NASA’s Advanced Environmental Barrier Coatings Development for SiC/SiC Ceramic Matrix Composites: Understanding Calcium Magnesium Alumino-Silicate (CMAS) Degradations and Resistance

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Currently related NASA CMAS research activities:

• Advanced EBC development – composition design and developments for improved CMAS resistance

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Outline

• Environmental barrier coating (EBC) development: the CMAS relevance

• Some generalized CMAS related failures

• CMAS degradation of environmental barrier coating (EBC) systems: rare earth silicates
  – Ytterbium silicate and yttrium silicate EBCs
  – Some reactions, kinetics and mechanisms

• Advanced EBCs, HfO$_2$- and Rare Earth - Silicon based 2700°F+ capable bond coats

• Summary
NASA Environmental Barrier Coatings (EBCs) and Ceramic Matrix Composite (CMC) System Development

- Emphasize material temperature capability, performance and long-term durability. Highly loaded EBC-CMCs with temperature capability of 2700°F (1482°C)

- 2700-3000°F (1482-1650°C) turbine and CMC combustor coatings
- 2700°F (1482°C) EBC bond coat technology for supporting next generation
  - Recession: <5 mg/cm² per 1000 h
  - Coating and component strength requirements: 15-30 ksi, or 100- 207 Mpa
  - Resistance to Calcium Magnesium Alumino-Silicate (CMAS)

Step increase in the material’s temperature capability

2700°F (1482°C) Gen I and Gen II SiC/SiC CMCs
2400°F (1316°C) Gen I and Gen II SiC/SiC CMCs
2800ºF combustor TBC
2700ºF Turbine TBC
2500ºF Turbine TBC
2800ºF combustor TBC

Increase in ΔT across T/EBC

3000°F+ (1650°C+)

3000°F SiC/SiC CMC airfoil and combustor technologies

2700°F SiC/SiC thin turbine EBC systems for CMC airfoils

2000°F (1093°C), PtAl and NiAl bond coats
EBC-CMAS Degradation is of Concern with Increasing Operating Temperatures

- Emphasize improving temperature capability, performance and long-term durability of ceramic turbine airfoils

  • Increased gas inlet temperatures for net generation engines lead to significant CMAS-related coating durability issues – CMAS infiltration and reactions


Calcium Magnesium Alumino-Silicate (CMAS) Systems Used in Laboratory Tests

- Synthetic CMAS, modified version (NASA), the Air Force PTI 02 CMAS currently being used
- Saudi Sands used for past turbine TBC studies
- CMAS SiO₂ content typically ranging from 43-49 mole%
- Collaborations on-going with the Air Force; also planned DLR, ONEA etc on Volcanic Ash Composition selections

ARFL PTI 02 is also used at NASA for CMAS studies
Thermal Gradient Tests - Thermal Barrier Coating Degradations

— Coating cyclic failure with CMAS

Steady state laser thermal gradient test

7YSZ turbine EB-PVD laser rig cyclic tested, after 50 hr cyclic test at $T_{\text{surface}}$ 1230°C and $T_{\text{interface}}$ 1170°C
Thermal Gradient Tests of Infiltrated and Reacted Apatite Phase under Cyclic Testing – Thermal Barrier Coating Degradations

— Coating surface layer spallation in infiltrated or highly reacted apatite phase layer in high rare earth dopant TBC systems
— Thermal gradient cyclic testing at $T_{\text{surface}}=1316^\circ\text{F}$, $T_{\text{interface}}=950^\circ\text{F}$

After thermal gradient cyclic testing
CMAS Related Degradations in EBCs

- **CMAS effects**
  - Significantly reduce melting points of the EBCs and bond coats
  - Cause more severe degradations with thin airfoil EBCs
  - CMAS increase EBC diffusivities and permeability, thus less protective as an environmental barrier
  - Reduced mechanical properties: such as strength and toughness reductions
  - Leads to grain boundary attack thus disintegrate EBCs
  - CMAS interactions with heat flux, thermal cycling, erosion and thermomechanical fatigue
CMAS Related Degradations in EBCs - Continued

- **CMAS effects on EBC temperature capability**
  - Silicate reactions with NaO₂ and Al₂O₃ silicate
EBC-CMAS Degradation under Thermal Gradients

- **Effect of CMAS concentration on EBC-CMC system cyclic durability**
  - CMAS reacts with high SiO$_2$ activity layer and reducing melting point
  - Low tough reaction layers such as apatite phases
  - Interactions with heat flux, thermal cycling, erosion and thermomechanical fatigue

More severe degradation and delamination:
- T$_{surface}$ 1500°C
- T$_{interface}$ 1316°C
NASA EBC Systems

