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JOHN F. KENNEDY SPACE CENTER
UNIVERSITY OF CENTRAL FLORIDA

SHOCK SPECTRA APPLICATIONS TO A CLASS OF MULTIPLE DEGREE-OF-FREEDOM STRUCTURES SYSTEM

Prepared By: Shoi Y. Hwang
Academic Rank: Professor
University and Department: South Carolina State College
Civil and Mechanical Engineering
NASA/KSC:
Division: Launch Equipment Test Lab
Branch: Mechanical Engineering
NASA Counterpart: Gary Lin
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ABSTRACT

The demand on safety performance of launching structure and equipment system from impulsive excitations necessitates a study which predicts the maximum response of the system as well as the maximum stresses in the system. A method to extract higher modes and frequencies for a class of multiple degree-of-freedom (MDOF) Structure system is proposed. And, along with the shock spectra derived from a linear oscillator model, a procedure to obtain upper bound solutions for the maximum displacement and the maximum stresses in the MDOF system is presented.
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1.1 INTRODUCTION

The prevention of structure and equipment from damage by impulsive excitations necessitates a study which will predict the maximum dynamic response of the system. Two kinds of impulsive excitations are considered in this study; a blast-pressure which acts directly on the structure or equipment, and a sudden acceleration of bases which support structure or equipment. The investigation can provide some useful information which is relevant to the KSC launching equipment shock design applications.

The purpose of this study is to develop a practical method which will efficiently extract higher modes and frequencies for a class of Multiple Degree-of-Freedom (MDOF) structures. When these higher modes and frequencies are used along with the shock spectra of a linear oscillator subjected to the same excitation, their contributions to the maximum stresses in the real structure could well be very significant.

2.1 ANALYSIS

At least for the purpose of estimate or in the initial design stage, a detailed dynamical analysis of a real structure system is rarely attempted. The usual practice is to choose an idealized mathematical model consisting of springs (or elastic elements), dampers, and lumped masses which closely perform in the same way as the real structure or equipment. Figure 1.1 shows how each real structure or equipment is represented by an idealized mathematical model. In this study, damping is excluded from the analysis, since only the maximum dynamical response of the system is of primary interest.

A class of structures considered in this study are beams and frames of various support conditions. These structures are the typical ones which support equipment or instruments, and in certain cases, represent the equipment itself. For the sake of simplicity and practicality, only up to three DOF structures are included in this study. Accordingly, the method is considered efficient when it is applied to these structures. In developing the method, with the exception of the first mode and frequency which require a few iterations, the solution extracts higher modes and frequencies directly from the frequency equation. The equations governing the motion of MDOF structures are written in terms of flexural modes, but they are equally applicable to the cases of torsional modes.
Table 1.1 shows the class of structures which are included in this study.

2.1.1 SINGLE DEGREE-OF-FREEDOM SYSTEM (SDOF). A brief description of the SDOF structure system is discussed first, because it can providemuch insights to the subsequent study of the MDOF structure system. The concept of SDOF model implies that a single coordinate is sufficient to describe the motion of a real structure. The equation of motion of an equivalent SDOF model is given by:

\[ m_e \ddot{y}(t) + \kappa_e y(t) = f_e(t) \]  

where \( m_e \), \( \kappa_e \) and \( f_e \) are parameters of the equivalent SDOF system, the values of which are evaluated on the basis of an assumed deflection shape of the real structure. Detailed expressions of these parameters are given in the Appendix A. The natural frequency, \( \omega_e \), of the equivalent SDOF system is simply

\[ \omega_e = \left( \frac{\kappa_e}{m_e} \right)^{\frac{1}{2}} \]  

whith \( \omega_e \) being known, the maximum dynamic magnification factor (DMF)\(_{\text{max}}\), which is defined as the ratio of the maximum dynamic deflection to the deflection which would have resulted from the static load application. It should be emphasized that the maximum dynamical response thus obtained for the equivalent SDOF system is identical to that in the real structure. The maximum dynamic stress is then given by:

