International Space Station
Lithium-Ion Battery

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ISS Li-Ion Battery - Outline

• Configuration of Existing ISS Electric Power System
• Timeline of Li-Ion Battery Development
• Battery Design Drivers
• Technical Definition Studies
• Cell Selection
• Safety Features
• Final Flight Adapter Plate and Battery Design
• Battery Charge Control and Low Earth Orbit (LEO) Cycle Test Data
• Current Status
ISS Configuration - Battery Locations

Batteries are located in the four Integrated Equipment Assemblies (IEAs)

Two Power Channels per IEA

Six Ni-H₂ Orbital Replacement Units (ORUs) per channel – 48 total

One Li-Ion and one Adapter Plate to replace two Ni-H₂ – 24 total Li-Ion batteries
ISS Configuration - EPS Schematic

Electrical Power Channel – 1 of 8

Note: 2-Battery ORUs will be replaced by 1 Li-ion Battery and an Adapter Plate

EPS:: Electric Power System  
BCDU: Battery Charge / Discharge Unit  
DCSU: DC Switching Unit  
DDCU: DC-to-DC Converter Unit  
MBSU: Main Bus Switching Units
Timeline of ISS Li-Ion Development

• **2009-2010** – Preliminary risk and feasibility studies
• **December 2011** - ISS Program Authority To Proceed with design, development and the fabrication of 27 Li-Ion ORUs and 25 on-orbit Adapter Plate ORUs
• **Jan-Jun 2012** - Cell Safety Testing and Cell Qualification
• **July 2012** - Final cell down-select
• **December 2012** - System Preliminary Design Review
• **November 2013** - System Critical Design Review
• **March 2016** - First flight Li-Ion battery delivered to Kennedy Space Center for shipment to Tanegashima, Japan
ISS Li-Ion Battery Key Design Drivers

- One Li-Ion battery ORU replaces two Ni-H$_2$ ORUs
- Launch on Japanese HTV
- Six year battery storage life requirement
- Ten year/60,000 cycle life target (minimum 48 A-hr capacity at end of life)
  - ORU will have cell balancing circuitry
  - ORU will have adjustable End of Charge Voltage (EOCV)
- Maximum battery ORU weight ~430 lbs
- Non-operating temperature range (Launch to Activation): -40 to +60 °C
- No changes to existing IEA interfaces and hardware
  - Use existing mounting, attachment, electrical and data connectors
  - Use existing Charge/Discharge Units and Thermal control systems
ISS Upgrade to Li-Ion

**Ni-H₂**
(76 cells in series)

**Li-Ion**
(30 cells in series)

**BCDU**: Battery Charge / Discharge Unit

**BIU**: Battery Interface Unit

**BSCCM**: Battery Signal Conditioning and Control Module

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**Battery A**

**Battery B**

**Commands & Data**

**Main Power Path**

**Existing**

**New**

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**Battery**

**Li-Ion Cells**

**Data Cable**

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BCDU: Battery Charge / Discharge Unit
BIU: Battery Interface Unit
BSCCM: Battery Signal Conditioning and Control Module
ISS Li-Ion Technical Definition Studies

- NASA Down Select to 4 cell candidates (April 2010)
- NASA Safety Risk Mitigation Activity (Jan 2009 – Sept 2010)
- Battery Mounting/ MOD Kit Feasibility Report (includes ORU Max Weight Assessment) (May 2010 – Sept 2010)
- Cell Selection NAR (Sept 2010)
- Battery ORU Specification and SOW Development (start Sept 2010)

Flow Diagram:
- 6 cell designs
- 4 cell designs
- 2 cell designs

- NASA Risk Mitigation Safety Report (Nov 2010)
ISS Li-Ion Cell Final Down-Select

- Two designs taken through qualification, with down-selection made prior to EM build

