery of 1st set of measurement data

2016-2017
- Initial data analyses for model improvements

2018-2019
- Fragment radar and optical measurements

Sep 2020
- Complete measurement data

2018-2020
- Debris model and SSA application updates
Hypervelocity Impact Facility at AEDC

- Range-G operates the largest two-stage light gas gun in the U.S.
- Standard diagnostic instruments include X-rays, high-speed Phantom cameras, and lasers
  - With additional Aerospace IR cameras, and witness plates
- Low-density polyurethane foam panels are installed inside target chamber to “soft catch” the fragments

Examples of the before (without target) and after impact views of the target chamber (10 ft × 20 ft).

Credit (all photos): Arnold Engineering Development Complex/Air Force
Pre-Test DebrisLV Shot

• To further increase the benefits of the project, Aerospace built a target resembling a launch vehicle upper stage ("DebrisLV") for the pre-test shot
  – DebrisLV: 17.1 kg, body dimensions ~ 88 cm (length) × 35 cm (diameter)
  – Test conditions were identical to the impact on DebriSat (facility setup, projectile, impact speed, soft catch, etc.)

• Pre-test shot was successfully conducted on 1 April 2014
  – Projectile impacted DebrisLV at 6.9 km/sec, completely fragmenting it
  – Fragments and soft catch foam panels/pieces were collected in boxes on 19 pallets for shipment
DebriSat Shot

- DebriSat shot was successfully conducted on 15 April 2014
  - Projectile impacted DebriSat at 6.8 km/sec, completely fragmenting it
  - Fragments and soft catch foam panels/pieces were collected in boxes on 21 pallets for shipment

A ~9 cm, 570-g projectile impacted DebriSat at 6.8 km/s

A ~9 cm, 598-g projectile impacted DebrisLV at 6.9 km/s

Credit (both videos): Arnold Engineering Development Complex/Air Force
Fragment Characterization Plan

• Measure fragments, including MLI and solar panel pieces, down to ~2 mm in size
  – Dimensions, mass, shape, density, composition

• Obtain 2D and 3D scanning data for fragments
  – Cross-sectional area, A/m, bulk density

• Conduct radar, photometric, and spectral measurements for selected fragments
  – Support improvements to radar and optical size-estimation models
DebriSat Project Summary

• Beginning in 2009, significant effort was put into design phase to provide a realistic representation of a modern LEO satellite

• Impact test conducted in April 2014; fragments recorded to date have far exceed the initial prediction of 85K

• Characterization process involves prepping soft-catch panels, extracting fragments, and acquiring measurements on all fragments ≥ 2 mm in size

• Subset of representative fragments will be analyzed in a radar facility and Optical Measurement Center at NASA\JSC to update the current radar-based Size Estimation Model and generate an optical SEM

• Goal is to use the data/analysis to update existing breakup models used by NASA and DoD and support future ORDEM development and damage assessments
Optical Laboratory Data Analyses: Optical Measurement Center (NASA/JSC)

- **Goal:** develop optical size estimation model eliminating need for assumptions in current calculations

\[
d = \frac{2 \cdot R}{\left[ \pi \cdot A_g \cdot \Psi(\alpha) \right]^{0.5} \cdot 10^{-5}} \left[ \frac{M_{\text{abs}}(\nu) + M_{\text{sun}}(\nu)}{5.0} \right]
\]

- **R:** 36,000 Km
- **$A_g$:** geometrical albedo = 0.175
- **$\Psi(\alpha)$**, Lambertian Phase Function

- **Acquire empirical-based bidirectional reflectance distribution function measurements to eliminate need to have aspect angle dependencies**

- **ASD field spectrometer:** high-resolution reflectance spectrometer.
- **SBIG CCD camera and attached filter wheel,** which uses Johnson/Bessell BVRI and SDSS $g'r'i'z'$
- **75 W Xenon arc lamp**
- **R17 robotic arm (5 DOF)**
- **Rotary arm with potentiometer**

Credit: NASA
Radar Facility Data Analyses

Historical:

- 1990s physical measurements acquired ($L_c$, area, and mass)
- Majority of objects investigated were aluminum or steel
- Radar measurements were taken at multiple angles in order to avoid undersampling the RCS aspect angle variations
- Data were taken at a wide range of frequencies from 2.4 to 18 GHz (S, C, X and Ku bands)

Plan Forward:

- Repeat the 1990s experiment with more modern materials, large frequency sweep, and collect RCS measurements in various aspect angles
- Similar to previous analysis, the physical parameters compared with RCS measurements to investigate the relationships for a broader subset of materials/shapes
- The DebriSat fragments will provide the opportunity to compare optical measurements against optical inferred sizes and radar cross sections