\[ \sigma_{dy} = \sigma_{st} \left( \text{DMF} \right)_{\text{max}} \]  

where \( \sigma_{st} \) is the maximum static stress and \( \sigma_{dy} \) the maximum dynamic stress, both are induced by the same impulsive excitation. An example is given in Appendix C which illustrates the application of the SDOF concept.
2.1.2 MULTIPLE DEGREE-OF-FREEDOM SYSTEM (MDOF). If a real structure system has more than one possible mode of displacement, then more than one independent coordinate is needed to describe its response. The structure system must now be represented by a MDOF model. In a MDOF system; determining frequencies and modes become exceedingly cumbersome, because one must deal with a complete set of equations of motion, one equation for each degree of freedom. The complexity, however, can be reduced greatly by using the modal analysis concept in which the response in the normal modes are determined separately, and then superimposed to provide the total response.

2.1.2.1 Fundamental Mode and Frequency. In most practical problems, usually a few of the lower modes are of interest. Therefore, the Rayleigh method is convenient to use, especially in finding the fundamental frequency. By this method, the natural frequency of the fundamental mode (first mode) can be obtained with considerable accuracy and yet with relative ease. Although the mode shape obtained is less accurate, that can be improved with few iterations. In Rayleigh, the equation used to obtain the natural frequency of fundamental mode is given by:

$$
\omega^2 = \frac{\sum_{r=1}^{j} F_r \phi_r^2}{\sum_{r=1}^{i} M_r \phi_r^2}
$$

(4)

where

$$
\phi_r = \text{displacement coordinate of rth mass}
$$

$$
F_r = \text{inertia force of rth mass}
$$

$$
A = \text{a constant}
$$

$$
M_r = \text{rth mass}
$$

$$
\omega = \text{natural frequency of fundamental mode}
$$

In many practical problems, a reasonable solution of fundamental frequency is
often obtained by assuming the static deflection curve as the mode shape, and the dynamic deflection curve is used in subsequent iterations if desirable.

2.1.2.2 High Modes and Frequencies. After the fundamental mode and frequency have been determined from the preceding section, the next higher modes and frequencies of a three DOF system are then directly extracted from the following frequency equation.

\[ g_n^6 + C_4 g_n^4 + C_2 g_n^2 + C_0 = 0 \]  

(5)

where \( g_n \) relates to the frequency of higher mode and \( C_4, C_2, C_0 \) are constants which relate to masses and flexibility coefficients of the particular structure concerned. Detailed descriptions of variables and constants in Eq.(5) are given in the Appendix B.

2.1.2.3 Upper Bound Maximum Displacement of Masses. The upper bound of the maximum displacement, \( y_{r, \text{max}} \), of \( r \)th mass due to all modes is given by:

\[ y_{r, \text{max}} = \frac{\sum A_{nst} \phi_{rn} (DMF)_{\text{max}, n}}{N} \]  

(6)

where

- \( A_{nst} \) = modal static desplacement
- \( \phi_{rn} \) = displacement coordinate of \( r \)th mass for \( n \)th mode
- \( N \) = number of modes
- \( (DMF)_{\text{max}} \) = maximum dynamic magnification factor

The \( y_{r, \text{max}} \) computed in this manner is a rather conservative estimate of the maximum displacement.

2.1.2.4 Maximum Dynamic Load. In order to find the maximum stress in the structure, the maximum relative displacement between two adjacent masses must be determined first, and which is given by the following equation.
where

\[ \Delta r,_{\text{max}} = \sum_{r=1}^{N} A_{\text{nst}} (\phi_{rn} - \phi_{(r-1)n}) (\text{DMF})_{\text{max},n} \]  

(7)

The maximum dynamic force, \( F_r \), which induces maximum dynamic stress in the real structure, is then given by:

\[ F_r = k_r \Delta r,_{\text{max}} \]

Where \( k_r \) is the spring constant between the \( r \)th mass and the \((r-1)\)th mass.