GS Yuasa 134 A-hr cells
- Li Cobalt Oxide / Carbon Graphite
- Wound elliptical prismatic electrode
- Internal Fusible link
- Aluminum Case, 50 x 130 x 263 mm
- Spec Mass: 3530 grams (~7.8 lb)
ISS Li-Ion Battery Safety Features

Battery-Level Safety Features

- Two independent controls vs. thermal runaway (two fault tolerant)
- Voltage and temperature monitoring of all 30 cells
- Circuit protection/fault isolation at the individual cell level for both high/low voltage and high temperature
- Physical separation between cell pairs and 10 packs
  - Thermal radiant barriers between cell pairs
- Controlled direction of cell vents - prevent damage to cold plate, adjacent cells and IEA hardware
  - ORU pressure relief/flame trap to prevent ORU over-pressurization but contain flame in the event of a cell vent
- MMOD shielding in ORU and empty ORU slot
- Dead face device to remove power from output connector during ground or EVA handling
- Non propagation of failures beyond Battery ORU
Safety Features - MMOD Shielding

MMOD Shielding

MMOD test setup

Ballistic Limit Testing

Over Match - Penetration testing
10 mm 2017-T4 Aluminum Sphere @ 6.86 km/s

Overcharge Containment Testing

Note: Existing Ni-H₂ does not have MMOD (Micro-Meteoroid Orbital Debris) protection
Safety Features - Radiant Heat Barriers

• ORU Layout – three Cell “10-Packs” and 12 Radiant Barriers

Radiant Heat Barrier (12 per ORU)

• Higher margin against thermal runaway propagation
• One barrier between each cell pair
• Reflects 787 reach-back safety additions

~3.5” Spacing between 10-Packs
~2” Spacing between Cells
ISS Li-ion Cell Safety Features

Cell-Level Safety Features and Controls

- Manufacturing Process controls include 100% materials screening and chemical analysis plus annual configuration/production line audits
- Acceptance testing of 100% of cells
- Simulated LEO life cycle testing in 2% of cells in each lot
- For 1% of cells in each lot, 100 cycles at 100% DOD are performed, followed by DPA
- Cell vent before burst and directional vent away from base plate and adjacent cells
- Individual cell fusing (internal fusible link)
- Shutdown separators between electrode windings
- Case neutral and electrically insulated from ORU structure
ISS Li-Ion ORUs

Heater Matt
Heater Plate Assembly

P4 Connector
(stowed for launch)

EVA
Hand Hold

P1 & P2
Connectors

J3 Test
Connector

J4
Connector

Adapter Plate ORU
Dimensions (LxWxH): ~ 41” x 36” x 15”
Spec Weight: 85 Lbs

Li-ion Battery ORU
Dimensions (LxWxH): ~ 41” x 37” x 21”
Spec Weight: 435 Lbs
ISS Li-Ion Charge Control and Cycling

- Li-Ion charge current profile based on cell voltages
- Cell bypass/balancing at EOCV every orbit
- EOCV is ground command-able

<table>
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<th>Highest of the Cell Terminal Voltages</th>
<th>Charge Current</th>
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ISS Li-Ion Flight Battery Status

- Six Flight Li-Ion Adapter Plates on-dock in Japan, Tomioka: April 2016
- Six Flight Li-Ion Batteries on-dock in Japan, Tanegashima: May 2016
- Final charge to 4.1 V: May-June 2016
- Launch on HTV: NET October 2016
  - Each IEA will have three Li-Ion ORUs and three Ni-H₂ ORUs (not electrically connected) stored on top of three On-Orbit Adapter Plate ORUs
- Installation and start-up on ISS: October 2016
ISS Li-Ion Battery Future Plans

• Thermal runaway propagation testing is scheduled for May 2016 at White Sands Test Facility
• Six Li-Ion Batteries and six Adapter Plates launch in 2017, 2018, 2019 to provide a full complement on ISS

➢ Design challenges have been addressed
➢ Ready for successful and safe operation
Acknowledgments

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