- **HfO$_2$ -RE$_2$O$_3$ -SiO$_2$/RE$_2$Si$_{2-x}$O$_{7-2x}$ environmental barrier systems**
  - Controlled silica content and transition element and rare earth dopants to improve EBC stability and toughness
  - Develop HfO$_2$-Si based + X (dopants) and more advanced rare earth composite compound composition systems for 2700°F+ long-term applications
  - Develop prime-reliant composite EBC-CMC interfaces for fully integrated EBC-bond coat systems
- **RE$_2$O$_3$-SiO$_2$-Al$_2$O$_3$ Systems**
- Develop advanced NASA high toughness alternating layered systems
- **Advanced 1500°C bond coats**
Strength Results of Selected EBC and EBC Bond Coats

- CMAS Reaction resulted in Strength Reduction in Silicates

Selected EBC systems
- HfO\textsubscript{2}-RE-Si, along with co-doped rare earth silicates and rare earth alumino-silicates, for optimized strength, stability and temperature capability
- CMAS infiltrations can reduce the strength

![Graph showing strength test data compared for various EBCs and EBC bond coat constituents.](image)

- 646-Specially alloy toughened t' like HfO2
- 648-EBC Bond Coat Constituent
- 658-EBC HfO2-Rare Earth-Alumino Silicate
- 660-Y2Si2O7
- 669-Yb2Si2O7
- 696-EBC Bond Coat Constituent
- 659-AE9762
- 681-HfO2-Si
- 669-9762 with CMAS 1350C 50hr reacted
- 693-HfO2-Si with CMAS 1350C 50hr reacted

![Images of Yb\textsubscript{2}Si\textsubscript{2}O\textsubscript{7} CMAS reacted tensile surface and Yb\textsubscript{2}Si\textsubscript{2}O\textsubscript{7} CMAS reacted specimen fracture surface.](image)
Effect of CMAS Reaction on Toughness of HfO$_2$-Si Bond Coat and Yb$_2$Si$_2$O$_7$ EBC

- HfO$_2$-Si bond coat and ytterbium di-silicate fracture toughness studied
  - HfO$_2$-Si toughness >4-5 MPa m$^{1/2}$ achieved at higher temperature
  - Annealing heat treatments at 1300°C improved lower temperature toughness
  - CMAS effect unclear due to the compounded effects of possible 1350°C CMAS reaction degradation and annealing
- Ytterbium silicate EBC toughness may also be reduced due to CMAS reactions
  - More measurements are needed

![HfO$_2$-Si illustrating notch distortion due to CMAS exposure at 1350°C for 50 hrs](image)

![Yb$_2$Si$_2$O$_7$ notch after CMAS exposure at 1350°C for 50 hrs](image)

![Graph showing fracture toughness](image)
EBC CMAS Surface Reactions

- Ytterbium and yttrium silicate reactions and dissolutions in CAMS

Ytterbium silicate surface CMAS melts: 50 hr 1300°C

Ytterbium silicate surface CMAS melts: 5 hr 1500°C

Yttrium silicate surface CMAS melts: 50 hr 1300°C

Yttrium silicate surface CMAS melts: 5 hr 1500°C
EBC Reacted Apatite Phases under Long-Term Testing at 1500°C – Ytterbium silicate EBC

- Non stoichiometric characteristics of the CMAS – rare earth silicate reacted apatite phases
- Difference in partitioning of ytterbium vs. yttrium in apatite

Composition in apatite (100 hr):
EBC Reacted Apatite Phases under Long-Term Testing at 1500°C: Yttrium Silicate EBC

- Non stoichiometric characteristics of the CMAS – rare earth silicate reacted apatite phases
- Difference in partition of ytterbium vs. yttrium
  - Average AEO/RE$_2$O$_3$ ratio ~ 0.68 for ytterbium silicate – CMAS system
  - Average AEO/RE$_2$O$_3$ ratio ~ 0.22 for yttrium silicate – CMAS system

Composition in apatite (100 hr):
Stoichiometry of the Reacted Apatite Phases under Long-Term Testing at 1500°C

- Non stoichiometric characteristics of the CMAS – rare earth silicate reacted apatite phases – up to 200 hr testing
- Difference in partitioning of ytterbium vs. yttrium in apatite
  - Average AEO/RE$_2$O$_3$ ratio $\sim 0.68$ for ytterbium silicate – CMAS system
  - Average AEO/RE$_2$O$_3$ ratio $\sim 0.22$ for yttrium silicate – CMAS system

Partitioning of Rare Earths in Apatite in Geo Systems: Medium Ionic Rare Earth Reported higher Partitioning Coefficients

Reported partition of Rare Earths in Apatite


Fig. 1. Comparison of $D_{REE}$ values calculated from SIMS analyses (open squares) and LA-ICPMS analyses.