Now by replacing the static force in the real structure with one, the maximum dynamic force, in the same structure, the computation of maximum dynamic stress can be proceeded in the same way as in the static case.

3.1 APPLICATION

Eight beams and frames of various support conditions are chosen in this study. They are grouped into three categories below and also are shown in Table 1.1

a. SDOF System

Simply Supported Beam

b. Two DOF System

Simply Supported Beam
Fixed Ends Beam
Overhanging Beam
Rigid Body on Flexible Supports

c. Three DOF System
Simply Supported Beam
Simply Supported and Fixed End Beam
Shear-Building Frame

Frequencies and modes are obtained for all eight cases. These cases, one in each category, are chosen in stresses computation. No attempts are made to include all possible cases, the method, however, is general enough in application that a modification on flexibility coefficients is all that required. A computer program is written for each case except the SDOF one. In the program, the flexibility coefficients are derived from the static deflection curve. Examples in Appendix D show modes and frequencies for all eight cases and stresses computation for three cases. Although programs and examples are written in flexural mode, they are equally valid in torsional mode. To obtain results in the torsional mode, simply substitute the mass, modulus of elasticity, and the area moment of inertia in the flexural mode with the mass moment of inertia, modulus of rigidity, and polar moment of inertia in the torsional mode, respectively.

4.1 RESULTS AND DISCUSSIONS

Modes and frequencies are obtained for all seven cases in the MDOF system. And stresses are computed for three cases, one in each category. The results are verified from some known sources. The method is general and yet efficient to extract higher modes and frequencies in a MDOF system. In application, flexibility coefficients must be obtained first for each structure concerned. The advantage of this proposed method is that modes and frequencies obtained in the MDOF system and the shock spectra developed in the linear oscillator can each serve as an independent module. Any change in one does not affect the other. But both must act together to obtain the maximum displacements and stresses in the MDOF structure. For illustrative purpose, some sample outputs of dynamical responses and of shock spectra for a linear oscillator are given in Figures 5.1 through 5.6.

5.1 SUMMARY OF RESULTS

Results of modes, frequencies, and stresses for the MDOF systems are summarized in Table 5.1. Verifications are made from several known sources.
6.1 FUTURE RESEARCH

Many more cases can be included in the future study. Tables and charts in each case can be generated for quick references in shock design applications. If enough cases are developed, most likely, one can model a real structure analogue to one of the cases.
REFERENCES


**APPENDIX A**

**Governing Equation of the Equivalent SDOF System**

\[ m_e \ddot{y}(t) + K_e y(t) = F_e(t) \]

where

\[ m_e = \text{Equivalent mass} = \int_0^L m(x) \left[ \phi(x) \right]^2 dx + \frac{1}{2} \sum_{i=1}^j M_i \left[ \phi(r) \right]^2 \]

\[ K_e = \text{Equivalent spring constant} = \int_0^L E I(x) \left[ \phi''(x) \right]^2 dx \]

\[ F_e(t) = \text{Equivalent force} = \int_0^L p_i(x) f(t) \phi(x) dx + \frac{1}{2} \sum_{i=1}^j F_i f(t) \phi(r) \]

\[ \phi(x) = \text{Assumed mode shape curve of the real structure (normalized deflection curve)} \]

\[ \phi(r) = \text{mode coordinate at rth mass} \]

\[ F_i f(t) \]

\[ p_i(x) f(t) \]

\[ m(x) \]

\[ m_e \]

\[ F_e(t) \]

\[ K_e \]

\[ L, E, I \]
APPENDIX B

Governing Equations of MDOF System

I. TWO DOF SYSTEM

\[ g_n^4 + c_2 g_n^2 + c_0 = 0 \]  \hspace{1cm} (1)

\[ a_{ij} = \text{flexibility coefficient} \]

\[ \psi_1 = \frac{a_{12}}{a_{11}} \hspace{0.5cm} \psi_2 = \frac{a_{22}}{a_{11}} \hspace{0.5cm} a_{21} = a_{12} \]

