Advanced Control and Autonomy Research

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NASA – DLR Meeting at NASA Ames
March 14, 2017
Outline

• Overview of ACES Group – Dr. Nhan Nguyen

• UAS Autonomy Research – Dr. Corey Ippolito

• Stall Recovery Guidance Research – Dr. Thomas Lombaerts

• Flexible Aircraft Flight Control Research – Dr. Sean Swei
Advanced Control and Evolvable Systems Group

• Advanced Control and Evolvable Systems (ACES) Group (21 people) within the Intelligent Systems Division (code TI) conducts advanced GNC research and multidisciplinary vehicle dynamic modeling and simulations

• > 90% of research are aeronautics with some space-related GNC

• Collaborate with other NASA centers (AFRC, LaRC, GRC), other government agencies (FAA, DHS), industry (Boeing and small business companies), and academia (U.S. universities and TU Delft)
Intelligent Adaptive Flight Control

• Core expertise of ACES group

Intelligent Flight Control System (IFCS) 2003 – 2006
• Sigma-Pi neural network MRAC (Model-Reference Adaptive Control)
• Team: NASA AFRC, NASA ARC, IV&V, Boeing

Integrated Resilient Aircraft Control (IRAC) 2007 – 2011
• NASA simplified MRAC / optimal control modification
• Team: NASA AFRC, NASA ARC
Adaptive Control & Guidance for Vehicle Autonomy

Adaptive control as enabler for vehicle autonomy

- Adaptation through closed-loop control and mission management
- Integration with vehicle adaptive physical hardware & software
- Interactions with other domains (human-machine interactions, prognostics, etc.)
Performance Adaptive Aeroelastic Wing (PAAW)

Overview
Conduct multidisciplinary research to develop advanced technologies for wing shaping control to reduce fuel consumption and improve safety of high aspect-ratio flexible wing transport aircraft.

Impacts
- 1% – 6% drag reduction which could translate into as much as $0.2B – $1.3B fuel savings (www.transtats.bts.gov/fuel.asp)
- Reduce aircraft weight and gust loads
- Improve passenger comfort and enable safe operation of flexible wing transport aircraft

Collaboration

Multi-Fidelity Modeling
- Multi-fidelity aero modeling (Cart3D, OverFlow, Cava, Vorlax, Vspaero)
- Coupled FEM (Beam3D, NAS3AN) with aero codes
- Aerelasticity / Aeroservoelasticity (ASE)

Multidisciplinary Optimization
- Aerodynamic design optimization for drag reduction
- MDO for drag minimization, load alleviation, and active ASE control

Performance Analysis
- Design trade-study
- Mission analysis / trajectory optimization to minimize fuel burn

ASE – Flight Dynamics
- Coupled ASE – rigid aircraft flight dynamics
- Gust modeling
- Actuator dynamics of ASE control effectors

ASE Flight Control
- ASE control (flutter suppression, load alleviation)
- Multi-objective flight control
- Real-time drag optimization

Control Actuation
- VCCTEF / continuous leading edge slit
- Distributed control surfaces
- Other novel actuation concepts

Aeroservoelasticity (ASE) Modeling
- Aerodynamic coefficients & stability & control derivatives
- Nonlinear 6-DOF flight dynamics

Variable Camber Continuous Trailing Edge Flap (VCCTEF) Drag Reduction Technology

Wind Tunnel Validation

Flexible Wing Modern Transport Aircraft

Integrated Vehicle Multidisciplinary Research Capabilities

Multi-Objective Wing Shaping Flight Control System

Collaboration with NASA AFRC

X-56A Collaboration with NASA AFRC
Mission Adaptive Digital Composite Aerostructure Technologies (MADCAT)

Objective:
Develop a revolutionary aerostructure concept by combining the lattice-based discrete construction and the multi-objective optimal flight controls to realize mission adaptive and aerodynamically efficient future air vehicles.

Approach:
To leverage emerging digital composite manufacturing and fabrication methods and utilize the “building-block” strategy to build high stiffness-to-density ratio, ultra-light aerostructures that can provide mission adaptivity for varying flight conditions.

Status:
A scaled UAV model, capable of wing morphing, was built and flight tested successfully.

Funding & Duration:
This project is funded by ARMD CAS Project. FY16-FY17.
Safe, Autonomous, and Routine Operation of Small-UAS in High-Density Low-Altitude Urban Environments

Corey Ippolito, Kalmanje Krishnakumar
December 5, 2016
SAFE50 Project Overview

Purpose
• Investigate onboard vehicle autonomy for safe, autonomous and routine small-UAS operations in high-density low-altitude urban environments, from the viewpoint of regulatory stakeholders and traffic system operators such as UTM.