Fig. 3. Plots of $D(REE)$ as a function of atomic number for series AP50 experiments. Note that AP50-2 and AP30-3 with low total REE contents have more pronounced enrichments in middle REEs than AP50-1 and AP50-4.
Effect of CMAS Reactions on Grain Boundary Phases

- CMAS and grain boundary phase has higher Al$_2$O$_3$ content (17-22 mole%)
  - Eutectic region with high Al$_2$O$_3$ content ~1200°C melting
  - Loss of SiO$_2$ due to volatility

NASA modified CMAS

Grain boundary final phase – low SiO$_2$ and high Alumina

200 hr, 1500°C
Rare Earth Apatite Grain Growth

- Grain growth of apatite phase at 1500°C at various times
Effect of CMAS Reaction on EBC Cyclic Durability in Thermal Gradient Laser Steam Rig

- Ytterbium silicate EBC Yb$_2$Si$_2$O$_7$/Si on CMC
- CMAS fully infiltrated
- Failed after 40 cycles (1hr cycle) under combined laser thermal gradient CMAS+steam at 1400-1500°C
- Accelerated recession leading to cracking and porous coatings

![Temperature vs Time Graph](image1.png)

- $T_{surface}$
- $T_{back}$

![Coating Spalled Image](image2.png)
- Coating spalled after 40 hr test
- Failed coating surface
HfO$_2$-Rare Earth Silicate Composite EBC with Yb-Si Bond Coat Systems

- Generally showed good resistance in CMAS and CMAS-steam tests
  - Composite system for achieving balanced CMAS resistance and water vapor stability at 1500°C
  - Compositions being further optimized
HfO$_2$-Rare Earth Silicate Composite EBC Systems - Continued

- Silica loss observed in the concentrated CMAS reacted regions

CMAS concentrated region, SiO$_2$ content 20-30 mol% (SiO$_2$ loss in the steam water vapor tests)
High Stability Rare Earth Silicon Bond Coat with High Melting Point Coating Compositions

- Thermogravimetric analysis (TGA) in dry O₂ at 1500°C, tested up to 500 hr
- “Protective” scale of rare earth di-silicate forms in oxidizing environments
- Furnace cyclic test life evaluated at 1500°C

FCT life of RE-Si coatings

TGA results vs. Si content
High Stability and CMAS Resistance Observed from the Rare Earth Silicon High Melting Point Coating Compositions

- Demonstrated CMAS resistance of NASA RESi System at 1500°C, 100 hr
- Silica-rich phase precipitation
- Rare earth element leaching into the melts (low concentration ~9mol%)
CMAS Reaction Kinetics in Bond Coats

- SiO$_2$ rich phase partitioning in the CMAS melts
- Rare earth content leaching low even at 1500°C

CMAS Partitioning on RE-Si bond coat, 1500°C, 100hr
Turbine TEBC Life Aspects due to CMAS Degradations and Other Mechanisms

- Reduced cyclic life due to the CMAS infiltration

- Surface vertical cracks
- Delamination cracks

Graph showing:

- In (Oxidation cycle time to failure), hours
- Erosion resistance, grams erodent required per 25 micrometer coating erosion
- Temperature/Interface temperature
- Mach 0.3
- Oxidation-erosion

Plot data points for:
- Zr2.5Y0.75Gd0.75Yb
- Zr2.0Y1.5Gd1.5Yb
- Zr1.6Y1.2Gd1.2Yb
- Zr2.5Y0.75Gd0.75Yb
- Zr2.0Y1.5Gd1.5Yb
- Zr1.6Y1.2Gd1.2Yb
- Zr2.5Y0.75Gd0.75Yb
- Zr2.0Y1.5Gd1.5Yb
- 7YSZ
- 7YSZ Laser heat flux rig
- 7YSZ burner heat flux rig
Summary

• CMAS degradation remains a challenge for emerging turbine engine environmental barrier coating – SiC/SiC CMC component systems
• CMAS leads to lower melting point of EBC and EBC bond coat systems, and accelerated degradations
• NASA advanced HfO$_2$-Si and, in particular Rare Earth - Silicon based bond coat compositions showed promise for CMAS resistance at temperatures up to 1500°C
• We have better understanding of CMAS integration with rare earth silicates, and in controlling the compositions for CMAS resistance while maintaining high toughness
• We are developing better standardized CMAS testing, and working on CMAS induced life reductions, helping validate life modeling