\[ c_4 = -\frac{1 + \psi_2 m_2}{m_2 (\psi_2 - \psi_1)} \hspace{0.5cm} c_0 = \frac{1}{m_2 (\psi_2 - \psi_1)} \]

\[ m_2 = \frac{M_2}{M_1} \hspace{0.5cm} A = \psi_2 - \psi_1 \]

\[ M_1 = \text{mass 1} \hspace{0.5cm} M_2 = \text{mass 2} \]

\[ \omega_1 = \text{reference frequency} = \left( \frac{1}{a_{11} M_1} \right)^{\frac{1}{2}} \]

\[ g_{n_2} = \left( \frac{1}{A m_2} \right) \frac{1}{g_R} \]

\[ g_R = \frac{\omega_R}{\omega_1} \]

\[ \omega_R = \text{fundamental frequency from Rayleigh method} \]

Second mode displacement coord. ratio

\[ \frac{\phi_2}{\phi_1} = \frac{1 - g_{n_2}}{\psi_1 m_2 g_{n_2}} \]
APPENDIX B (CONTINUED)

II. THREE DOF SYSTEM

Governing equations of three DOF System

\[ g_n^6 + C_4 g_n^4 + C_2 g_n^2 + C_0 = 0 \quad (1) \]

\( a_{ij} = \text{flexibility coefficient} \)

\( \psi_1 = \frac{a_{12}}{a_{11}}, \quad \psi_2 = \frac{a_{22}}{a_{11}}, \quad \psi_3 = \frac{a_{33}}{a_{11}} \)

\( \psi_4 = \frac{a_{31}}{a_{11}}, \quad \psi_5 = \frac{a_{32}}{a_{11}} \)

\( a_{12} = a_{21}, \quad a_{13} = a_{31}, \quad a_{23} = a_{32} \)

\( M_1, M_2, M_3 = \text{masses} \)

\( m_2 = \frac{M_2}{M_1}, \quad m_3 = \frac{M_3}{M_1} \)

\( \psi_{ij} = \psi_i \psi_j, \quad \psi_{ijk} = \psi_i \psi_j \psi_k \)

\( A_1 = (\psi_{34} + 2 \psi_{135}) - (\psi_2 \psi_5 + \psi_1 \psi_4 + \psi_3^2) \)

\( B_1 = \psi_4 - \psi_5^2, \quad D_1 = \psi_{35} - \psi_{14} \)

\( B_2 = \psi_2 - \psi_1^2, \quad D_2 = \psi_{13} - \psi_{25} \)

\( B_3 = \psi_{24} - \psi_3^2, \quad D_3 = \psi_{15} - \psi_3 \)
APPENDIX B (CONTINUED)

\[ C_4 = -\frac{m_3 b_1 + m_2 b_2 + m_2 m_3 b_3}{m_2 m_3 A_1} \]

\[ C_2 = \frac{m_2 \psi_2 + m_3 \psi_4 + 1}{m_2 m_3 A_1} \]

\[ C_0 = -\frac{1}{m_2 m_3 A_1} \]

\[ w_1 = \text{reference frequency} = \left(\frac{1}{a_{11} M_1}\right)^{\frac{1}{2}} \]

\[ b = C_4 + g R^2, \quad c = -C_0 \frac{1}{g R^2} \]

\[ g_R = \frac{w_R}{w_1} \]

\[ w_R = \text{Fundamental Frequency from Rayleigh method} \]

\[ g_{n2} = \frac{-b - (b^2 - 4c)^{\frac{1}{2}}}{2}, \quad w_{n2} = w_1 g_{n2} \]

\[ g_{n3} = \frac{-b + (b^2 - 4c)^{\frac{1}{2}}}{2}, \quad w_{n3} = w_1 g_{n3} \]

\[ \left(\frac{\phi_2}{\phi_{1n}}\right)_{ni} = \frac{\psi_3 + D_3 g_{ni}}{\psi_5 + m_2 D_2 g_{ni}} \quad i = 2, \text{ second mode} \]

\[ \left(\frac{\phi_3}{\phi_{1n}}\right)_{ni} = \frac{\psi_3 + D_3 g_{ni}}{\psi_1 + m_3 D_1 g_{ni}} \quad i = 3, \text{ third mode} \]
APPENDIX C