Goal
• Develop vehicle-centric autonomy requirements allowing safe operations
• Investigate the trade-space (including capabilities, challenges, implications, and alternatives)
• Incorporate and disseminate into UTM for TCL-3, TCL-4

Approach
• Top-down analysis, requirements-driven approach
• Develop prototypes, perform simulation and flight testing experiments, work with external partners
Motivating Scenario and Characteristics

Through out the day, thousands of operators and vehicles utilize the airspace above the city. UAS requests appear randomly throughout the day, requesting navigation between random location/address within in the city.

- Urban canyons
- Constrained spaces
- High-density
- High-demand
- Large-scale concurrent operations
- Operating over high-valued assets
Scope and Definitions

- **Routine Operations**
- **Small Unmanned Aerial Systems (UAS)**
  - Mass up to 25 kg to 150 kg *
  - Airspeed up to 40 m/s *
- **Low-Altitude Operations**
  - Altitude up to 200m or 400m **
- **High-Density Urban Environments**
  - “Non-trivial density” of humans, human structures, infrastructure, and competing air traffic
  - Includes residential/commercial/high-rise buildings, towers, roads, bridges, railways, manned aircraft (particularly rotorcraft), and other UAS
  - Density metrics to be defined, may include
    - Population density greater than 1,000 people per square mile ***
    - Perhaps a minimum threshold UAS capacity per city block, such as 10 UAS per city block

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* NASA UAS Classification Matrix

<table>
<thead>
<tr>
<th>Type</th>
<th>Model or sUAS</th>
<th>sUAS</th>
<th>UAS</th>
</tr>
</thead>
<tbody>
<tr>
<td>Category</td>
<td>I (1)</td>
<td>II (2)</td>
<td>III (3)</td>
</tr>
<tr>
<td>Weight Limit</td>
<td>≤ 25 kg</td>
<td>25-150 kg</td>
<td>&gt; 150 kg</td>
</tr>
<tr>
<td>Airspeed Limit</td>
<td>≤ 40 m/s</td>
<td>≤ 100 m/s</td>
<td>&gt; 100 m/s</td>
</tr>
</tbody>
</table>

* per NASA NPR 7900.3C, Appendix I
** per U. S. Class G Airspace, typ. below 700ft/1200ft
*** per Geographic Areas Reference Manual (GARM), U.S. Census Bureau, 1994
Challenges for High-Density Urban Operations

- Flight operations occur almost entirely beyond RF communications line-of-sight from ground operators to the vehicle. Limited point-to-point or satellite line-of-site. **Need for Autonomy.**

- **Atmospheric uncertainties** may have major impact on safety.

- Flight operations occur in a **GPS-denied** (or at-best a GPS-degraded) environment. Limited satellite line-of-site.

- Obstacles and hazards are not known with certainty ahead of time. Autonomous **onboard see-and-avoid** may an onboard vehicle requirement.

- Vehicle system **failures** has major impact on safety (high failure rates, single-string architectures).

- Real-time ground-based surveillance is not easily accomplished. **Separation assurance** may be an onboard vehicle autonomy requirement.
SAFE50 Vehicle Autonomy Requirements

- **Detect**: equivalent to ‘see’ or ‘sense’, cooperative or noncooperative, technology limits, SWaP constraints, cost implications
- **Operate-Near**: more stringent than ‘avoid’
- **Avoid-Endangering**: responsibility of risk and damage assigned to vehicles and operators
Tall-Poles and Research Focus

- Detection of objects (static, dynamic, other air vehicles)
- Classification of objects (as needed to satisfy requirements)
- Relative control to objects
- Decision making under uncertainty
- Resilience to wind gusts and micro-weather effects
- Resilience to onboard failures
- Risk minimizing nominal/off-nominal control
- Alternative/augmentations to GNSS-derived position, navigation, and timing
Approach
Research Challenges

1. UrbanScape Wind Uncertainties

High velocity region

2. GPS Denied/Degraded

3. Static/Dynamic Obstacles
UrbanScape Wind Uncertainties

Urban Architecture and CFD Simulation of Wind Profiles.
GPS Denied/Degraded Navigation
Powerline Identification and Reconstruction. Raw LiDAR point clouds (left), voxel processing (middle), reconstructed powerlines (right), at 20m (top)
Test Environments

Ames NUARC Facility

Ames Roverscape

S1000 Tethered Flight Test Roverscape 2010/09/09

Ames DART Facility
Stall Recovery Guidance

NASA-DLR meeting on Unmanned Aircraft research topics
Stefan Schuet, John Kaneshige, Thomas Lombaerts,
Kimberlee Shish, Vahram Stepanyan, Gordon Hardy,
Peter Robinson
NASA Ames Research Center, Moffett Field
Structure