5DOF system Example

Fixed Ends Beam

Equivalent Force $F_e = F_c \phi \left( \frac{L}{2} \right) = F_c$

Equivalent mass $m_e = \int_0^L m(\phi(x))^2 dx + M[\phi(\frac{L}{2})]^2$

Equivalent stiffness $K_e = \int_0^L [\phi''(x)]^2 dx$

$\phi(x) = \frac{4x^2(3L-4x)}{L^3}, \quad \phi(\frac{L}{2}) = 1, \quad M = \frac{25.6}{g}$

$m_e = 0.077 \frac{K-sec^2}{in}, \quad K_e = 258 \frac{k}{in}$

Natural frequency $\omega = (\frac{K_e}{m_e})^{1/2} = 58 \text{ rps}, \quad T = 0.109 \text{ sec}$

$t_d = 0.735$, From shock spectra, $(\text{DFD})_{max} = 2$

$F_{p, \text{max}} = (\text{DFD})_{max} F = 100K, \quad \sigma_{\text{max}} = \frac{F_{p, \text{max}} L}{6B} = 25.7 \text{ ksi}$

Impulsive Force $t_d = 0.08 \text{ sec}$ $F = 50K$
**APPENDIX D**

**Two DOF System Example**

![Diagram of a simply supported beam](image)

**Simply Supported Beam**

Shock Spectra
\[ F_1 = 1^k, \ F_2 = 0.8^k, \ t_d = 0.02 \text{sec} \]

<table>
<thead>
<tr>
<th>Mode</th>
<th>( \omega )</th>
<th>( \phi_1 )</th>
<th>( \phi_2 )</th>
<th>( \frac{t_d}{T} )</th>
<th>( \text{(OMF)}_{\text{max}} )</th>
<th>( M_T )</th>
<th>( F_T )</th>
<th>( \text{Anst} )</th>
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<td>1</td>
<td>1.07</td>
<td>1.51</td>
<td>1.7</td>
<td>0.0447</td>
<td>0.1295</td>
<td>0.039</td>
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<tr>
<td>2</td>
<td>1619</td>
<td>1</td>
<td>-0.465</td>
<td>5.16</td>
<td>1.9</td>
<td>0.0447</td>
<td>0.1295</td>
<td>0.0026</td>
</tr>
</tbody>
</table>

Unit: \( \omega \) - rps, \( M_T = \frac{k - s^2}{in} \), \( F_T = k \), \( \text{Anst} \) - in

\[
A_{\text{Anst}} = \sum_{n=1}^{\frac{3}{2}} \frac{F_T \phi_{\text{in}}}{\omega^2 M_T d_n^2}
\]

Total displacement of mass for all modes:

Mass 1: \( D_{m1} = \sum_{n=1}^{\frac{3}{2}} A_{\text{Anst}} \phi_{\text{in}} (\text{OMF})_{\text{max}}, m = 0.071 \text{in} \)

Mass 2: \( D_{m2} = \sum_{n=1}^{\frac{3}{2}} A_{\text{Anst}} \phi_{\text{in}} (\text{OMF})_{\text{max}}, m = 0.0685 \text{in} \)

Maximum Dynamic Force:
\[
F_{d1} = \frac{a_{12} D_{m1} - a_{12} D_{m2}}{a_{11} a_{22} - a_{12}^2}, \quad F_{d2} = \frac{a_{11} D_{m2} - a_{12} D_{m1}}{a_{11} a_{22} - a_{12}^2}
\]

Maximum Dynamic Stress: \( \sigma_{\text{max}} = \frac{MC}{I} = 1.61 \text{ ksi} \)
### Three DOF System Example

#### Three-Story Building Frame

<table>
<thead>
<tr>
<th>Mode</th>
<th>From proposed Method</th>
<th>From Shock Spectra</th>
<th>From Modal Analysis</th>
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<tr>
<td></td>
<td>( \omega )</td>
<td>( \phi_1 )</td>
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<tr>
<td>1</td>
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<td>1</td>
<td>1.471</td>
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<td>2</td>
<td>24</td>
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<tr>
<td>3</td>
<td>35</td>
<td>1</td>
<td>-2.22</td>
</tr>
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</table>

Units: \( \omega \) - rps, \( M_r \) - \( \frac{K \cdot s^2}{in} \), \( F_r \) - \( K \), \( A_{nst} \) - in
\textbf{APPENDIX D}

\[ A_{\text{nst}} = \frac{\sum_{r=1}^{3} F_r \phi_{r \text{n}}}{W_n \sum_{r=1}^{3} M_r \phi_{r \text{n}}^2} \]

The maximum roof displacement for all modes:

\[ y_{3, \text{max}} = \frac{3}{11} A_{\text{nst}} \phi_{3 \text{n}} (D M F)_{\text{max}, n} = 1.13 \text{ in} \]

The maximum relative displacement between the roof and the second floor for all modes

\[ A_{3, \text{max}} = \frac{3}{11} A_{\text{nst}} [\phi_{3 \text{n}} - \phi_{2 \text{n}}] (D M F)_{\text{max}, n} = 0.107 \text{ in} \]

The maximum moment at column end in the roof is

\[ M_{\text{max}} = \frac{6 E I A_{3, \text{max}}}{L_3^2} = 142.5 \text{ k-in} \]

The maximum dynamic stress

\[ \sigma_{\text{max}} = \frac{M_{\text{max}}}{S} = 3.54 \text{ ksi} \]
Simplified Model Showing Human Head dynamical response due to sudden base acceleration (Jet seat Ejection)

A more complex model representing the dynamical system of a human body

**Figure 1.1 REAL SYSTEM AND MATHEMATICAL MODEL**
Simplified model showing an instrument package inside a rocket nose subjected to sudden lift-off.

Simplified model representing a two-story building due to impulsive vertical Foundation Acceleration.

Simplified model showing a multistory building subjected to impulsive horizontal Loads.

**Figure 1.1 (Continue)**
Simplified model representing a
overhead crane lifting a heavy
object

Simplified model representing
a bearing support for a
rotating shaft

Simplified model representing a
turbine-generator foundation system

FIGURE 1.1 (CONTINUE)
Simplified Model representing propellers and shaft of a ship

Fig. 1-6

Simplified model representing an Automobile

Model of missile represented by a simplified model of lumped masses and elastic elements with bending stiffness

FIGURE 1.1 (CONTINUE)
FIG. 5.1 DYNAMIC RESPONSE
FIG. 5-2 DYNAMIC RESPONSE
FIG. 5.3 DYNAMIC RESPONSE
FIG. 5.4 SHOCK SPECTRA

ORIGINAL PAGE IS OF POOR QUALITY
FIG. 5.5 SHOCK SPECTRA
FIG. 5.6 SHOCK SPECTRA
TABLE 1.1
A CLASS OF TWO DOF STRUCTURE SYSTEM
BEAM AND FRAME

FORTRAN FILE
NS2.FOR
RF2.FOR
MOH2.FOR
MSP2.FOR
MS3.FOR
MSF3.FOR

THREE DOF

TWO DOF

SINGLY OCCUPIED
TABLE 5.1

TWO DOF SIMPLY SUPPORTED BEAM

INPUT DATA

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<th>load 1</th>
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<td>inch</td>
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<td>in**4</td>
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OUTPUT DATA

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<tr>
<td>2</td>
<td>1619.361</td>
<td>1.000000</td>
<td>-0.4650053</td>
</tr>
</tbody>
</table>