• Introduction
• Guidance strategies
• Display
• Simulation results
• Two implementation stages
Introduction

- Loss of control in flight remains the most frequent primary cause of accidents

- Stall related accidents: Colgan Air 3407, AirAsia 8501, XL Airways Germany 888T, Air Algerie 5017, Air France AF447,…

CAST studies:
- Effective Upset Prevention and Recovery Training
- Airplane State Awareness by aircrew (SE207)
  - Algorithms and display strategies to provide control guidance for recovery from approach-to-stall or stall
Introduction

Research subtopics, based on CAST directives on safety enhancements:

1. **Upset prevention**
   - Adaptive safe flight envelope estimation
   - Autoflight trajectory prediction and alerting
   - Adaptive envelope protection

2. **Upset recovery**
   1. Stall recovery guidance
   2. Unusual attitude recovery

Previous work published in the literature:

Sequence of events for stall recovery

<table>
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<th>Onset to Stall</th>
<th>Stall Occurrence</th>
<th>Stall Recovery</th>
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<tbody>
<tr>
<td>Decreasing airspeed, increasing angle of attack</td>
<td>Speed below stall speed, alpha exceeds stall value</td>
<td>Accelerating dive</td>
</tr>
<tr>
<td>FAA stall recovery template:</td>
<td>Speed below stall speed, alpha exceeds stall value</td>
<td>Trade altitude for speed, potential → kinetic energy</td>
</tr>
</tbody>
</table>
Guidance strategies

3 strategies under consideration:
• Fast model predictive control (AIAA-2017-1513)
• Energy based guidance (AIAA-2017-1021)
• Constrained control approach (AIAA-2016-0878)

Display
Simulation results

Angle of attack and calibrated airspeed

Energy transfers

Intro – Guidance strategies – Display – **Results** – Implementation
Two implementation stages

- Advanced Controls Technologies (ACT) lab
  - Fall 2016

- Vertical Motion Simulator (VMS)
  - Spring 2017

Intro – Guidance strategies – Display – Results – Implementation
Milestones

1. Complete an initial multi-objective optimization study with aeroelastic finite-element wing model coupled with longitudinal flight dynamic model, and subject to appropriate constraints.

2. Complete an initial robust control and distributed parameter control system design for coupled aircraft dynamics with aeroelastic wing structures to suppress fluttering motion.

Milestones:
Aeroelastic Stability Augmentation/Flutter Suppression Control

National Aeronautics and Space Administration
Flight Controls for Flexible Air Vehicles

Prepared by
Sean Swei and Kenneth Cheung
NASA Ames Research Center

in collaboration with
NASA, MIT, Michigan State U., UCSC, U. of Alabama

March 14, 2017
Outline

◆ Mission Adaptive Digital Aerostructure Technologies (MADCAT)

◆ Linear Parameter Varying (LPV) Modeling & Controls
  ➢ Model Alignment
  ➢ Adaptive Grid Step Size Determination

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MADCAT v0

OPTIMAL WING SHAPING CONTROL
Mission Adaptive Digital Aerostructure Technologies (MADCAT)

- Develop a revolutionary aerostructure concept for future air vehicles by combining:
  - lattice-based discrete/digital construction approach
  - multi-objective optimal flight controls

_to realize mission adaptive and aerodynamically efficient air vehicles_

Component Development  Digital Construction  Performance Assessment (wind tunnel tests)
Mission Adaptive Digital Aerostructure Technologies (MADCAT)

Performance Assessment: MADCAT v0 Flight Tests

N719NU
Linear Parameter Varying (LPV) Modeling

- Flexible Model Alignment

NASA Generic Transport Model (GTM)
Note: $\sigma$-shifted $\mathcal{H}_2$-norm, denoted by $\mathcal{H}_{2-\sigma}$-norm, is defined by

$$\left\| G_j^i(s) \right\|_{2-\sigma}^2 = \left\| G_j^i(s + \sigma) \right\|_2^2$$
Switch LPV Controls

- Sequential design of hysteresis switching LPV controllers

**Control Design Objectives:**

*Sequently design a family of LPV controllers, such that: 1) the closed-loop systems are stable; 2) the controllers switch smoothly between neighboring controllers; 3) the flight performances are improved.*

**LMI Characterization**
Switching LPV Controls

§ Sequential hysteresis switching LPV controllers

Scheduling parameter division

Control Objective:
Sequentially design a family of LPV controllers such that:
1) the closed-loop systems are stable;
2) the controllers switch smoothly between neighboring controllers;
3) the performances are improved.

LMI Characterization

Thank You!

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Thank You