TWO DOF FIXED ENDS BEAM

INPUT DATA

<table>
<thead>
<tr>
<th>load 1</th>
<th>load 2</th>
<th>span</th>
<th>load 1</th>
<th>load 2</th>
<th>modulus</th>
<th>area</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>lb/in**2</td>
<td>in**4</td>
</tr>
<tr>
<td>25</td>
<td>50</td>
<td>35</td>
<td>25</td>
<td>10</td>
<td>1.E6</td>
<td>24</td>
</tr>
</tbody>
</table>

OUTPUT DATA

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>1121.801</td>
<td>1.000000</td>
<td>1.202247</td>
</tr>
<tr>
<td>2</td>
<td>2336.245</td>
<td>1.000000</td>
<td>-0.3883002</td>
</tr>
</tbody>
</table>
TABLE 5.1 (Continue)

TWO DOF OVERHANGING BEAM

INPUT DATA

<table>
<thead>
<tr>
<th>load 1</th>
<th>load 2</th>
<th>span length</th>
<th>load 1 from left support</th>
<th>load 1 from right support</th>
<th>load 2 from right support</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
</tr>
<tr>
<td>30</td>
<td>7.5</td>
<td>24</td>
<td>12</td>
<td>12</td>
<td>12</td>
</tr>
</tbody>
</table>

Modulus of elasticity = 1.0E6 lb/in**2
Area moment of inertia = 24 in**4

OUTPUT DATA

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency (rad/sec)</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>100.3552</td>
<td>1.000000</td>
<td>-2.774851</td>
</tr>
<tr>
<td>2</td>
<td>211.4977</td>
<td>1.000000</td>
<td>1.441519</td>
</tr>
</tbody>
</table>

TWO DOF RIGID BODY BEAM ON FLEXIBLE SUPPORTS

INPUT DATA

<table>
<thead>
<tr>
<th>load 1</th>
<th>radius of gyration</th>
<th>span length</th>
<th>c.g. from left support</th>
<th>c.g. from right support</th>
<th>spring constant 1</th>
<th>spring constant 2</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>lb/in</td>
<td>lb/in</td>
</tr>
<tr>
<td>5.6</td>
<td>3.9</td>
<td>10</td>
<td>6.93</td>
<td>3.07</td>
<td>15</td>
<td>5</td>
</tr>
</tbody>
</table>

OUTPUT DATA

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency (rad/sec)</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>23.30480</td>
<td>1.000000</td>
<td>-4.971451</td>
</tr>
<tr>
<td>2</td>
<td>65.67800</td>
<td>1.000000</td>
<td>0.2011484</td>
</tr>
</tbody>
</table>
### TABLE 5.1 (CONTINUE)

**THREE DOF SHEAR BUILDING FRAME**

**INPUT DATA**

<table>
<thead>
<tr>
<th>load 1</th>
<th>load 2</th>
<th>load 3</th>
<th>span length</th>
<th>load 1 from right</th>
<th>load 2 from right</th>
<th>load 3 from right</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>lb</td>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
</tr>
<tr>
<td>650</td>
<td>19300</td>
<td>38600</td>
<td>466</td>
<td>346</td>
<td>173</td>
<td>0</td>
</tr>
</tbody>
</table>

modulus of elasticity = 1.0E6 lb/in**2
area moment of inertia = 4320 in**4

**OUTPUT DATA**

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency rad/sec</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
<th>displacement coordinate mass 3</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>6.007885</td>
<td>1.000000</td>
<td>3.906237</td>
<td>6.108354</td>
</tr>
<tr>
<td>2</td>
<td>20.01055</td>
<td>1.000000</td>
<td>2.996532</td>
<td>-0.9990512</td>
</tr>
<tr>
<td>3</td>
<td>40.79921</td>
<td>1.000000</td>
<td>-0.1600301</td>
<td>1.0241550E-02</td>
</tr>
</tbody>
</table>
### THREE DOF SIMPLY SUPPORTED AND FIXED END BEAM

#### INPUT DATA

<table>
<thead>
<tr>
<th>load 1</th>
<th>load 2</th>
<th>load 3</th>
<th>span length from right support</th>
<th>load 1</th>
<th>load 2</th>
<th>load 3</th>
<th>span length from right support</th>
<th>span length from right support</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>lb</td>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>480</td>
<td>390</td>
</tr>
<tr>
<td>1.5</td>
<td>1.0</td>
<td>2.0</td>
<td>1.0</td>
<td>1.5</td>
<td>2.0</td>
<td>1.0</td>
<td>1.0</td>
<td>1.5</td>
</tr>
</tbody>
</table>

- modulus of elasticity = $1.0 \times 10^6$ lb/in$^2$
- area moment of inertia = 30 in$^4$

#### OUTPUT DATA

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency (rad/sec)</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
<th>displacement coordinate mass 3</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>32.31675</td>
<td>1.000000</td>
<td>1.400000</td>
<td>1.073848</td>
</tr>
<tr>
<td>2</td>
<td>114.0660</td>
<td>1.000000</td>
<td>3.2434639E-02</td>
<td>1.000000</td>
</tr>
<tr>
<td>3</td>
<td>275.5626</td>
<td>1.000000</td>
<td>-2.138684</td>
<td>1.000000</td>
</tr>
</tbody>
</table>

### THREE DOF SIMPLY SUPPORTED BEAM

#### INPUT DATA

<table>
<thead>
<tr>
<th>load 1</th>
<th>load 2</th>
<th>load 3</th>
<th>span length from right support</th>
<th>load 1</th>
<th>load 2</th>
<th>load 3</th>
<th>span length from right support</th>
<th>span length from right support</th>
</tr>
</thead>
<tbody>
<tr>
<td>lb</td>
<td>lb</td>
<td>lb</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>inch</td>
<td>480</td>
<td>360</td>
</tr>
<tr>
<td>3</td>
<td>2</td>
<td>3</td>
<td>1.5</td>
<td>2.0</td>
<td>480</td>
<td>360</td>
<td>240</td>
<td>120</td>
</tr>
</tbody>
</table>

- modulus of elasticity = $1.0 \times 10^6$ lb/in$^2$
- area moment of inertia = 90 in$^4$

#### OUTPUT DATA

<table>
<thead>
<tr>
<th>mode</th>
<th>natural frequency (rad/sec)</th>
<th>displacement coordinate mass 1</th>
<th>displacement coordinate mass 2</th>
<th>displacement coordinate mass 3</th>
</tr>
</thead>
<tbody>
<tr>
<td>1</td>
<td>114.4588</td>
<td>1.000000</td>
<td>1.504153</td>
<td>1.073848</td>
</tr>
<tr>
<td>2</td>
<td>313.5506</td>
<td>1.000000</td>
<td>0.4624190</td>
<td>-1.234211</td>
</tr>
<tr>
<td>3</td>
<td>796.5601</td>
<td>1.000000</td>
<td>-1.574680</td>
<td>0.4098067</td>
</tr>
</tbody>
</table>
1988

NASA/ASEE SUMMER FACULTY RESEARCH FELLOWSHIP PROGRAM

JOHN F. KENNEDY SPACE CENTER
UNIVERSITY OF CENTRAL FLORIDA

STUDIES OF CRYOGENIC PROPELLANT STORAGE AND HANDLING
FOR THE LUNAR LANDING AND LAUNCH FACILITY (COMPLEX 39L)

Prepared By: Jerald Linsley

Academic Rank: Associate Professor

University and Department: Florida Institute of Technology
Chemical Engineering

NASA/KSC:
Division: Advanced Projects, Technology
and Commercialization Office
Branch: Advanced Systems and Technology

NASA Counterpart: Dennis Matthews

Date: August 19, 1988

Contract No.: University of Central Florida
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